

THE

Cherokee **ARCHER**

PILOT'S OPERATING MANUAL



BY



This manual is incomplete without an APPROPRIATE FAA APPROVED AIRPLANE FLIGHT MANUAL and an APPROPRIATE WEIGHT AND BALANCE REPORT.

WARNING

EXTREME CARE MUST BE EXERCISED TO LIMIT THE USE OF THIS MANUAL TO APPLICABLE AIRCRAFT. THIS MANUAL REVISED AS INDICATED BELOW OR SUBSEQUENTLY REVISED IS VALID FOR USE WITH THE AIRPLANE IDENTIFIED BELOW WHEN APPROVED BY PIPER AIRCRAFT CORPORATION. SUBSEQUENT REVISIONS SUPPLIED BY PIPER AIRCRAFT CORPORATION MUST BE PROPERLY INSERTED.

MODEL PA-28-180

AIRCRAFT SERIAL NO. 28-7505222 REGISTRATION NO. N1185X
PR750120
PILOT'S OPERATING MANUAL, PART NUMBER 761 556 REVISION 1-20-75

PIPER AIRCRAFT CORPORATION
APPROVAL SIGNATURE AND STAMP


T.W. Reed, Jr.

Assurance that the airplane is in an airworthy condition is the responsibility of the owner. The pilot in command is responsible for determining that the airplane is safe for flight. The pilot is also responsible for remaining within the operating limitations outlined by the Airplane Flight Manual, instrument markings, and placards.

This Pilot's Operating Manual is not designed as a substitute for adequate and competent flight instruction, knowledge of the current airworthiness directives, applicable federal air regulations, or advisory circulars. It is not intended to be a guide for basic flight instruction or a training manual for transition from single to multi-engine flying.

If an inconsistency of information exists between the Pilot's Operating Manual and the Airplane Flight Manual approved by the FAA, the Airplane Flight Manual shall be the authority.

A complete or partial replacement of this manual, Part No. 761 556, may be obtained only from Piper Customer Services.

Published by
PUBLICATIONS DEPARTMENT
Piper Aircraft Corporation
761 556
Issued: July 1973

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APPLICABILITY

This manual is applicable to Piper Model PA-28-180 aircraft having serial numbers 28-7405001 through 28-7505259. Contact Piper Customer Services for specific information on the application of this manual.

REVISIONS

The information compiled in the Pilot's Operating Manual will be kept current by revisions distributed to the airplane owners.

Revision material will consist of information necessary to update the text of the present manual and/or to add information to cover added airplane equipment.

I. Revisions

Revisions will be distributed whenever necessary as complete page replacements or additions and shall be inserted into the manual in accordance with the instructions given below:

1. Revision pages will replace only pages with the same page number.
2. Insert all additional pages in proper numerical order within each section.
3. Page numbers followed by a small letter shall be inserted in direct sequence with the same common numbered page.

II. Identification of Revised Material

Revised text and illustrations shall be indicated by a black vertical line along the left hand margin of the page, opposite revised, added or deleted material. A line opposite the page number or section title and printing date, will indicate that the text or illustration was unchanged but material was relocated to a different page or that an entire page was added.

Black lines will indicate only current revisions with changes and additions to or deletions of existing text and illustrations. Changes in capitalization, spelling, punctuation or the physical location of material on a page will not be identified by symbols.

III. Original Pages Issued

The original pages issued for this manual prior to revision are given below:

1-1 through 1-3, 2-1 through 2-18, 3-1 through 3-18, 4-1 through 4-6, 5-1 through 5-27, 7-1 through 7-10, 8-1 through 8-2, 9-1 through 9-10, 10-1 through 10-15.

PILOT'S OPERATING MANUAL LOG OF REVISIONS

Current Revisions to the PA-28-180 Cherokee Archer Pilot's Operating Manual, 761 556, issued July 9, 1973.

Revision	Revised Pages	Description	Date
Rev. 1 - 761 556 (PR731106)	2-8	Revised Overvoltage Relay reset time.	Nov. 6, 1973
	4-6	Revised Alternator Failure Item 3.	
	W/B	Added Rev. 1 to Report: VB-547	
	7-7	Added Airspeed Data; Relocated E.L.T.	
	7-8	Added E.L.T.; Relocated Air Conditioning and Air Conditioner Operational Check Procedure.	
	7-9	Added Air Conditioning; Relocated Air Conditioner Effects on Airplane Performance.	
	7-10	Added Air Conditioner Operational Check Procedure and Air Conditioner Effects on Airplane Performance.	
	7-11	Added page.	
	7-12	Added page.	
	8-1	Revised item 6.	
Rev. 2 - 761 556 (PR740530)	ii	Added PAC Approval Form.	May 30, 1974
	iii	Added Applicability and Item III. Original Pages Issued.	
	A F/M W/B	Added Rev. 1 to Report: VB-558. Added Rev. 2 to Report: VB-547. (NOTE: AIRCRAFT DELIVERED WITH MANUALS PRIOR TO THIS REVISION DO NOT REQUIRE THIS REVISION.)	
Rev. 3 - 761 556 (PR740617)	2-8	Added Annunciator Panel information and footnote.	June 17, 1974
	2-8a	Added page (added material from page 2-8).	
	2-9	Relocated page (Alternator-Starter Schematic).	
	2-9a	Added page (Alternator-Starter Schematic).	
	2-11	Added Annunciator Panel information and footnote.	
	2-12	Revised Instrument Panel illustration.	
	2-16	Added Seat Removal instructions.	
	A F/M	Added Rev. 2 to Report: VB-558.	
	W/B	Added Rev. 3 to Report: VB-547.	
	7-4	Added Annunciator Panel check to Warm-up and Ground Check; added footnote.	
8-1	Revised item 6; added footnote.		
8-2	Relocated material from page 8-1.		

PILOT'S OPERATING MANUAL LOG OF REVISIONS (cont)

Revision	Revised Pages	Description	Date
Rev. 3 (cont)	9-i	Revised Performance Charts Index.	
	9-2	Revised Takeoff Chart.	
	9-3	Revised Climb Chart.	
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	9-5	Revised Engine Chart.	
	9-6	Revised Airspeed Chart.	
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	9-9	Revised Glide Chart.	
	9-10	Revised Landing Chart.	
	10-11	Revised Battery Service.	
Rev. 4 - 761 556 (PR750120)	1-2	Added O-360-A4M engine designation and footnote.	Jan. 20, 1975
	2-1	Added O-360-A4M engine designation and footnote.	
	A F/M	Added Rev. 3 to Report: VB-558.	
	W/B	Added Rev. 4 to Report: VB-547.	
	7-4	Added new item 9.; revised existing item nos. under TAKEOFF.	
	7-8	Revised ELT info.	
	8-2	Added item 11.	
	10-8	Added O-360-A4M engine designation under Oil Requirements.	
Rev. 5 - 761 556 (PR751121)	iii	Added concluding ser. no. to Applicability.	Nov. 21, 1975
	1-1	Revised Cruise Speed and Range figures; revised Empty Weight and Useful Load; deleted footnote.	
	1-2	Deleted footnote.	
	2-1	Revised Airframe info; deleted footnote.	
	A F/M	Added Rev. 4 to Report: VB-558.	
	W/B	Added Rev. 5 to Report: VB-547.	
	8-1	Revised item 8 (Fuel Warning Tip).	
	9-4	Revised Cruise Performance - Range chart.	
	9-6	Revised Cruise Performance - True Airspeed chart.	
Rev. 6 - 761 556 (PR790326)	2-8	Added Warning.	March 26, 1979
	2-8a	Added info.	
	2-14	Added Caution.	
	W/B	Added Rev. 6 to Report: VB-547.	
	7-2	Revised items.	
	7-8, 7-9	Revised ELT info.	

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Revision	Revised Pages	Description	Date
Rev. 7 - 761 556 (PR840316)	1-2 2-16 AFM W/B 7-1 7-4 9-i 10-1 10-9 10-11 10-12, 10-13	Revised fuel and oil. Revised para. Added Rev. 5 to Report VB-558. Added Rev. 7 to Report VB-547. Added to preflight. Added to text info. Added Warning. Added to text info. Added text info. Revised para. Revised preventive maintenance info.	March 16, 1984

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GENERAL SPECIFICATIONS

PERFORMANCE

Published figures are for standard airplanes flown at gross weight under standard conditions at sea level, unless otherwise stated. Performance for a specific airplane may vary from published figures depending upon the equipment installed, the condition of engine, airplane and equipment, atmospheric conditions and piloting technique. Each performance figure below is subject to the same conditions as on the corresponding performance chart from which it is taken in the Performance Charts Section.

Takeoff Ground Run, 25° flaps, sea level (ft)	720
Takeoff Distance Over 50-ft Obstacle, 25° flaps, sea level (ft)	1625
Best Rate of Climb Speed (mph)	85
Rate of Climb (ft per min)	725
Best Angle of Climb Speed, sea level (mph)	76
Max Speed, sea level (mph)	148*
Max Speed Optimum Altitude 9300 ft, 75% power (TAS) (mph)	141*
Service Ceiling (ft)	14,150
Absolute Ceiling (ft)	16,500
Cruise Speed at best power mixture (mph)	
65% power, 12,800 ft	132*
60% power, 14,000 ft	125*
Range at best power mixture (mi)**	
75% power, 9300 ft	645
65% power, 12,800 ft	705
Cruise speed at best economy mixture (mph)	
75% power, 9300 ft	131*
65% power, 12,600 ft	123*
Range at best economy mixture (mi)**	
75% power, 9300 ft	715
65% power, 12,600 ft	774
Stalling Speed, flaps down (CAS) (mph)	61
Stalling Speed, flaps up (CAS) (mph)	68
Landing Roll, sea level, flaps down (ft)	635
Landing Distance Over 50-ft Obstacle, sea level (ft)	1185

*All speeds stated are with optional wheel fairings installed. Subtract 3 mph if wheel fairings are not installed.

**No reserve.

WEIGHTS

Gross Weight (lbs)	2450
Standard Empty Weight (lbs)	1404
Maximum Useful Load (lbs)	1046

CHEROKEE ARCHER

POWER PLANT

Engine (Lycoming)	O-360-A4A or O-360-A4M
Rated Horsepower	180
Rated Speed (rpm)	2700
Bore (inches)	5.125
Stroke (inches)	4.375
Displacement (cubic inches)	361.0
Compression Ratio	8.5:1
Dry Weight (pounds)	285
Propeller	76EM8S5-0-60

FUEL

AVGAS ONLY

Fuel Capacity (U.S. gal) Usable	48
Fuel, Aviation Grade (min octane)	100/130

OIL

Oil Capacity (qts)	8
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BAGGAGE

Maximum Baggage (lbs)	200
Baggage Space (cubic ft)	24
Baggage Door Size (in.)	20 x 22

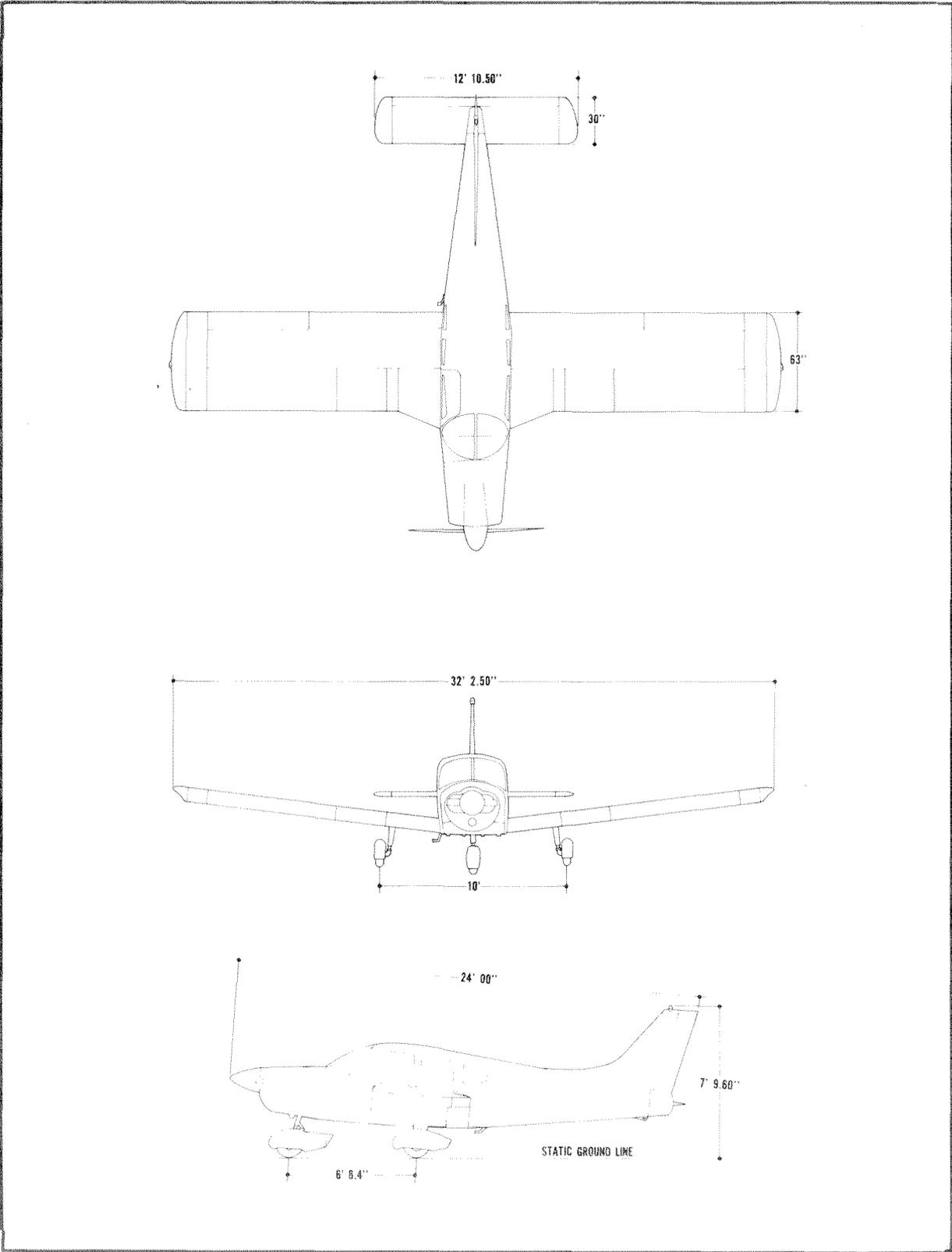
DIMENSIONS

Wing Span (ft)	32
Wing Area (sq ft)	170
Wing Loading (lbs per sq ft)	14.4
Length (ft)	24.0
Height (ft)	7.8
Power Loading (lbs per hp)	13.6

LANDING GEAR

Wheel Base (ft)	6.7	
Wheel Tread (ft)	10	
Tire Pressure (psi)	24	
	Nose	24
	Main	24
Tire Size	Nose (4 ply rating)	6.00 x 6
	Main (4 ply rating)	6.00 x 6

GENERAL SPECIFICATIONS
REVISED: MARCH 16, 1984



DESCRIPTION

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DESCRIPTION

AIRPLANE AND SYSTEMS

THE AIRPLANE

The PA-28-180 Cherokee is a single-engine, low-wing monoplane of all metal construction. Its full 4-place seating, two hundred pound baggage capacity, 645 mile range and economical operation, coupled with the lively performance of a 180 horsepower engine, make this Cherokee a versatile airplane in the business and personal aviation fields.

AIRFRAME

The basic airframe, except for a tubular steel engine mount, steel landing gear struts, and other miscellaneous steel parts, is of aluminum alloy construction. The extremities - the wing tips, the cowling, the tail surfaces - are of tough fiberglass.

The wings are attached to each side of the fuselage by insertion of the butt ends of the respective main spars into a spar box carry-through which is an integral part of the fuselage structure, providing, in effect, a continuous main spar with splices at each side of the fuselage. There are also fore and aft attachments at the rear spar and at an auxiliary front spar.

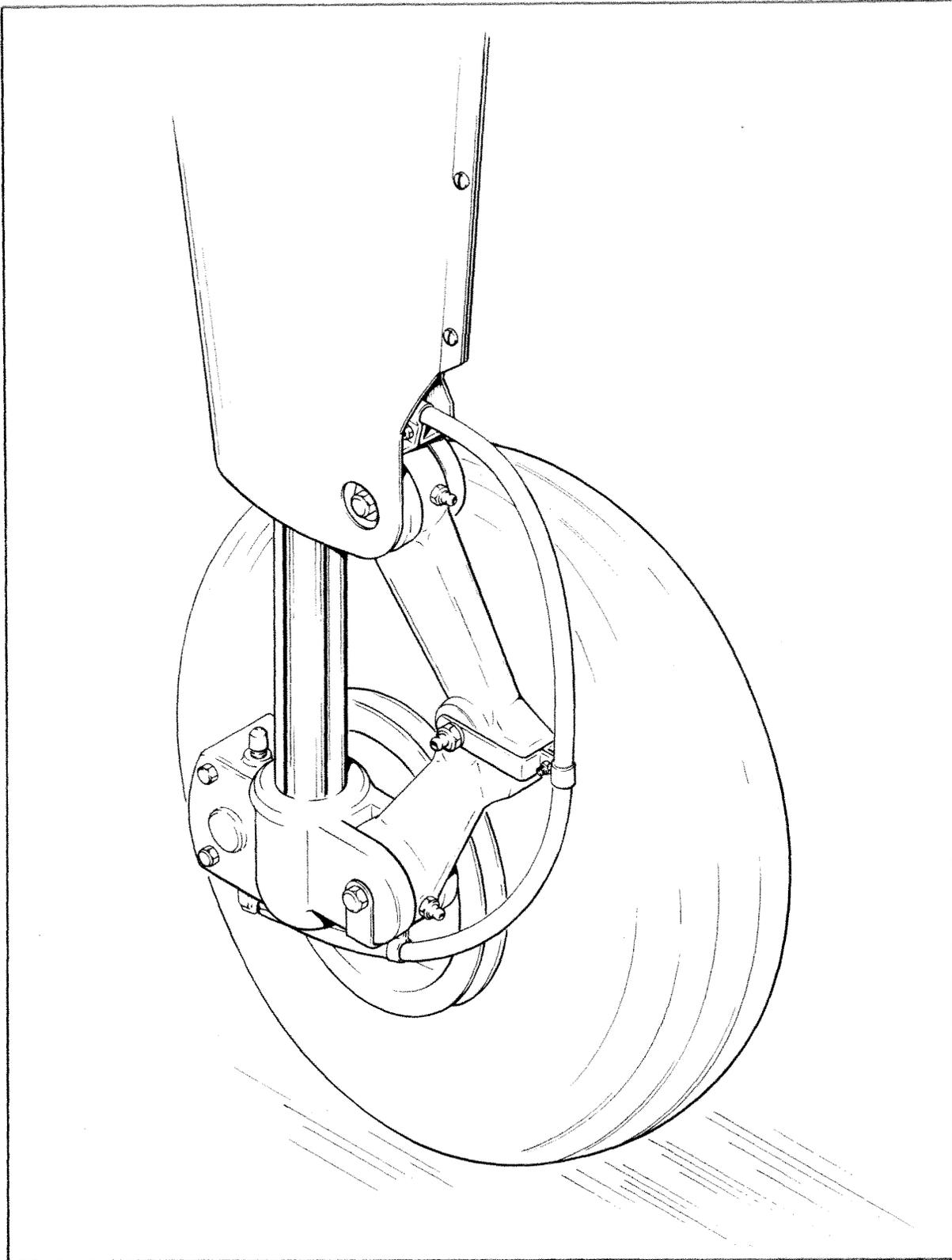
The wing airfoil section is a laminar flow type, NACA65₂-415 with the maximum thickness about 40% aft of the leading edge. This permits the main spar carry-through structure to be located under the rear seat, providing unobstructed cabin floor space ahead of the rear seat.

ENGINE AND PROPELLER

The Cherokee 180 is powered by a Lycoming O-360-A4A or O-360-A4M four cylinder, direct drive, horizontally opposed engine rated at 180 horsepower at 2700 rpm. It is furnished with a starter, a 60 ampere, 14 volt alternator, a shielded ignition, vacuum pump drive, a fuel pump, and a dry, automotive type carburetor air filter.

The exhaust system is of the cross-over type to reduce back pressure and improve performance. It is made entirely from stainless steel and is equipped with dual mufflers. A heater shroud around the mufflers is provided to supply heat for the cabin and windshield defrosting.

The Sensenich 76EM8S5-0-60 fixed-pitch **propeller** is made from a one-piece alloy forging.



Main Wheel Assembly

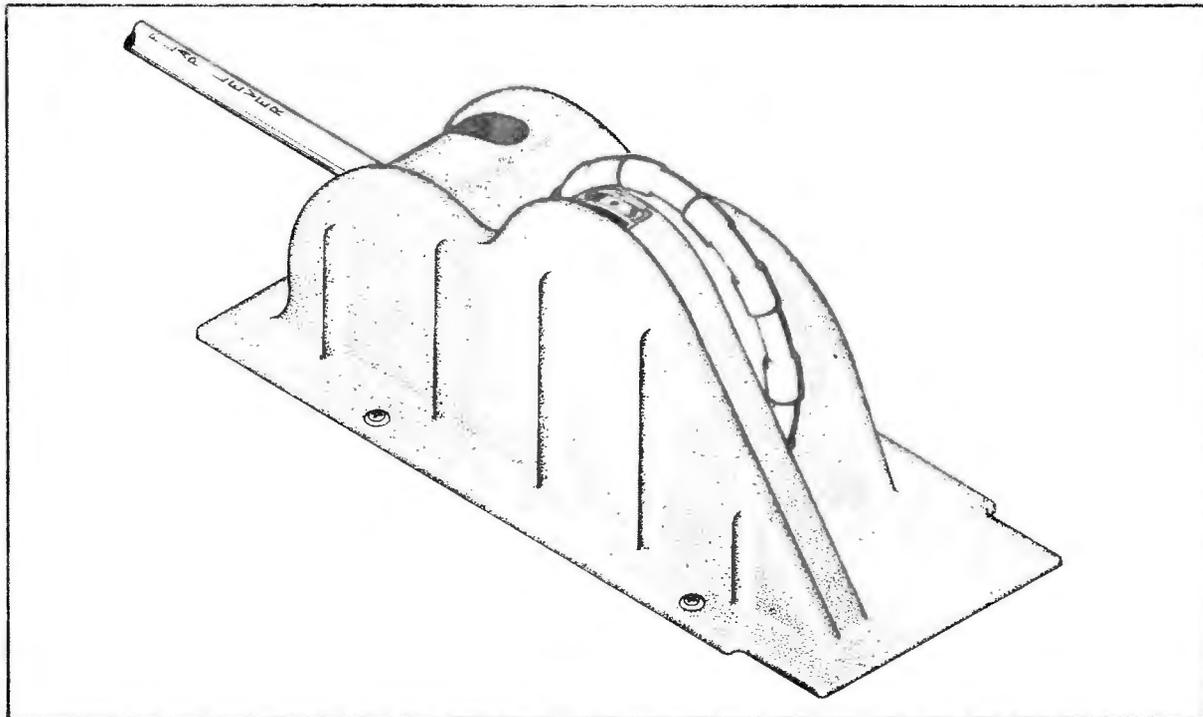
LANDING GEAR

The three **landing gears** use Cleveland 6.00 x 6 wheels, the main gear wheels being provided with brake drums and Cleveland single disc hydraulic brake assemblies. All three wheels use 6.00 x 6, four-ply rating, Type III tires with tubes.

The **nose gear** is steerable through a 30 degree arc either side of center by use of the rudder pedals and brakes. A spring device incorporated in the rudder pedal torque tube assembly aids in rudder centering and provides rudder trim. The nose gear steering mechanism also incorporates a bungee assembly to reduce steering effort and to dampen shocks and bumps during taxiing. A shimmy dampener is included in the nose gear.

The three struts are of the air-oil type, with a normal extension of 3.25 inches for the nose gear and 4.50 inches for the main gear.

The standard **brake system** for this Cherokee consists of dual toe brakes attached to the rudder pedals and a hand lever and master cylinder located below and behind the left center of the instrument sub-panel. The toe brakes and the hand brake have their own brake cylinders, but they share a common reservoir. The brake fluid reservoir is installed on the top left front face of the fire wall. The **parking brake** is incorporated in the master cylinder and is actuated by pulling back on the brake lever, depressing the knob attached to the left side of the handle, and releasing the brake lever. To release the parking brake, pull back on the brake lever to disengage the catch mechanism and allow the handle to swing forward.



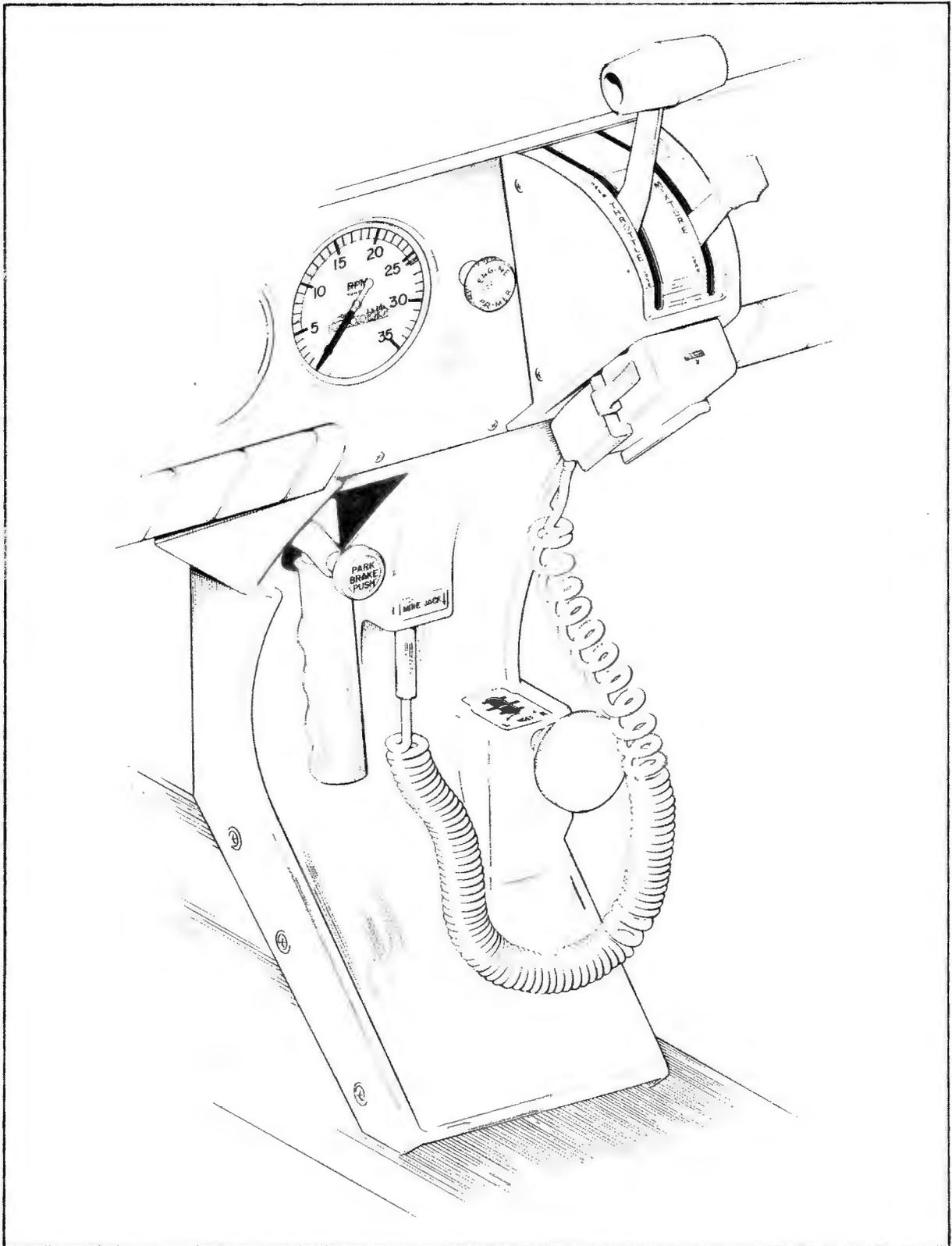
Console

FLIGHT CONTROLS

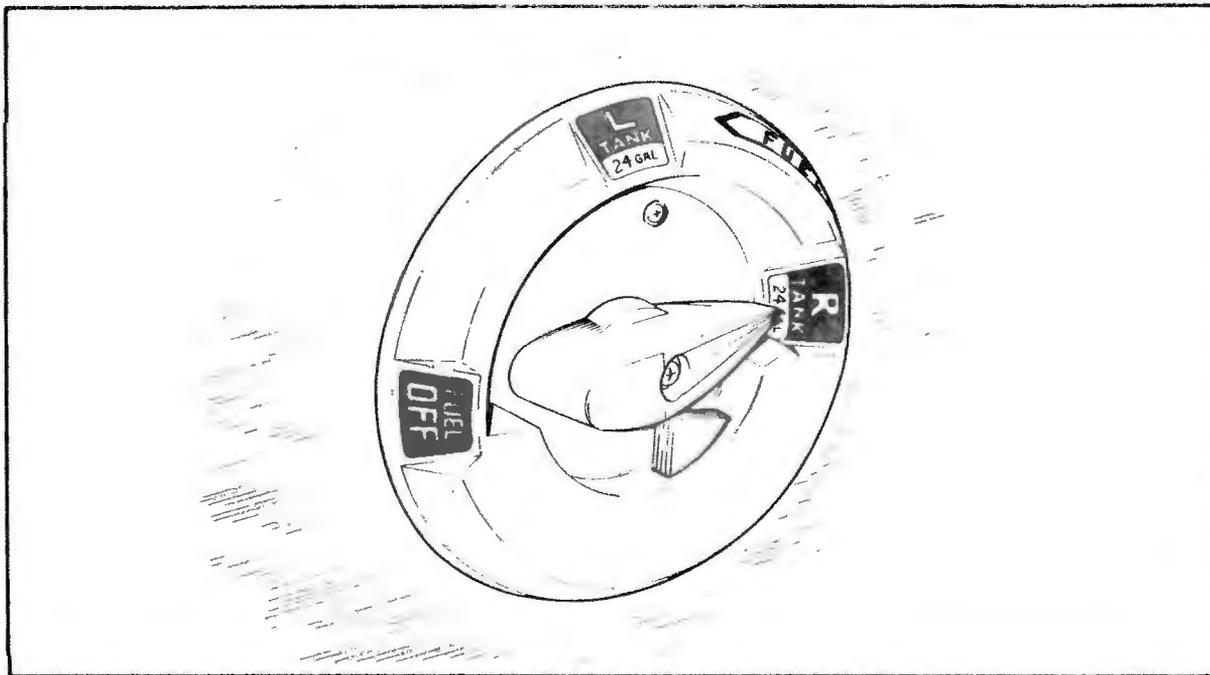
Dual controls are provided as standard equipment, with a cable system used between the controls and the surfaces. The horizontal tail (**stabilator**) is of the all-movable slab type with a **trim tab** mounted on the trailing edge of the stabilator to reduce the control system forces. This tab is actuated by a control wheel on the floor between the front seats.

The **stabilator** provides extra stability and controllability with less size, drag and weight than conventional tail surfaces. The **ailerons** are provided with a differential action which tends to reduce adverse yaw in turning maneuvers, and which also reduces the amount of coordination required in normal turns. A rudder trim adjustment is mounted on the right side of the pedestal below the throttle quadrant and permits directional trim as needed in flight.

The **flaps** are manually operated, balanced for light operating forces and spring-loaded to return to the up position. A past-center lock incorporated in the actuating linkage holds the flap when it is in the up position so that it may be used as a step on the right side. The flap will not support a step load except when in the full up position, so it must be completely retracted when used as a step. The flaps have three extended positions, 10, 25 and 40 degrees.



Throttle Quadrant and Console



Fuel Selector

FUEL SYSTEM

Fuel is stored in two twenty-five gallon (24 gallons usable) tanks which are secured to the leading edge structure of each wing by screws and nut plates. This allows easy removal for service or inspection.

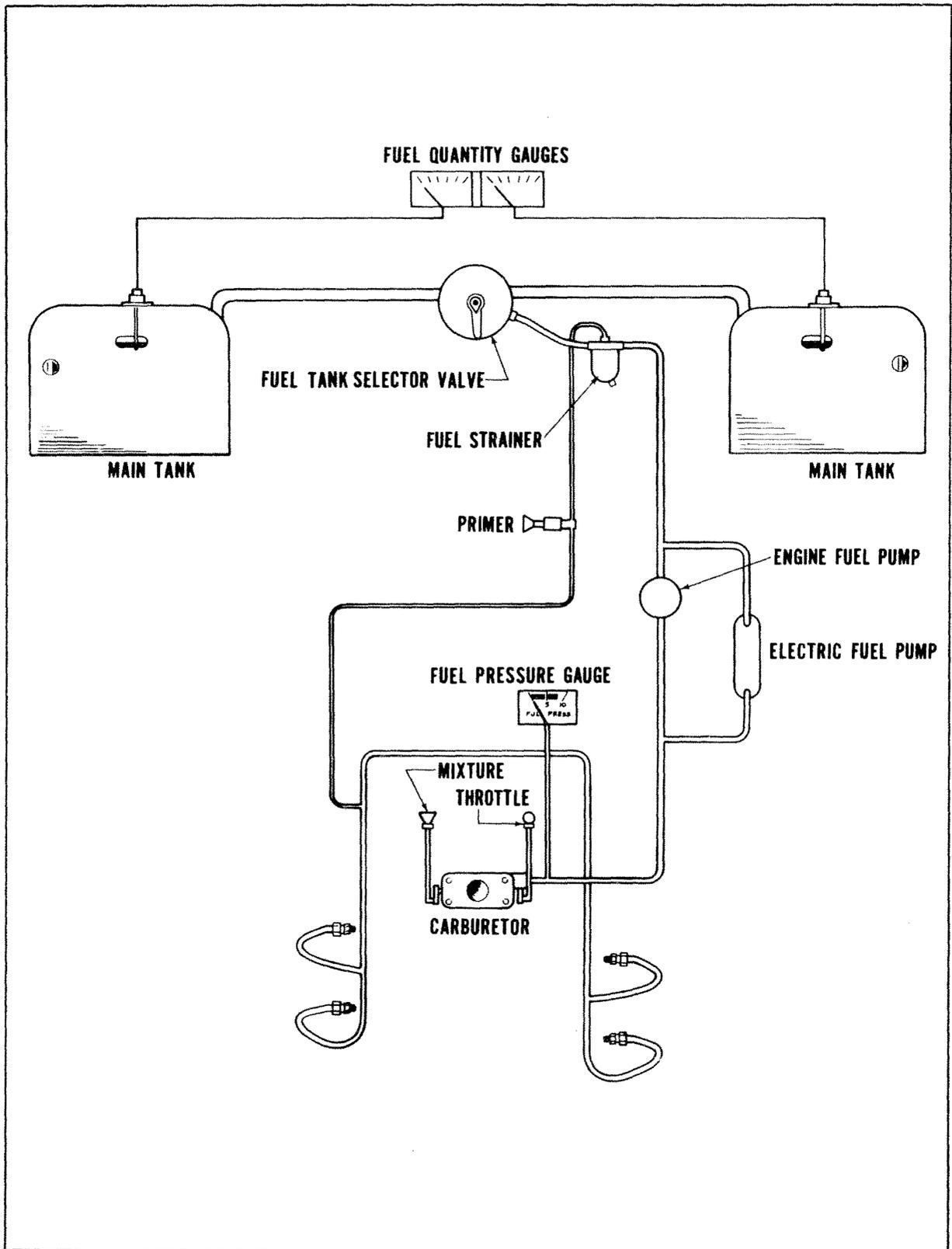
The **fuel selector control** is located on the left side-panel, forward of the pilot's seat. The button on the selector cover must be depressed and held while the handle is moved to the OFF position. The button releases automatically when the handle is moved back into the ON position.

An **auxiliary electric fuel pump** is provided in case of failure of the engine driven pump. The electric pump should be on for all takeoffs and landings, and when switching tanks. The pump switch is located in the switch panel above the throttle quadrant.

Each tank has an **individual quick drain** located at the bottom, inboard rear corner, and should be drained to check for water before each flight (a special container is furnished for this operation). The **fuel strainer**, which is also equipped with a quick drain, is located on the front lower left corner of the fire wall. This strainer should be drained regularly to check for water or sediment accumulation. To drain the lines from the tanks, the tank selector valve must be switched to each tank in turn, with the electric pump on, and the gascolator drain valve opened. (Refer to the Handling and Servicing Section for the complete procedure.)

Fuel quantity and pressure are indicated on gauges located in a cluster on the left side of the instrument panel.

An optional **engine priming system** is available to facilitate starting. The primer pump is located to the immediate left of the throttle quadrant.



Fuel System Schematic

ELECTRICAL SYSTEM

The electrical system includes a 14-volt, 60 amp alternator, a 12-volt battery, a voltage regulator, an overvoltage relay and a master switch relay. The battery is mounted in a stainless steel box immediately aft of the baggage compartment. The regulator and overvoltage relay are located on the forward left side of the fuselage behind the instrument panel.

Electrical switches are located on the right center instrument panel, and the circuit breakers are located on the lower right instrument panel. A rheostat switch on the left side of the switch panel controls the navigational lights and the radio lights. The similar switch on the right side controls and dims the panel lights.

The alternator system offers many advantages over the generator system both in operation and maintenance. The main advantage is full electrical power output at lower engine RPM. This is a great improvement for radio and electrical equipment operation. Since the alternator output is available at all times, the battery will be charging for a greater percentage of use. This will make cold weather starting easier.

Standard electrical accessories include a starter, electric fuel pump, stall warning indicator, cigar lighter, fuel gauge, ammeter, and annunciator panel*.

The annunciator panel includes alternator and low oil pressure indicator lights. When the optional gyro system is installed, the annunciator panel also includes a low vacuum indicator light. The annunciator panel lights are provided only as a warning to the pilot that a system may not be operating properly, and that he should check and monitor the applicable system gauge to determine when or if any necessary action is required.

Optional electrical accessories include navigation lights, anti-collision light, landing light, instrument lighting, and cabin dome light. Circuits will handle a full complement of communications and navigational equipment.

WARNING

Strobe lights should not be operating when flying through overcast and clouds since reflected light can produce spacial disorientation. Do not operate strobe lights in close proximity to ground, during takeoff and landing.

The words "**master switch**" used hereafter in this manual indicate both sides of the switch; battery side "BAT" and alternator side "ALT" are to be depressed simultaneously to OFF or ON as directed.

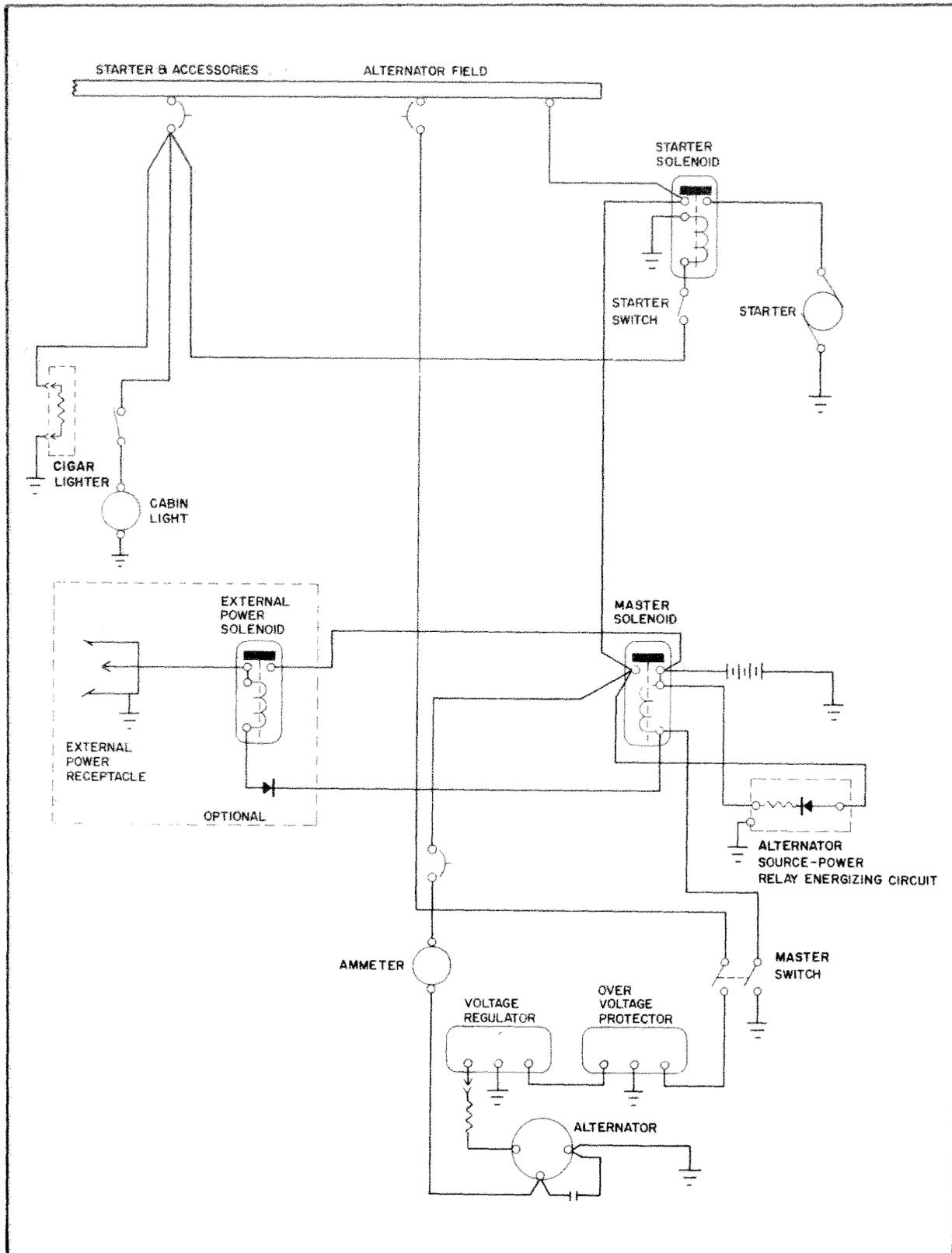
*Serial nos. 7505001 and up

Unlike previous generator systems, the ammeter does not indicate battery discharge; rather it displays in amperes the load placed on the alternator. With all electrical equipment off (except master switch) the ammeter will be indicating the amount of charging current demanded by the battery. As each item of electrical equipment is turned on, the current will increase to a total appearing on the ammeter. This total includes the battery. The maximum continuous load for night flight, with radios on, is about 30 amperes. This 30 ampere value, plus approximately two amperes for a fully charged battery, will appear continuously under these flight conditions. The amount of current shown on the ammeter will tell immediately if the alternator system is operating normally, as the amount of current shown should equal the total amperage drawn by the equipment which is operating.

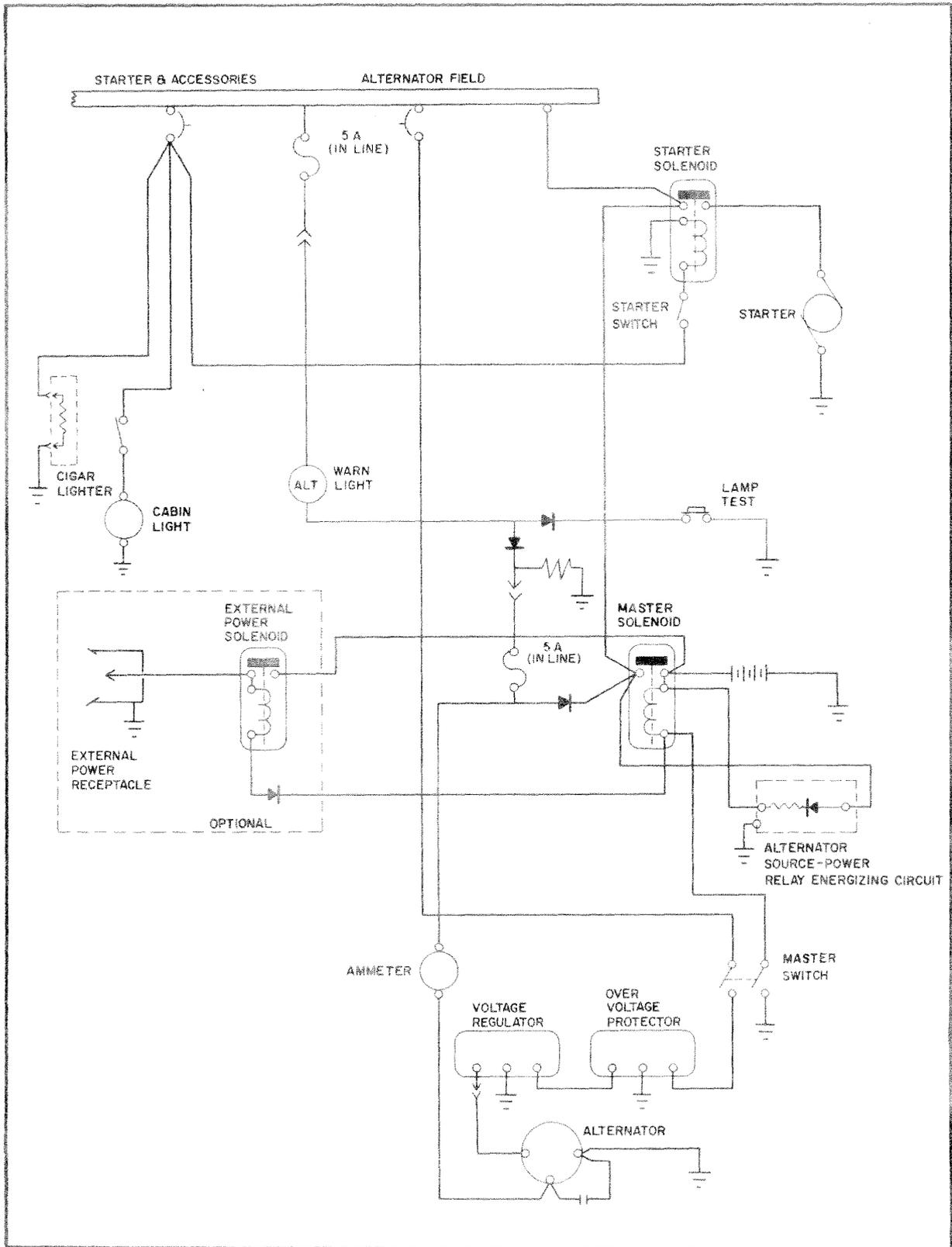
If no output is indicated on the ammeter during flight, reduce the electrical load by turning off all unnecessary electrical equipment. Check both 5 ampere field breaker and 60 ampere output breaker and reset if open. If neither circuit breaker is open, turn off the "ALT" switch for 1 second to reset the overvoltage relay. If ammeter continues to indicate no output, maintain minimum electrical load and terminate flight as soon as practical.

Maintenance on the alternator should prove to be a minor factor. Should service be required, contact the local Piper Dealer.

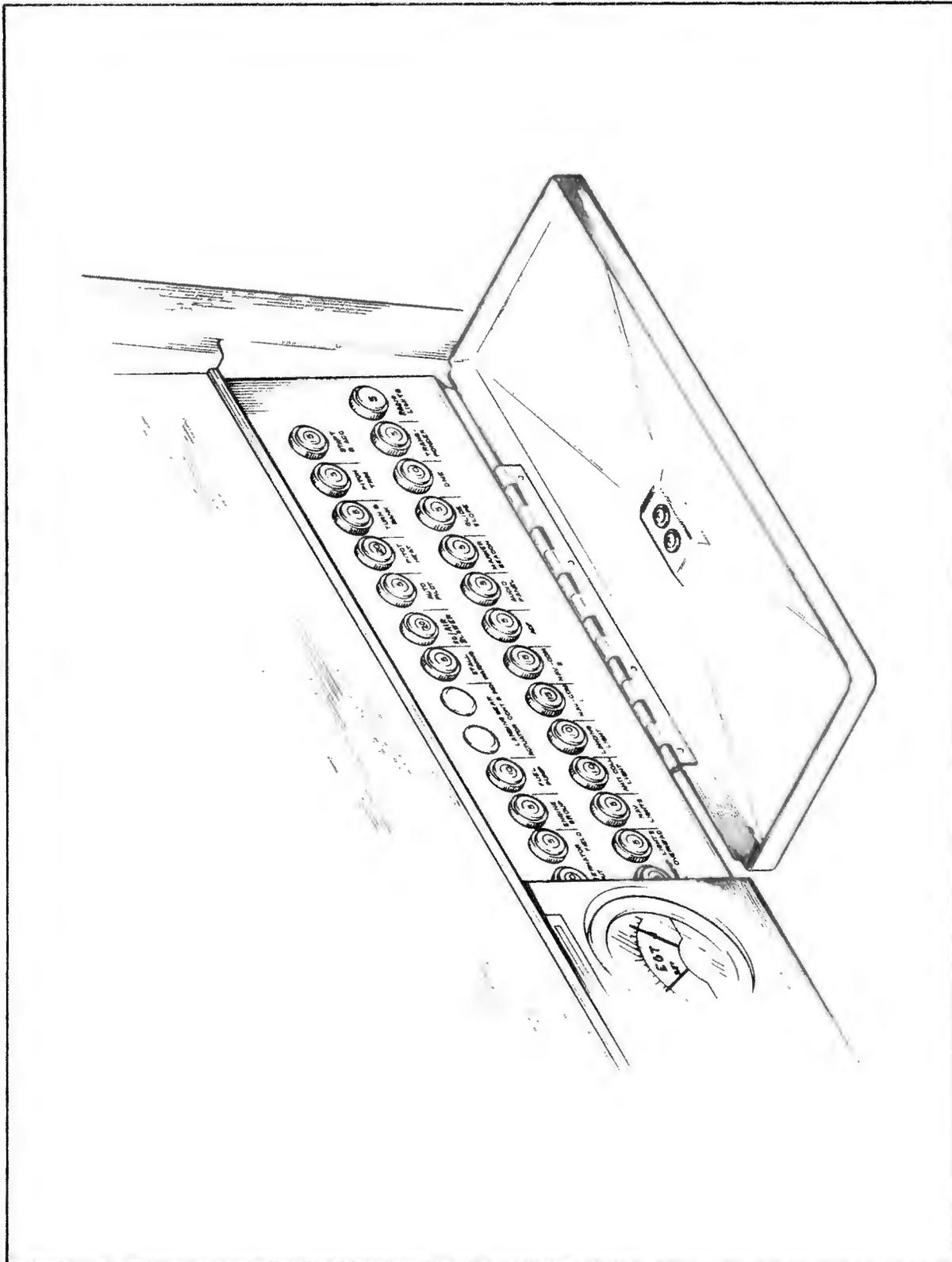
CHEROKEE ARCHER



Alternator and Starter Schematic (Ser. nos. 7405001 through 7405290)



Alternator and Starter Schematic (Ser. nos. 7505001 and up)



Circuit Breaker Panel

VACUUM SYSTEM

The vacuum system is designed to operate the air driven gyro instruments. This includes the directional and attitude gyros when installed. The system consists of an engine driven vacuum pump, a vacuum regulator, a filter and the necessary plumbing.

The vacuum pump is a dry type pump which eliminates the need for an air/oil separator and its plumbing. A shear drive protects the pump from damage. If the drive shears, the gyros will become inoperative.

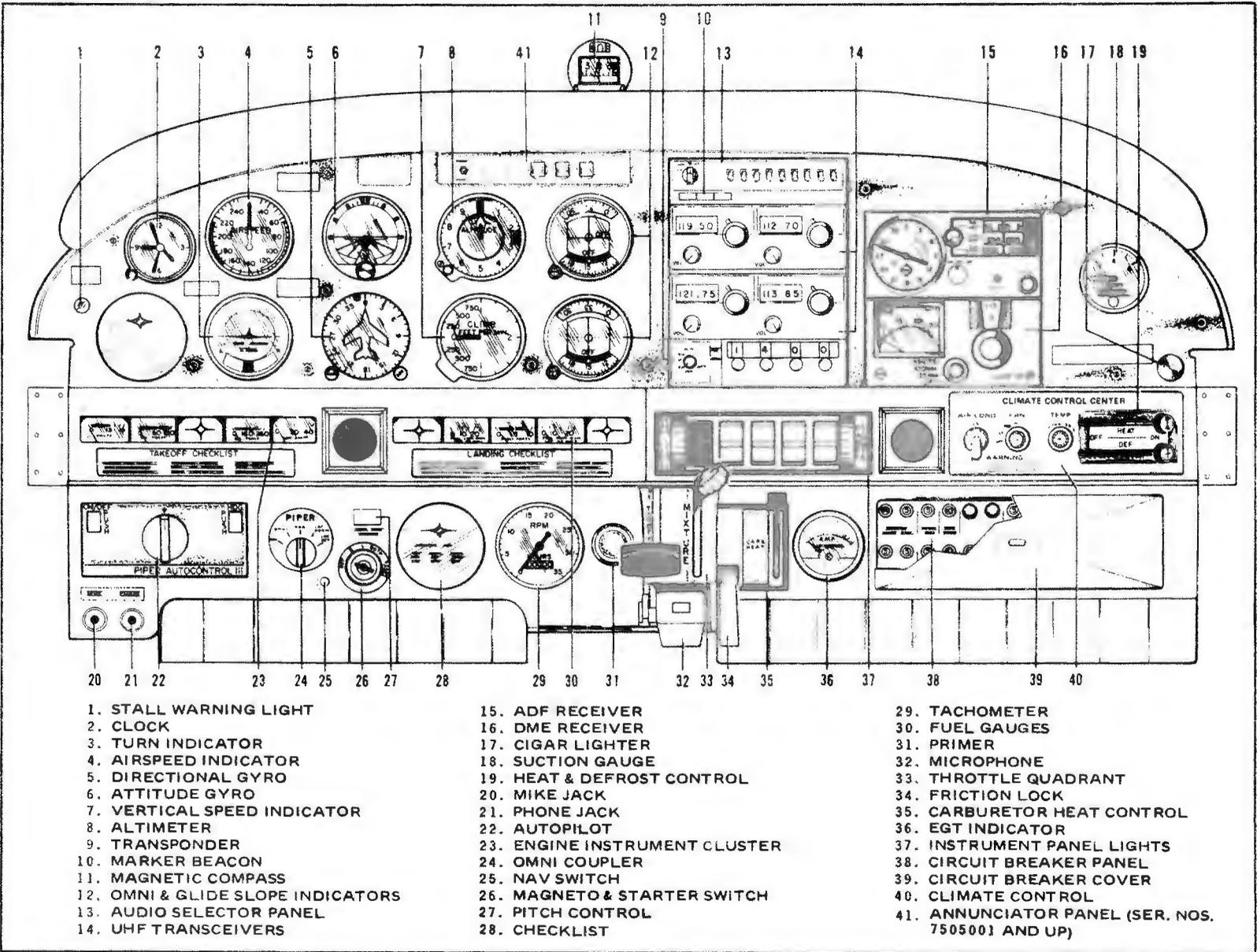
The vacuum gauge, mounted on the right instrument panel to the right of the radios, provides valuable information to the pilot about the operation of the vacuum system. A decrease in pressure in a system that has remained constant over an extended period may indicate a dirty filter, dirty screens, possibly a sticking vacuum regulator or leak in system (a low vacuum indicator light is provided in the annunciator panel*). Zero pressure would indicate a sheared pump drive, defective pump, possibly a defective gauge or collapsed line. In the event of any gauge variation from the norm, the pilot should have a mechanic check the system to prevent possible damage to the system components or eventual failure of the system.

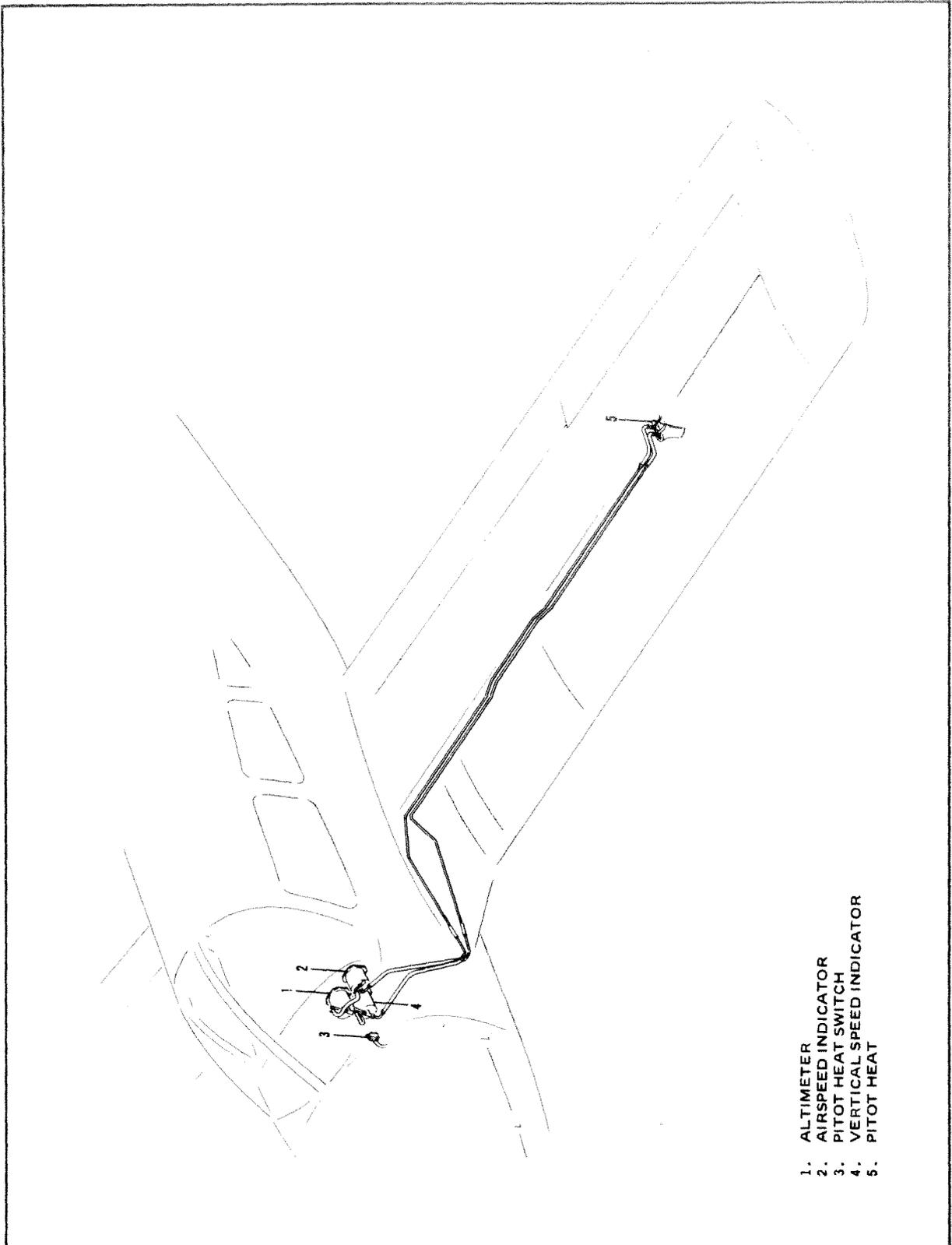
A vacuum regulator is provided in the system to protect the gyros. The valve is set so the normal vacuum reads $5.0 \pm .1$ inches of mercury, a setting which provides sufficient vacuum to operate all the gyros at their rated RPM. Higher settings will damage the gyros and with a low setting the gyros will be unreliable. The regulator is located behind the instrument panel and is accessible from below the instrument panel.

INSTRUMENT PANEL

The instrument panel of the Cherokee is designed to accommodate the customary advanced flight instruments and the normally required power plant instruments. The artificial horizon and directional gyro are vacuum operated through use of a vacuum pump installed on the engine, while the turn and bank instrument is electrically operated. A vacuum gauge is mounted on the far right side of the instrument panel. A natural separation of the flight group and power group is provided by the placement of the flight group in the upper instrument panel and the power group in the center and lower instrument panels. The radios and circuit breakers are on the right hand instrument panel. Extra circuits are provided for a complete line of optional radio equipment. An annunciator panel* is mounted in the upper instrument panel to warn the pilot of a possible malfunction in the alternator, oil pressure, or vacuum systems.

*Serial nos. 7505001 and up





Pitot - Static System

PITOT-STATIC SYSTEM

The system supplies both pitot and static pressure for the airspeed indicator, altimeter and vertical speed indicator (when installed).

Pitot and static pressure are picked up by the pitot head on the bottom of the left wing. An optional heated pitot head, which alleviates problems with icing or heavy rain, is available. The switch for pitot heat is located on the switch panel above the throttle quadrant.

To prevent bugs and water from entering the pitot and static pressure holes, when the airplane is parked, a cover should be placed over the pitot head. A partially or completely blocked pitot head will give erratic or zero readings on the instruments.

NOTE

During the preflight, check to make sure the pitot cover is removed.

HEATING AND VENTILATING SYSTEM

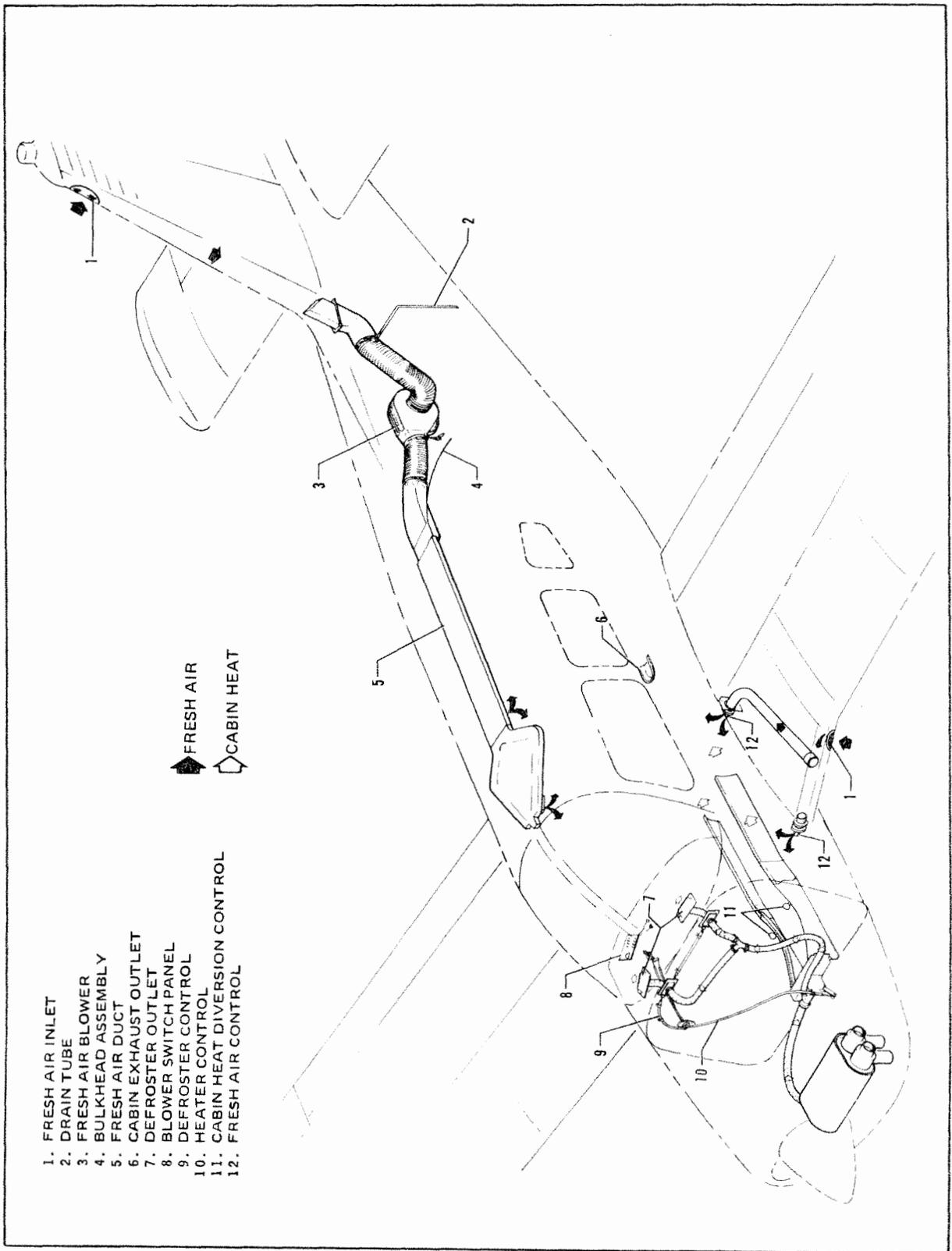
Heat for the cabin interior and the defroster system is provided by a **heater muff** attached to the exhaust system. The amount of heat desired can be regulated with the controls located on the far right side of the instrument panel.

The air flow can be regulated between the front and rear seats by levers located on top of the heat ducts next to the console.

Fresh **air inlets** are located in the leading edge of the wing at the intersection of the tapered and straight sections. A large adjustable outlet is located on the side of the cabin near the floor at each seat location; overhead air outlets are offered as optional equipment. Air is exhausted through an outlet under the rear seat. A **cabin air blower**, incorporated in the ventilating system, is also available as optional equipment. An optional overhead ventilating system with a cabin air blower is available on models without air conditioning. This blower is operated by a "FAN" switch with 4 positions - "OFF," "LOW," "MED," or "HIGH."

CAUTION

When cabin heat is operated, heat duct surface becomes hot. This could result in burns if arms or legs are placed too close to heat duct outlets or surface.



Heating and Ventilating System

CABIN FEATURES

For ease of entry and exit and pilot-passenger comfort, the front seats are adjustable fore and aft. The back of the right front seat contains two latches, an outboard latch which allows the seat to be moved forward or aft for ease of entry, and an inboard latch which allows the seat back to be tilted forward to allow easy entry to the rear seats. The rear seats are easily removed to provide room for bulky items. Some rear seat installations incorporate leg retainers with latching mechanisms which must be released before the rear seats can be removed. Releasing the retainers is easily accomplished by turning the latching mechanisms 90° with a coin or screwdriver. Armrests are also provided for the front seats. All seats are available with optional headrests and optional vertical adjustment may be added to the front seats.

The cabin interior includes a pilot storm window, two sun visors, ash trays, two map pockets, and pockets on the backs of each front seat.

A single strap shoulder harness controlled by an inertia reel is standard equipment for the front seats, and is offered as an option for the rear seats. The shoulder strap is routed over the shoulder adjacent to the windows and attached to the lap belt in the general area of the person's inboard hip.

A check of the inertia reel mechanism is made by pulling sharply on the strap. The reel will lock in place under this test and prevent the strap from extending. Under normal movement the strap will extend and retract as required.

BAGGAGE AREA

A 24 cubic foot baggage area, located behind the rear seats, is accessible either from the cabin or through a large 20 x 22 inch outside baggage door on the right side of the aircraft. Maximum capacity is 200 pounds. Tie-down straps are provided and should be used at all times.

NOTE

It is the pilot's responsibility to be sure when the baggage is loaded that the aircraft C.G. falls within the allowable C.G. Range. (See Weight and Balance Section.)

STALL WARNING

An approaching stall is indicated by a stall warning light located on the left side of the instrument panel. The red light illuminates automatically between five and ten miles per hour above stall speed.

FINISH

All exterior surfaces are primed with etching primer and finished with a durable acrylic lacquer in a variety of tasteful colors to suit individual owners.

AIR CONDITIONING*

The air conditioning system is a recirculating air system. The major items include; evaporator, condenser, compressor, blower, switches and temperature controls.

The evaporator is located behind the left rear side of the baggage compartment. This cools the air that is used for air conditioning.

The condenser is mounted on a retractable scoop located on the bottom of the fuselage and to the rear of the baggage compartment area. The scoop extends when the air conditioner is "ON" and retracts to a flush position when the system is "OFF."

The compressor is mounted on the forward right underside of the engine. It has an electric clutch which automatically engages or disengages the compressor to the belt drive system of the compressor.

An electrical blower is mounted on the aft side of the rear cabin panel. Air from the baggage area is drawn through the evaporator by the blower and distributed through an overhead duct to individual outlets located adjacent to each occupant.

The switches and temperature control are located on the lower right side of the instrument panel in the climate control center panel. The temperature control regulates the desired temperature of the cabin. Turn the control clockwise for increased cooling, counterclockwise for decreased cooling.

Located inboard of the temperature control is the fan speed switch and the air conditioning "ON-OFF" switch. The fan can be operated independently of the air conditioning. However, it must be on for air conditioner operation. Turning either switch off will disengage the compressor clutch and retract the condenser door. Cooling air should be felt within one minute after the air conditioner is turned on.

NOTE

If the system is not operating in 5 minutes, turn the system "OFF" until the fault is corrected.

The "FAN" switch allows operation of the fan with the air conditioner turned "OFF" to aid cabin air circulation if desired. A "LOW," "MED" or "HIGH" flow of air can be selected to the air conditioner outlets located in the overhead duct. The outlets can be adjusted or turned off by each occupant to regulate individual cooling effect.

The "DOOR OPEN" indicator light is located to the left of the radio stack in front of the pilot. The light illuminates whenever the condenser door is open and remains on until the door is closed.

A circuit breaker located on the circuit breaker panel protects the air conditioning electrical system.

*Optional equipment

CHEROKEE ARCHER

Whenever the throttle is in the full throttle position, it actuates a micro switch which disengages the compressor and retracts the scoop. This is done to obtain maximum power and maximum rate of climb. The fan continues to operate and the air will remain cool for approximately one minute. When the throttle is retarded approximately 1/4 inch, the clutch will engage and the scoop will extend, again supplying cool, dry air.

PIPER EXTERNAL POWER*

An optional starting installation known as Piper External Power (PEP) is accessible through a receptacle located on the right side of the fuselage aft of the wing. An external battery can be connected to the socket, thus allowing the operator to crank the engine without having to gain access to the airplane's battery.

*Optional equipment

AIRPLANE FLIGHT MANUAL

FOR

CHEROKEE ARCHER

APPLICABLE TO SERIAL NUMBERS 28-7405001 THROUGH 28-7505259

WARNING

EXTREME CARE MUST BE EXERCISED TO LIMIT THE USE OF THIS MANUAL TO APPLICABLE AIRCRAFT. THIS MANUAL REVISED AS INDICATED BELOW OR SUBSEQUENTLY REVISED IS VALID FOR USE WITH THE AIRPLANE IDENTIFIED BELOW WHEN APPROVED BY PIPER AIRCRAFT CORPORATION. SUBSEQUENT REVISIONS SUPPLIED BY PIPER AIRCRAFT CORPORATION MUST BE PROPERLY INSERTED.

MODEL PA-28-180

AIRCRAFT SERIAL NO. 28-7505222 REGISTRATION NO. 1185X

AIRPLANE FLIGHT MANUAL, REPORT NUMBER VB-558 REVISION _____

PIPER AIRCRAFT CORPORATION
APPROVAL SIGNATURE AND STAMP _____

NOTE

THIS MANUAL MUST BE KEPT IN THE AIRPLANE AT ALL TIMES

FAA APPROVED BY:

H. W. Barnhouse

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DATE OF APPROVAL: MAY 14, 1973

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AIRPLANE FLIGHT MANUAL

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AIRPLANE FLIGHT MANUAL LOG OF REVISIONS

Revision	Revised Pages	Description and Revision	FAA Approved Date
1	Title	Added PAC Approval Form. (NOTE: AIRCRAFT DELIVERED WITH MANUALS PRIOR TO THIS REVISION DO NOT REQUIRE THIS REVISION.)	<i>D. H. Trompler</i> D. H. Trompler May 30, 1974
2	3-i 3-11 3-19, 3-20, 3-21, 3-22	Added item F. Installation of Piper Auto-Control IIIB to Supplements. Added item F. Installation of Piper Auto-Control IIIB. Added Pages (AutoControl IIIB Supplement).	<i>D. H. Trompler</i> D. H. Trompler June 17, 1974
3	3-i 3-1 3-11 3-13 3-14 3-15 3-16 3-17 3-19 3-22	Changed Section IV title from Supplements to Optional Equipment; deleted (With Pitch Trim Switch) from item A.; deleted items B. and C.; revised existing item letters; added AutoControl III to new item D. Added O-360-A4M engine designation and ser. nos. Changed Section IV title from Supplements to Optional Equipment; revised NOTE; deleted items B. and C.; revised existing item letters; added AutoControl III to new item D. Deleted (With Pitch Trim Switch) from title. Deleted item B. AutoFlite Installation. Deleted item C. AutoControl III Installation. Changed item letter (D. to B.); revised item 2. b. (3). Changed item letter (E. to C.). Changed item letter (F. to D.); added Auto-Control III to title. Deleted IIIB designation from items 2. c. (1) and (2).	<i>Ward Evans</i> Ward Evans Jan. 20, 1975
4	Title 3-1	Added Applicable Serial Nos. Removed Serial Nos. from item A.	<i>Ward Evans</i> Ward Evans Nov. 21, 1975

AIRPLANE FLIGHT MANUAL LOG OF REVISIONS

Revision	Revised Pages	Description and Revision	FAA Approved Date
5	3-1	Revised item B.	<i>Ward Evans</i> Ward Evans March 16, 1984

SECTION I
LIMITATIONS

The following limitations must be observed in the operation of this airplane:

- A. **ENGINE**
Lycoming O-360-A4A or O-360-A4M with carburetor setting IO-3878

ENGINE LIMITS
For all operations 2700 RPM, 180 HP

- B. **FUEL (AVGAS ONLY)**
100/130 minimum octane aviation fuel
- C. **PROPELLER**
Sensenich M76EMMS or 76EM8S5. Maximum diameter 76 inches, minimum diameter 76 inches. Static RPM at maximum permissible throttle setting. Not over 2425, not under 2325. No additional tolerance permitted.
- D. **POWER INSTRUMENTS**

OIL TEMPERATURE

Green Arc (Normal Operating Range)	75° F to 245° F
Red Line (Maximum)	245° F

OIL PRESSURE

Green Arc (Normal Operating Range)	60 PSI to 90 PSI
Yellow Arc (Caution Range)	25 PSI to 60 PSI
Red Line (Minimum)	25 PSI
Red Line (Maximum)	90 PSI

FUEL PRESSURE

Green Arc (Normal Operating Range)	.5 PSI to 8 PSI
Red Line (Minimum)	.5 PSI
Red Line (Maximum)	8 PSI

TACHOMETER

Green Arc (Normal Operating Range)	500 to 2700 RPM
Red Line (Maximum Continuous Power)	2700 RPM

CHEROKEE ARCHER

E. AIRSPEED LIMITATIONS AND AIRSPEED INSTRUMENT MARKINGS

NEVER EXCEED	171 MPH
MAXIMUM STRUCTURAL CRUISE	140 MPH
MANEUVERING	127 MPH
FLAPS EXTENDED	115 MPH
MAXIMUM POSITIVE LOAD FACTOR	(Normal Category) 3.8
MAXIMUM POSITIVE LOAD FACTOR	(Utility Category) 4.4
MAXIMUM NEGATIVE LOAD FACTOR	No inverted maneuvers approved

AIRSPEED INSTRUMENT MARKINGS

Red Radial Line (Never Exceed)	171 MPH (148 KTS)
Yellow Arc (Caution Range) (Smooth Air Only)	140 MPH to 171 MPH (121 KTS to 148 KTS)
Green Arc (Normal Operating Range)	68 MPH to 140 MPH (59 KTS to 121 KTS)
White Arc (Flaps Down Range)	61 MPH to 115 MPH (53 KTS to 100 KTS)

F. MAXIMUM WEIGHT

Normal Category	2450 LBS
Utility Category	1950 LBS

G. BAGGAGE CAPACITY

200 LBS

H. C. G. RANGE

The datum used is 78.4 inches ahead of wing leading edge at the intersection of the straight and tapered section.

1. Normal Category

<u>Weight (Pounds)</u>	<u>Forward Limit (In. Aft of Datum)</u>	<u>Rearward Limit (In. Aft of Datum)</u>
2450	87.4	93.0
2050	82.0	93.0

2. Utility Category

<u>Weight (Pounds)</u>	<u>Forward Limit (In. Aft of Datum)</u>	<u>Rearward Limit (In. Aft of Datum)</u>
1950	82.0	86.5

Straight line variation between points given.

NOTE

It is the responsibility of the airplane owner and the pilot to insure that the airplane is properly loaded. See Weight and Balance Section for proper loading instructions.

I. MANEUVERS

1. Normal Category - All acrobatic maneuvers including spins prohibited.
2. Utility Category - Approved maneuvers for Utility Category only.

	Entry Speed
Steep Turns	127 MPH
Lazy Eights	127 MPH
Chandelles	127 MPH

J. PLACARDS

In full view of the pilot:

“THIS AIRPLANE MUST BE OPERATED AS A NORMAL OR UTILITY CATEGORY AIRPLANE IN COMPLIANCE WITH THE OPERATING LIMITATIONS STATED IN THE FORM OF PLACARDS, MARKINGS AND MANUALS.

ALL MARKINGS AND PLACARDS ON THIS AIRPLANE APPLY TO ITS OPERATION AS A UTILITY CATEGORY AIRPLANE. FOR NORMAL AND UTILITY CATEGORY OPERATIONS, REFER TO THE AIRPLANE FLIGHT MANUAL.

NO ACROBATIC MANEUVERS ARE APPROVED FOR NORMAL CATEGORY OPERATIONS. SPINS ARE PROHIBITED FOR BOTH NORMAL AND UTILITY CATEGORIES.”

In full view of the pilot, the following takeoff and landing check lists will be installed:

TAKEOFF CHECK LIST

Fuel on proper tank	Mixture set	Fasten belts/harness
Electric fuel pump on	Seat backs erect	Trim tab - set
Engine gauges checked		Controls - free
Flaps - set		Door - latched
Carb heat off		Air Conditioner - off

LANDING CHECK LIST

Fuel on proper tank		Flaps - set (115 mph)
Mixture rich	Seat backs erect	Fasten belts/harness
Electric fuel pump on		Air Conditioner - off

The “AIR COND OFF” item in the above takeoff and landing check lists is mandatory for air conditioned aircraft only.

CHEROKEE ARCHER

In full view of the pilot, in the area of the air conditioner control panel when the air conditioner is installed:

“WARNING - AIR CONDITIONER MUST BE OFF TO INSURE NORMAL TAKEOFF CLIMB PERFORMANCE.”

Adjacent to upper door latch:

“ENGAGE LATCH BEFORE FLIGHT.”

On inside of the baggage compartment door:

“BAGGAGE MAXIMUM 200 LBS”
“UTILITY CATEGORY OPERATION - NO BAGGAGE OR AFT PASSENGERS ALLOWED. NORMAL CATEGORY OPERATION - SEE AIRPLANE FLIGHT MANUAL WEIGHT AND BALANCE SECTION FOR BAGGAGE AND AFT PASSENGER LIMITATIONS.”

In full view of the pilot:

“ROUGH AIR OR MANEUVERING SPEED - 127 MPH.”

“UTILITY CATEGORY OPERATION - NO AFT PASSENGERS ALLOWED.”

On the instrument panel in full view of the pilot when the oil cooler winterization kit is installed:

“OIL COOLER WINTERIZATION PLATE TO BE REMOVED WHEN AMBIENT TEMPERATURE EXCEEDS 50°F.”

On the instrument panel in full view of the pilot when the AutoFlite is installed:

“FOR HEADING CHANGES: PRESS DISENGAGE SWITCH ON CONTROL WHEEL. CHANGE HEADING, RELEASE DISENGAGE SWITCH.”

In full view of the pilot:

“UTILITY CATEGORY ONLY.”

ACROBATIC MANEUVERS ARE LIMITED TO THE FOLLOWING:

	ENTRY SPEED
STEEP TURNS	127 MPH
LAZY EIGHTS	127 MPH
CHANDELLES	127 MPH

On the instrument panel in full view of the pilot when the AutoFlite II is installed:

“TURN AUTOFLITE ON. ADJUST TRIM KNOB FOR MINIMUM HEADING CHANGE. FOR HEADING CHANGE, PRESS DISENGAGE SWITCH ON CONTROL WHEEL, CHANGE HEADING, RELEASE SWITCH. ROTATE TURN KNOB FOR TURN COMMANDS. PUSH TURN KNOB IN TO ENGAGE TRACKER. PUSH TRIM KNOB IN FOR HI SENSITIVITY. LIMITATIONS: AUTOFLITE OFF FOR TAKEOFF AND LANDING.”

On the instrument panel in full view of the pilot when the supplementary white strobe lights are installed:

“WARNING - TURN OFF STROBE LIGHTS WHEN TAXIING IN VICINITY OF OTHER AIRCRAFT, OR DURING FLIGHT THROUGH CLOUD, FOG OR HAZE.”

K. AIR CONDITIONED AIRPLANES

Air Conditioner must be off for takeoff and landing.

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SECTION II
PROCEDURES

1. The stall warning system is inoperative with the master switch off.
2. Electric fuel pump must be on for both landing and takeoff.
3. The PA-28-180 airplane is approved under FAA Regulation CAR 3 which prohibits intentional spins for both normal and utility category operation. The following information is noteworthy:
 - a. The stall characteristics of the PA-28-180 are normal with the nose pitching down moderately following the stall, occasionally with a moderate roll which can be corrected by normal use of ailerons and rudder against the roll.
 - b. Prolonged use of full rudder during stall practice may result in a rapid roll followed by a spin and should be avoided. Recovery from an incipient spin may be effected in less than one additional turn by use of opposite rudder followed by full forward control wheel.
 - c. In the event that a fully developed spin is inadvertently experienced, recovery is best made by using full opposite rudder followed by full forward wheel and full opposite aileron. The control positions against the spin should be maintained during the entire recover, which may require several turns and a substantial loss of altitude if the airplane is loaded heavily with a rearward center of gravity.
4. Except as noted above, all operating procedures for this airplane are normal.
5. Air Conditioned Models only: Warning - The air conditioner must be off to insure normal takeoff performance.

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SECTION III
PERFORMANCE

The following performance figures were obtained during FAA type tests and may be realized under conditions indicated with the airplane and engine in good condition and with average piloting technique. All performance is given for 2450 pounds.

Loss of altitude during stalls varied from 100 to 250 feet, depending on configuration and power.

Stalling speeds, in mph, power off, versus angle of bank (Calibrated Airspeed):

Angle of Bank	0°	20°	40°	50°	60°
Flaps Up	68	70	78	85	96
Flaps Down	61	-	-	-	-

Air Conditioned Models only:

When the full throttle position is not used or in the event of a malfunction which causes the compressor to operate and the condenser door to remain extended, a decrease in rate of climb of as much as 100 fpm can be expected at all altitudes.

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SECTION IV
OPTIONAL EQUIPMENT

NOTE

THE INFORMATION CONTAINED IN THIS SECTION
APPLIES WHEN THE RELATED EQUIPMENT IS INSTALLED
IN THE AIRCRAFT.

- A. Electric Pitch Trim Installation
- B. AutoFlite II Installation
- C. Air Conditioner Installation
- D. Installation of Piper AutoControl III and/or AutoControl IIIB

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I A. ELECTRIC PITCH TRIM INSTALLATION

The following emergency information applies in case of electric pitch trim malfunction:

1. In case of malfunction, disengage electric pitch trim by pushing pitch trim switch on instrument panel to OFF position.
2. In an emergency, electric pitch trim may be overpowered using manual pitch trim.
3. In cruise configuration, malfunction results in 10° pitch change and 200 ft altitude variation.
4. In approach configuration, a malfunction can result in a 5° pitch change and 50 ft altitude loss.

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C. AIR CONDITIONER INSTALLATION

Prior to takeoff, the air conditioner should be checked for proper operation as follows:

1. Check aircraft master switch on.
2. Turn the air conditioner control switch to "ON" and the fan switch to one of the operating positions - the "AIR COND DOOR OPEN" warning light will turn on, thereby indicating proper air conditioner condenser door actuation.
3. Turn the air conditioner control switch to OFF - the "AIR COND DOOR OPEN" warning light will go out, thereby indicating the air conditioner condenser door is in the up position.
4. If the "AIR COND DOOR OPEN" light does not respond as specified above, an air conditioner system or indicator bulb malfunction is indicated and further investigation should be conducted prior to flight.

The above operational check may be performed during flight if an in flight failure is suspected.

WARNING

The air conditioner must be off to insure normal takeoff performance.

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I B. AUTOFLITE II INSTALLATION

1. LIMITATIONS

AutoFlite off for takeoff and landing.
AutoFlite use prohibited above 160 mph CAS.

2. PROCEDURES

a. Normal Operation

Refer to Manufacturer's Operation Manual.

b. Emergency Operation

- (1) In case of malfunction, PRESS disconnect switch on pilot's control wheel.
- (2) Rocker switch on instrument panel - OFF.
- (3) Unit may be overpowered manually.
- (4) In climb, cruise or descending flight an autopilot runaway, with a 3 second delay could result in 60° bank, and 190 ft altitude loss.
- (5) In approach configuration an autopilot runaway, with a 1 second delay results in 15° bank and 40 ft altitude loss.

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D. INSTALLATION OF PIPER AUTOCONTROL III AND/OR AUTOCONTROL IIIB

1. LIMITATIONS

- a. Autopilot OFF during takeoff and landing.
- b. Autopilot use prohibited above 160 MPH CAS.

2. PROCEDURES

a. PREFLIGHT

(1) Roll Section

- (a) Place Radio Coupler in "Heading" mode and place A/P ON/OFF switch in the "ON" position to engage roll section. Rotate roll command knob Left and Right and observe control wheel describes a corresponding Left and Right turn, then center knob.
- (b) Set proper D.G. Heading on D.G. and turn Heading Indice to aircraft heading. Engage "Heading" mode switch and rotate Heading Indice right and left. Aircraft control wheel should turn same direction as Indice. While D.G. indice is set for a left turn, grasp control wheel and override the servo to the right. Repeat in opposite direction for right turn.
- (c) If VOR signal available check Omni mode on Radio Coupler by swinging Omni needle left and right slowly. Observe that control wheel rotates in direction of needle movement.
- (d) Disengage by placing the A/P ON/OFF switch to the "OFF" position.

b. IN-FLIGHT

- (1) Trim airplane (ball centered).
- (2) Check air pressure or vacuum to ascertain that the Directional Gyro and Attitude Gyro are receiving sufficient air.
- (3) Roll Section
 - (a) To engage, center Roll Command Knob, place the A/P ON/OFF switch to the "ON" position. To turn rotate roll command knob in desired direction. (Maximum angle of bank should not exceed 30°.)
 - (b) For heading mode, set Directional Gyro with Magnetic Compass. Push directional gyro HDG knob in, rotate to aircraft heading. Place the console HDG ON/OFF switch to the "ON" position. To select a new aircraft heading, push D.G. heading knob IN and rotate, in desired direction of turn, to the desired heading.

NOTE

In HDG mode the maximum bank angles are limited to approximately 20° and single command, heading changes should be limited to 150°. (HDG Indice not more than 150° from actual aircraft heading.)

(4) VOR

(a) To Intercept:

1. Using OMNI Bearing Selector, dial desired course, inbound or outbound.
2. Set identical heading on Course Selector D.G.
3. After aircraft has stabilized, position coupler mode selector knob to OMNI mode. As aircraft nears selected radial, interception and crosswind correction will be automatically accomplished without further switching.

NOTE

If aircraft position is less than 45° from selected radial, aircraft will intercept before station. If position is more than 45°, interception will occur after station passage. As the aircraft nears the OMNI station, (1/2 mile) the zone of confusion will direct an "S" turn in alternate directions as the OMNI indicator needle swings. This alternate banking limited to the standard D.G. bank angle, is an indication of station passage.

(b) To select new course:

1. To select a new course or radial, rotate the HDG indice to the desired HDG (match course).
2. Rotate OBS to the new course. Aircraft will automatically turn to the intercept heading for the new course.

(c) To change stations:

1. If same course is desired, merely tune receiver to new station frequency.
2. If different course is desired, position coupler mode selector to HDG mode. Dial course selector D.G. to new course. Dial OBS to new course and position coupler mode selector to OMNI mode.

(5) VOR Approach

Track inbound to station as described in VOR navigation section.

After station passage:

- (a) Dial outbound course on Course Selector D.G., then dial same course on OBS.
- (b) After established on outbound radial, position coupler mode selector to HDG mode and select outbound procedure turn heading. After 40 seconds to 1 minute select a turn in the desired direction with the Course Selector D.G. to the inbound procedure turn heading.
- (c) Set OBS to inbound course.
- (d) When aircraft heading is 45° to the inbound course, dial Course Selector D.G. to inbound course and position coupler mode selector to OMNI mode.

NOTE

For precise tracking over OMNI station, without "S" turn, position coupler mode selector to HDG mode just prior to station passage. If holding pattern is desired, position coupler mode selector to HDG mode at station passage inbound and select outbound heading in direction of turn. After elapsed time, dial inbound course on Course Selector D.G. When aircraft heading is 45° to radial, position coupler mode selector to OMNI mode.

(6) LOC Approach Only

- (a) To intercept dial ILS outbound course on Course Selector D.G. When stabilized, position coupler mode selector to LOC REV mode.
- (b) After interception and when beyond outer marker, position coupler mode selector to HDG mode and dial outbound procedure turn heading. After one minute, dial inbound procedure turn heading in direction of turn.
- (c) When aircraft heading is 45° to ILS inbound course dial inbound course on Course Selector D.G. and position coupler mode selector to LOC NORM mode.
- (d) At the missed approach point (M.A.P.), or when missed approach is elected, position coupler mode selector to HDG mode and execute missed approach procedure.

(7) LOC Approach - Back Course (Reverse)

- (a) To intercept dial ILS Back Course outbound heading on Course Selector D.G. When stabilized, position coupler mode selector to LOC NORM mode.
- (b) After interception and when beyond fix, position coupler mode selector to HDG and dial outbound procedure turn heading. After one minute, dial inbound procedure turn heading in direction of turn.
- (c) When heading 45° to inbound course, dial inbound course on Course Selector D.G. and position coupler mode selector to LOC REV mode.
- (d) Approximately 1/2 mile from runway, position coupler mode selector to HDG mode to prevent "S" turn over ILS station near runway threshold.
- (e) Missed approach - same as Front Course. (See (6) d)

CHEROKEE ARCHER

c. EMERGENCY OPERATION

- (1) In an emergency the AutoControl can be disconnected by placing the A/P ON/OFF switch to the "OFF" position.
- (2) The AutoControl can be overpowered at either control wheel.
- (3) An Autopilot runaway, with a 3 second delay in the initiation of recovery, while operating in a climb, cruise or descending flight could result in a 60° bank and 100 foot altitude loss.
- (4) An Autopilot runaway, with a 1 second delay in the initiation of recovery, during an approach operation, coupled or uncoupled, could result in a 10° bank and 10 foot altitude loss.

3. PERFORMANCE

No change.

**F.A.A. APPROVED
EMERGENCY PROCEDURES**

NONE APPLICABLE TO THIS AIRPLANE

EMERGENCY PROCEDURES

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EMERGENCY PROCEDURES

INTRODUCTION

This section contains procedures that are recommended if an emergency condition should occur during ground operation, takeoff, or in flight. These procedures are suggested as the best course of action for coping with the particular condition described, but are not a substitute for sound judgment and common sense. Since emergencies rarely happen in modern aircraft, their occurrence is usually unexpected, and the best corrective action may not always be obvious. Pilots should familiarize themselves with the procedures given in this section and be prepared to take appropriate action should an emergency arise.

Most basic emergency procedures, such as power off landings, are a normal part of pilot training. Although these emergencies are discussed here, this information is not intended to replace such training, but only to provide a source of reference and review, and to provide information on procedures which are not the same for all aircraft. It is suggested that the pilot review standard emergency procedures periodically to remain proficient in them.

ENGINE POWER LOSS DURING TAKEOFF

The proper action to be taken if loss of power occurs during takeoff will depend on circumstances.

1. If sufficient runway remains for a normal landing, land straight ahead.
2. If insufficient runway remains, maintain a safe airspeed and make only a shallow turn if necessary to avoid obstructions. Use of flaps depends on circumstances. Normally, flaps should be fully extended for touchdown.
3. If you have gained sufficient altitude to attempt a restart, proceed as follows:
 - a. MAINTAIN SAFE AIRSPEED
 - b. FUEL SELECTOR - SWITCH TO ANOTHER TANK CONTAINING FUEL
 - c. ELECTRIC FUEL PUMP - CHECK ON
 - d. MIXTURE - CHECK RICH
 - e. CARBURETOR HEAT - ON

NOTE

If engine failure was caused by fuel exhaustion, power will not be regained after tanks are switched until empty fuel lines are filled, which may require up to ten seconds.

If power is not regained, proceed with the POWER OFF LANDING procedure.

ENGINE POWER LOSS IN FLIGHT

Complete engine power loss is usually caused by fuel flow interruption, and power will be restored shortly after fuel flow is restored. If power loss occurs at low altitude, the first step is to prepare for an emergency landing (See POWER OFF LANDING). Maintain an airspeed of at least 80 MPH IAS, and if altitude permits, proceed as follows:

1. Fuel Selector - Switch to another tank containing fuel.
2. Electric Fuel Pump - On
3. Mixture - Rich
4. Carburetor Heat - On
5. Engine Gauges - Check for an indication of the cause of power loss.
6. Primer - Check Locked
7. If no fuel pressure is indicated, check tank selector position to be sure it is on a tank containing fuel.

When power is restored:

8. Carburetor Heat - Off
9. Electric Fuel Pump - Off

If the above steps do not restore power, prepare for an emergency landing.

If time permits:

1. Ignition Switch - "L" then "R" then back to "BOTH."
2. Throttle and Mixture - Different settings. (This may restore power if the problem is too rich or too lean a mixture, or if there is partial fuel system restriction.)
3. Try other fuel tank. (Water in the fuel could take some time to be used up, and allowing the engine to windmill may restore power. If power loss is due to water, fuel pressure indications will be normal.)

NOTE

If engine failure was caused by fuel exhaustion, power will not be regained after tanks are switched until empty fuel lines are filled, which may require up to ten seconds.

If power is not restored, proceed with POWER OFF LANDING procedure.

POWER OFF LANDING

If loss of power occurs at altitude, trim the aircraft for best gliding angle (80 MPH IAS - Air Cond. OFF) and look for a suitable field. If measures taken to restore power are not effective, and if time permits, check your charts for airports in the immediate vicinity; it may be possible to land at one if you have sufficient altitude. If possible, notify the FAA by radio of your difficulty and intentions. If another pilot or passenger is aboard, let them help.

When you have located a suitable field, establish a spiral pattern around this field. Try to be at 1000 feet above the field at the downwind position to make a normal approach. Excess altitude may be lost by widening your pattern, using flaps or slipping, or a combination of these.

Touchdown should normally be made at the lowest possible airspeed, with full flaps.

When committed to landing:

1. Ignition - Off
2. Master Switch - Off
3. Fuel Selector - Off
4. Mixture - Idle Cut-Off
5. Seat Belt (and harness if available) - Tight

SPINS

Intentional spins are prohibited in this aircraft. If a spin is inadvertently entered, immediately use the following recovery procedures:

1. THROTTLE - IDLE
2. RUDDER - FULL OPPOSITE TO DIRECTION OF ROTATION
3. CONTROL WHEEL - FULL FORWARD
4. RUDDER - NEUTRAL (WHEN ROTATION STOPS)
5. CONTROL WHEEL - AS REQUIRED TO SMOOTHLY REGAIN LEVEL FLIGHT ATTITUDE

OPEN DOOR

The cabin door on the Cherokee is double latched, so the chances of its springing open in flight at both the top and bottom are remote. However, should you forget the upper latch, or not fully engage the lower latch, the door may spring partially open. This will usually happen at takeoff or soon afterward. A partially open door will not affect normal flight characteristics, and a normal landing can be made with the door open.

If both upper and lower latches are open, the door will trail slightly open, and airspeed will be reduced slightly.

To close the door in flight, proceed as follows:

1. Slow aircraft to 100 MPH IAS.
2. Cabin Vents - Close
3. Storm Window - Open
4. If upper latch is open - latch. If lower latch is open - open top latch, push door further open, and then close rapidly. Latch top latch.

A slip in the direction of the open door will assist in latching procedure.

FIRE

The presence of fire is noted through smoke, smell, and heat in the cabin. It is essential that the source of the fire be promptly identified through instrument readings, character of the smoke, or other indications, since the action to be taken differs somewhat in each case.

SOURCE OF FIRE - CHECK

1. Electrical Fire (Smoke in Cabin):
 - a. Master Switch - Off
 - b. Vents - Open
 - c. Cabin Heat - Off
 - d. Land as soon as possible.
2. Engine Fire (In Flight):
 - a. Fuel Selector - Off
 - b. Throttle - Closed
 - c. Mixture - Idle Cut-Off
 - d. Heater - Off (In all cases of fire)
 - e. Defroster - Off (In all cases of fire)
 - f. If terrain permits, land immediately.

NOTE

The possibility of an engine fire in flight is extremely remote. The procedure given above is general and pilot judgment should be the deciding factor for action in such an emergency.

3. Engine Fire (During Start):

Engine fires during start are usually the result of overpriming. The following procedure is designed to draw the excess fuel back into the induction system.

 - a. If engine has not started:
 - (1) Mixture - Idle Cut-Off
 - (2) Throttle - Open
 - (3) Turn engine with starter (This is an attempt to pull the fire into the engine.)

- b. If engine has already started and is running, continue operating to try pulling the fire into the engine.
- c. In either case stated in (a) and (b), if the fire continues longer than a few seconds, the fire should be extinguished by the best available external means.
- d. If external fire extinguishing is to be applied:
 - (1) Fuel Selector Valves - Off
 - (2) Mixture - Idle Cut-Off

LOSS OF OIL PRESSURE

Loss of oil pressure may be either partial or complete. A partial loss of oil pressure usually indicates a malfunction in the oil pressure regulating system, and a landing should be made as soon as possible to investigate the cause and prevent engine damage.

A complete loss of oil pressure indication may signify oil exhaustion or may be the result of a faulty gauge. In either case, proceed toward the nearest airport, and be prepared for a forced landing. If the problem is not a pressure gauge malfunction, the engine may stop suddenly. Maintain altitude until such time as a dead stick landing can be accomplished. Don't change power settings unnecessarily, as this may hasten complete power loss.

Depending on the circumstances, it may be advisable to make an off airport landing while power is still available, particularly if other indications of actual oil pressure loss, such as sudden increases in temperatures, or oil smoke, are apparent, and an airport is not close.

If engine stoppage occurs, proceed to POWER OFF LANDING.

LOSS OF FUEL PRESSURE

1. Electric Boost Pump - On
2. Fuel Selector - Check on full tank

If problem is not an empty fuel tank, land as soon as practical and have engine-driven fuel pump checked.

HIGH OIL TEMPERATURE

An abnormally high oil temperature indication may be caused by a low oil level, an obstruction in the oil cooler, damaged or improper baffle seals, a defective gauge, or other causes. Land as soon as practical at an appropriate airport and have the cause investigated.

A steady, rapid rise in oil temperature is a sign of trouble. Land at the nearest airport and let a mechanic investigate the problem. Watch the oil pressure gauge for an accompanying loss of pressure.

ALTERNATOR FAILURE

Loss of alternator output is detected through a zero reading on the ammeter. Before executing the following procedure, insure that the reading is zero and not merely low by actuating an electrically powered device, such as the landing light. If no increase in the ammeter reading is noted, alternator failure can be assumed.

1. Reduce Electrical Load.
2. Alternator Circuit Breakers - Check
3. "Alt" Switch - Off (for 1 second), then On

If the ammeter continues to indicate no output, or alternator will not stay reset, turn off "Alt" switch, maintain minimum electrical load and land as soon as practical. All electrical load is being supplied by the battery.

ENGINE ROUGHNESS

Engine roughness is usually due to carburetor icing which is indicated by a drop in RPM, and may be accompanied by a slight loss of airspeed or altitude. If too much ice is allowed to accumulate, restoration of full power may not be possible; therefore, prompt action is required.

Turn carburetor heat on (See Note). RPM will decrease slightly and roughness will increase. Wait for a decrease in engine roughness or an increase in RPM, indicating ice removal. If no change in approximately one minute, return the carburetor heat to COLD. If the engine is still rough, try steps below:

1. Mixture - Adjust for maximum smoothness. Engine will run rough if too rich or too lean.
2. Electric Fuel Pump - On
3. Fuel Selector - Change tanks to see if fuel contamination is the problem.
4. Engine Gauges - Check for abnormal readings. If any gauge readings are abnormal, proceed accordingly.
5. Magneto Switch - "L" then "R", then back to "BOTH." If operation is satisfactory on either magneto, proceed on that magneto at reduced power, with mixture full rich, to a landing at the first available airport.

If roughness persists, prepare for a precautionary landing at pilot's discretion.

NOTE

Partial carburetor heat may be worse than no heat at all, since it may partially melt ice, which will refreeze in the intake system. When using carburetor heat, therefore, always use full heat, and when ice is removed return the control to the full cold position.

NOTE

Pilot's Operating Manual Revision Only

This Is Not A Complete Manual

Rev. 8 - 761 556 (PR900320) Dated March 20, 1990

This revision shall be inserted into the current PA-28-180 Cherokee Archer Pilot's Operating Manual, P/N 761 556, issued July 9, 1973.

PILOT'S OPERATING MANUAL LOG OF REVISIONS (cont)

Revision	Revised Pages	Description	Date
Rev. 3 (cont)	9-i 9-2 9-3 9-4 9-5 9-6 9-7 9-8 9-9 9-10 10-11	Revised Performance Charts Index. Revised Takeoff Chart. Revised Climb Chart. Revised Range Chart. Revised Engine Chart. Revised Airspeed Chart. Revised Stall Speed vs Weight Chart. Revised Stall Speed vs Angle of Bank Chart. Revised Glide Chart. Revised Landing Chart. Revised Battery Service.	
Rev. 4 - 761 556 (PR750120)	1-2 2-1 AF/M W/B 7-4 7-8 8-2 10-8	Added 0-360-A4M engine designation and footnote. Added 0-360-A4M engine designation and footnote. Added Rev. 3 to Report: VB-558. Added Rev. 4 to Report: VB-547. Added new item 9.; revised existing item nos. under TAKEOFF. Revised ELT info. Added item 11. Added 0-360-A4M engine designation under Oil Requirements.	Jan. 20, 1975
Rev. 5 - 761 556 (PR751121)	iii 1-1 1-2 2-1 AF/M W/B 8-1 9-4 9-6	Added concluding ser. no. to Applicability. Revised Cruise Speed and Range figures; revised Empty Weight and Useful Load; deleted footnote. Deleted footnote. Revised Airframe info; deleted footnote. Added Rev. 4 to Report: VB-558. Added Rev. 5 to Report: VB-547. Revised item 8 (Fuel Warning Tip). Revised Cruise Performance - Range chart. Revised Cruise Performance - True Airspeed chart.	Nov. 21, 1975
Rev. 6 - 761 556 (PR790326)	2-8 2-8a 2-14 W/B 7-2 7-8, 7-9	Added Warning. Added info. Added Caution. Added Rev. 6 to Report: VB-547. Revised items. Revised ELT info.	March 26, 1979

PILOT'S OPERATING MANUAL LOG OF REVISIONS (cont)

Revision	Revised Pages	Description	Date
Rev. 7 - 761 556 (PR840316)	1-2 2-16 AFM W/B 7-1 7-4 9-i 10-1 10-9 10-11 10-12, 10-13	Revised fuel and oil. Revised para. Added Rev. 5 to Report VB-558. Added Rev. 7 to Report VB-547. Added to preflight. Added to text info. Added Warning. Added to text info. Added text info. Revised para. Revised preventive maintenance info.	March 16, 1984
Rev. 8 - 761 556 (PR900320)	10-7 10-8 10-11, 10-12 10-13	Relocated info. from p. 10-8. Moved info. to p. 10-7. Revised para. Revised para. Revised para.	March 20, 1990

BRAKE SERVICE

The brake system is filled with MIL-H-5606 (petroleum base) hydraulic brake fluid. This should be checked at every 50 hour inspection and replenished when necessary by filling the brake reservoir on the firewall to the indicated level. If the entire system has to be refilled, it should be done by filling from the brake end of the system with fluid under pressure. This will eliminate air from the system.

No adjustment of brake clearances is necessary on the Cherokee. If after extended service the brake blocks become worn excessively, they are easily replaced with new segments.

LANDING GEAR SERVICE

The three landing gears use Cleveland 6.00 x 6 wheels, the main gear wheels being provided with brake drums and Cleveland single disc hydraulic brake assemblies. All three wheels use 6.00 x 6, four-ply rating, Type III tires with tubes.

Main wheels are removed by taking off the wheel fairings, hub cap, axle nut, and the two bolts holding the brake segment into place. The wheel will slip easily from the axle.

Tires are removed from the wheels by first deflating the tire, then removing the through bolts, and separating the wheel halves.

Landing gear oleo struts should be checked for proper strut exposures and fluid leaks. The required extensions for the strut when under normal static load (empty weight of airplane plus full fuel and oil) are 3-1/4 inches for the nose gear and 4-1/2 inches for the main gear. Should the strut exposure be below that required, it should be determined whether air or oil is required by first raising the airplane on jacks. Depress the valve core to allow air to escape from the strut housing chamber. Remove the filler plug and slowly raise the strut to full compression. If the strut has sufficient fluid, it will be visible up to the bottom of the filler plug hole and will then require only proper inflation.

Should fluid be below the bottom of the filler plug hole, oil should be added. Replace the plug with valve core removed; attach a clear plastic hose to the valve stem of the filler plug and submerge the other end in a container of hydraulic fluid (MIL-H-5606). Fully compress and extend the strut several times, thus drawing fluid from the container and expelling air from the strut chamber. To allow fluid to enter the bottom chamber of the main gear strut housing, the torque link assembly must be disconnected to let the strut be extended a minimum of 10 inches (the nose gear torque links need not be disconnected). Do not allow the strut to extend more than 12 inches. When air bubbles cease to flow through the hose, compress the strut fully and again check fluid level. Reinstall the valve core and filler plug, and the main gear torque links, if disconnected.

With fluid in the strut housing at the correct level, attach a strut pump to the air valve and with the airplane on the ground, inflate the oleo strut to the correct height.

In jacking the Cherokee for landing gear service, a jack kit (available through the Piper Dealers and Distributors) should be used. This kit consists of two hydraulic jacks and a tail stand. At least 250 pounds of ballast should be placed on the tail stand before jacking the aircraft. The jacks should be placed under the jack points on the wing and the airplane jacked up until the tail skid is at the right height to attach the tail stand. After attaching the tail stand and adding ballast, jacking may be continued until the aircraft is at the height desired.

The steering arms from the rudder pedals to the nose wheel are adjusted at the rudder pedals or at the nose wheel by turning in or out the threaded rod end bearings. Adjustment is normally accomplished at the forward end of the rods and should be done in such a way that the nose wheel is in line with the fore and aft axis of the plane when the rudder pedals and rudder are centered. Alignment of the nose wheel can be checked by pushing the airplane back and forth with the rudder centered to determine that the plane follows a perfectly straight line. The turning arc of the nose wheel is 30 degrees in either direction and is factory adjusted at stops on the bottom of the forging.

The steering arm stops should be carefully adjusted so that the nose wheel reaches its full travel just after the rudder hits its stops. This guarantees that the rudder will be allowed to move through its full travel.

PROPELLER SERVICE

The spinner and backing plate should be cleaned and inspected frequently for cracks. The propeller should be inspected before each flight for nicks, scratches, and corrosion. If found, they should be taken care of as soon as possible by a rated mechanic, because nicks and scratches cause areas of increased stress which can cause serious damage or loss of a propeller tip. The back face of the blades should be painted when necessary with flat black paint to retard glare to the pilot's eyes. To prevent corrosion, the surface should be cleaned and waxed periodically.

OIL REQUIREMENTS

The oil capacity of the Lycoming O-320-A4A and O-360-A4M series engines is 8 quarts, and the minimum safe quantity is 2 quarts. It is recommended that engine oil be drained and renewed every 50 hours. The oil filter element should be changed every 50 hours of operation. The interval between oil and oil filter changes should not exceed a total of four months. Under unfavorable dusty conditions, the oil and oil filter should be changed more frequently. Should fuel other than the specified octane rating for the power plant be used, refer to the latest issue of Lycoming Service Letter No. L185 for additional information and recommended service procedures.

The following seasonal aviation oil grades and seasonal ambient temperature ranges are recommended:

Average Ambient Temperature	MIL-L-6082B Mineral SAE Grade	MIL-L-22851 Ashless Dispersant SAE Grades
All Temperatures	--	15W-50 or 20W-50
Above 80°F	60	60
Above 60°F	50	40 or 50
30°F to 90°F	40	40
0°F to 70°F	30	30, 40 or 20W-40
0°F to 90°F	20W-50	20W-50 or 15W-50
Below 10°F	20	30 or 20W-30

When operating temperatures overlap indicated ranges, use the lighter grade oil.

NOTE

Refer to the latest issue of Textron Lycoming Service Instruction 1014 (Lubricating Oil Recommendations) for further information.

BATTERY SERVICE

Access to the 12-volt battery is through the removal of the panel at the right rear side of the baggage compartment. The battery box has a plastic drain tube which should be opened occasionally to drain off any accumulation of liquid. Check the battery for proper fluid level. (Do not fill above the baffle plates.) Use only water - no acid. A hydrometer check should be performed to determine the percent of charge present in the battery.

If the battery is not up to charge, recharge starting at a 4 amp rate and finishing with a 2 amp rate. Quick charges are not recommended.

FACTS YOU SHOULD KNOW

The Federal Aviation Administration (FAA) occasionally publishes Airworthiness Directives (ADs) that apply to specific groups of aircraft. They are mandatory changes and are to be complied with within a time limit set by the FAA. When an AD is issued, it is sent by the FAA to the latest registered owner of the affected aircraft and also to subscribers of their service. Owners should periodically check with their Piper Service Center or Piper's Customer Services Department to see whether they have the latest AD against their airplane. The owner is solely responsible for keeping up with ADs.

Piper Aircraft Corporation takes a continuing interest in having owners get the most efficient use from their airplane and keeping it in the best mechanical condition. Consequently, Piper Aircraft, from time to time, issues service releases including Service Bulletins, Service Letters, Service Spares Letters, and others relating to the airplane.

Piper Service Bulletins are of special importance and Piper considers compliance mandatory. These are sent directly to the latest FAA-registered owners in the United States (U.S.) and Piper Service Centers worldwide. Depending on the nature of the release, material and labor allowances may apply. This information is provided to all authorized Piper Service Centers.

Service Letters deal with product improvements and servicing techniques pertaining to the airplane. They are sent to Piper Service Centers and, if necessary, to the latest FAA-registered owners in the U.S. Owners should give careful attention to Service Letter information.

Service Spares Letters offer improved parts, kits, and optional equipment which were not available originally, and which may be of interest to the owner.

Piper Aircraft Corporation offers a **subscription service** for Service Bulletins, Service Letters, and Service Spares Letters. This service is available to interested persons such as owners, pilots, and mechanics at a nominal fee, and may be obtained through an authorized Piper Service Center or Piper's Customer Services Department.

Service manuals, parts catalogs, and revisions to both, are available from Piper Service Centers or Piper's Customer Services Department. Any correspondence regarding the airplane should include the airplane model and serial number to ensure proper response.

Pilot's Operating Manual supplements are distributed by the manufacturer as necessary. These revisions and additions should be studied and put into the operating manual to keep it up to date. This manual contains important information about the operation of the aircraft and should be kept with the aircraft at all times, even after resale. Every owner, to avail themselves of the latest information concerning their airplane, should stay in close contact with an authorized Piper Service Center or Piper's Customer Services Department.

If the owner desires to have his aircraft modified, he must obtain FAA approval for the alteration. Major alterations accomplished in accordance with Advisory Circular 43.13-2, when performed by an A & P mechanic, may be approved by the local FAA office. Major alterations to the basic airframe or systems not covered by AC43.13-2 require a Supplemental Type Certificate.

The owner or pilot is required to ascertain that the following Aircraft Papers are in order and in the aircraft.

- a. To be displayed in the aircraft at all times:
 1. Aircraft Airworthiness Certificate Form FAA-1362B.
 2. Aircraft Registration Certificate Form FAA-500A.
 3. Aircraft Radio Station License FCC-404A, if transmitters are installed.
- b. To be carried on the aircraft at all times:
 - (1) Aircraft Flight Manual.
 - (2) Weight and Balance Data plus a copy of the latest Repair and Alteration Form FAA-337, if applicable.
 - (3) Aircraft equipment list.

Although the aircraft and engine logbooks are not required to be in the aircraft, they should be made available upon request. Logbooks should be complete and up to date. Good records will reduce maintenance cost by giving the mechanic information about what has or has not been accomplished.

PREVENTIVE MAINTENANCE

The holder of a Pilot Certificate issued under FAR Part 61 may perform certain preventive maintenance described in FAR Part 43. This maintenance may be performed only on an aircraft which the pilot owns or operates and which is not used to carry persons or property for hire, except as provided in applicable FAR's. Although such maintenance is allowed by law, each individual should make a self-analysis as to whether he has the ability to perform the work.

All other maintenance required on the airplane should be accomplished by appropriately licensed personnel.

If maintenance is accomplished, an entry must be made in the appropriate logbook. The entry should contain:

- (a) The date the work was accomplished.
- (b) Description of the work.
- (c) Number of hours on the aircraft.
- (d) The certificate number of pilot performing the work.
- (e) Signature of the individual doing the work.

REQUIRED SERVICE AND INSPECTION PERIODS

The Owner Service Agreement which the owner receives upon delivery of the aircraft should be kept in the aircraft at all times. This identifies him to authorized Piper dealers and entitles the owner to receive service in accordance with the regular service agreement terms. This agreement also entitles the transient owner full warranty by any Piper dealer in the world.

Piper Aircraft Corporation has developed inspection items and required inspection intervals for the PA-28 (see PA-28 Service and Inspection Manuals). The PA-28 Inspection Manual contains appropriate forms, and all inspection procedures should be complied with by a properly trained, knowledgeable, and qualified mechanic at an authorized Piper Service Center or a reputable repair shop. Piper Aircraft Corporation cannot accept responsibility for the continued airworthiness of any aircraft not maintained to these standards, and/or not brought into compliance with applicable Service Bulletins issued by Piper Aircraft Corporation, instructions issued by the engine, propeller, or accessory manufacturers, or Airworthiness Directives issued by the FAA.

A Programmed Inspection, approved by the Federal Aviation Administration (FAA), is also available to the owner. This involves routine and detailed inspections to allow maximum utilization of the airplane. Maintenance inspection costs are reduced, and the maximum standard of continued airworthiness is maintained. Complete details are available from Piper Aircraft Corporation.

In addition, but in conjunction with the above, the FAA requires periodic inspections on all aircraft to keep the Airworthiness Certificate in effect. The owner is responsible for assuring compliance with these inspection requirements and for maintaining proper documentation in logbooks and/or maintenance records.

A spectrographic analysis of the engine oil is available from several sources. This inspection, if performed properly, provides a good check of the internal condition of the engine. To be accurate, induction air filters must be cleaned or changed regularly, and oil samples must be taken and sent in at regular intervals.

TYPE OF LUBRICANT				SPECIAL INSTRUCTIONS (cont)	
IDENTIFICATION LETTER	LUBRICANT	SPECIFICATION	PREFERRED PRODUCT AND VENDOR		
A	LUBRICATING OIL, GENERAL PURPOSE, LOW TEMP.	MIL-L-7870		8. STABILATOR TRIM PULLEYS - LUBRICATION MAY BE EXTENDED TO 250 HOURS WHEN DUSTY CONDITIONS ARE AT A MINIMUM.	
B	LUBRICATING OIL, AIRCRAFT RECIPROCATING ENGINE (PISTON) GRADE AS SPECIFIED SAE 50 ABOVE 90°F AIR TEMP. SAE 40 30° TO 90°F AIR TEMP. SAE 30 0° TO 70°F AIR TEMP. SAE 20 BELOW 10°F AIR TEMP.	MIL-L-6082		9. AILERON HINGES WITH TEFLON SLEEVES SHOULD NOT BE LUBRICATED. AILERON HINGES WITHOUT TEFLON SLEEVES SHOULD FIRST BE CLEANED WITH A DRY TYPE SOLVENT THEN LUBRICATED WITH MIL-L-7870 LUBRICATING OIL.	
C	HYDRAULIC FLUID, PETROLEUM BASE	MIL-H-5606		10. THIS TRANSMISSION TO BE 1/2 FULL OF GREASE. APPLY GREASE DURING ASSEMBLY AND LUBRICATE TRANSMISSION BALL NUT AND SCREW WITH MIL-G-23827 GREASE.	
D	GREASE, AIRCRAFT AND INSTRUMENT, GEAR AND ACTUATOR SCREW	MIL-G-23827			
E	GREASE, AIRCRAFT, HIGH TEMP.		TEXACO MARFAK ALL PURPOSE GREASE, MOBIL GREASE 77 (OR MOBILUX EP2), SHELL ALVANIA EP GREASE 2		
F	PARKER "O" RING LUBRICANT				
G	AERO LUBRIPLATE		FISKE BROS. REFINING CO.		
H	FLUOROCARBON RELEASE AGENT DRY LUBRICANT	*MS-122			
I	GREASE - LUBRICATION GEN. PURPOSE AIRCRAFT	MIL-G-7711			

SPECIAL INSTRUCTIONS	
1. AIR FILTER - TO CLEAN FILTER, TAP GENTLY TO REMOVE DIRT PARTICLES. DO NOT BLOW OUT WITH COMPRESSED AIR OR USE OIL. REPLACE FILTER IF PUNCTURED OR DAMAGED.	
2. BEARINGS AND BUSHINGS - CLEAN EXTERIOR WITH A DRY TYPE SOLVENT BEFORE LUBRICATING.	
3. WHEEL BEARINGS - DISASSEMBLE AND CLEAN WITH A DRY TYPE SOLVENT. ASCERTAIN THAT GREASE IS PACKED BETWEEN THE BEARING ROLLER AND CONE. DO NOT PACK GREASE IN WHEEL HOUSING.	
4. OLEO STRUTS, AND BRAKE RESERVOIR - FILL PER INSTRUCTIONS ON UNIT OR CONTAINER, OR REFER TO SERVICE MANUAL, SECTION II.	
5. "O" RING, CONTROL SHAFT BUSHING (WITH 1.125 INCH SHAFT ONLY) - DISASSEMBLE "O" RING RETAINER PLATES FROM INSTRUMENT PANEL, LUBRICATE "O" RING AND REASSEMBLE.	
6. LUBRICATION POINTS - WIPE ALL LUBRICATION POINTS CLEAN OF OLD GREASE, OIL, DIRT, ETC. BEFORE LUBRICATING.	
7. INTERVALS BETWEEN OIL CHANGES CAN BE INCREASED AS MUCH AS 100% ON ENGINES EQUIPPED WITH FULL FLOW (CARTRIDGE TYPE) OIL FILTERS - PROVIDED THE ELEMENT IS REPLACED EACH 50 HOURS OF OPERATION.	

NOTES	
1. PILOT AND PASSENGER SEATS - LUBRICATE TRACK ROLLERS AND STOP PINS AS REQUIRED. (TYPE OF LUBRICANT: "A")	
2. WHEEL BEARINGS REQUIRE CLEANING AND REPACKING AFTER EXPOSURE TO AN ABNORMAL QUANTITY OF WATER.	
3. FUEL SELECTOR VALVE - LUBRICATE FUEL SELECTOR VALVE AS REQUIRED. REFER TO PIPER SERVICE LETTER NO. 351.	
4. SEE LYCOMING SERVICE INSTRUCTIONS NO. 1014 FOR USE OF DETERGENT OIL.	

CAUTIONS	
1. DO NOT USE HYDRAULIC FLUID WITH A CASTOR OIL OR ESTER BASE.	
2. DO NOT OVER-LUBRICATE COCKPIT CONTROLS.	
3. DO NOT APPLY LUBRICANT TO RUBBER PARTS.	

EXAMPLE

Lubrication Nomenclature

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA779NE

This certificate, issued to U.S. Propeller Service of Connecticut
P.O. Box 415
East Haddam, Connecticut 06423

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 3 of the Civil Air Regulations.

Original Product — Type Certificate Number: 2A13
Make: Piper
Model: PA-28-180, -181

Description of Type Design Change:

Installation of McCauley 1A170/BMS Propeller in accordance with U.S. Propeller Service of Connecticut Report No 28181, dated August 27, 1990, or later FAA approved revisions. S/N MFOR8

Limitations and Conditions:

1. Propeller diameter: not over 76", not under 74"
2. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined that the interrelationship between this change and any of those other previously approved modifications will introduce no adverse effect upon the airworthiness of those aircraft.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: July 9, 1990

Date issued:

Date of issuance: September 10, 1990

Date amended:



By direction of the Administrator

Ronald L. Vavruska
(Signature)

Ronald L. Vavruska
Manager, Boston Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

U.S. PROPELLER SERVICE OF CT.
P.O. BOX 415
GOODSPEED AIRPORT
EAST HADDAM, CT. 06423

Dated 08-27-90

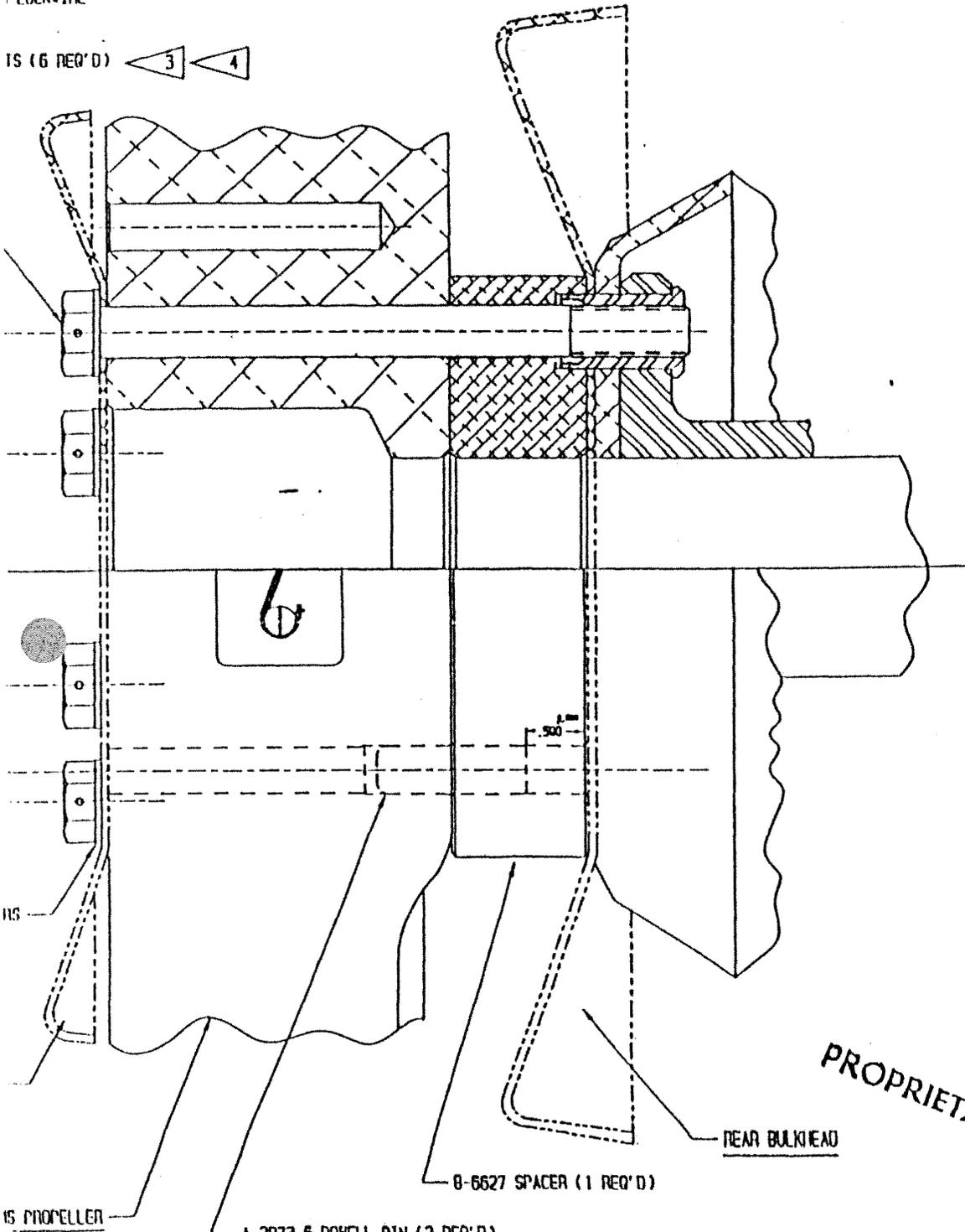
Report # USP 28181

Descriptive Data and Installation Instructions for conversion of Piper PA 28-180 and PA 28-181 airplanes to the McCauley 1A170/BMS 7660 Fixed Pitch Propeller.

1. Remove existing Sensenich propeller per instructions in Piper PA28-180 Maintenance Manuals.
2. Install McCauley Propeller, model number 1A170/BMS 7660 as per instruction on McCauley drawing #C6626 titled "Propeller Installation 1A170/BMS" dated May 22nd 1990 or later FAA approved revisions.
3. Fill out a 337 form for the aircraft alteration as appropriate, and make all necessary changes to aircraft records. Referencing this STC.
4. Revise the aircraft weight and balance. The McCauley propeller weighs 36 lbs.

DO NOT WEAR
LOCKWIRE

IS (6 REQ'D)  



PROPRIETARY

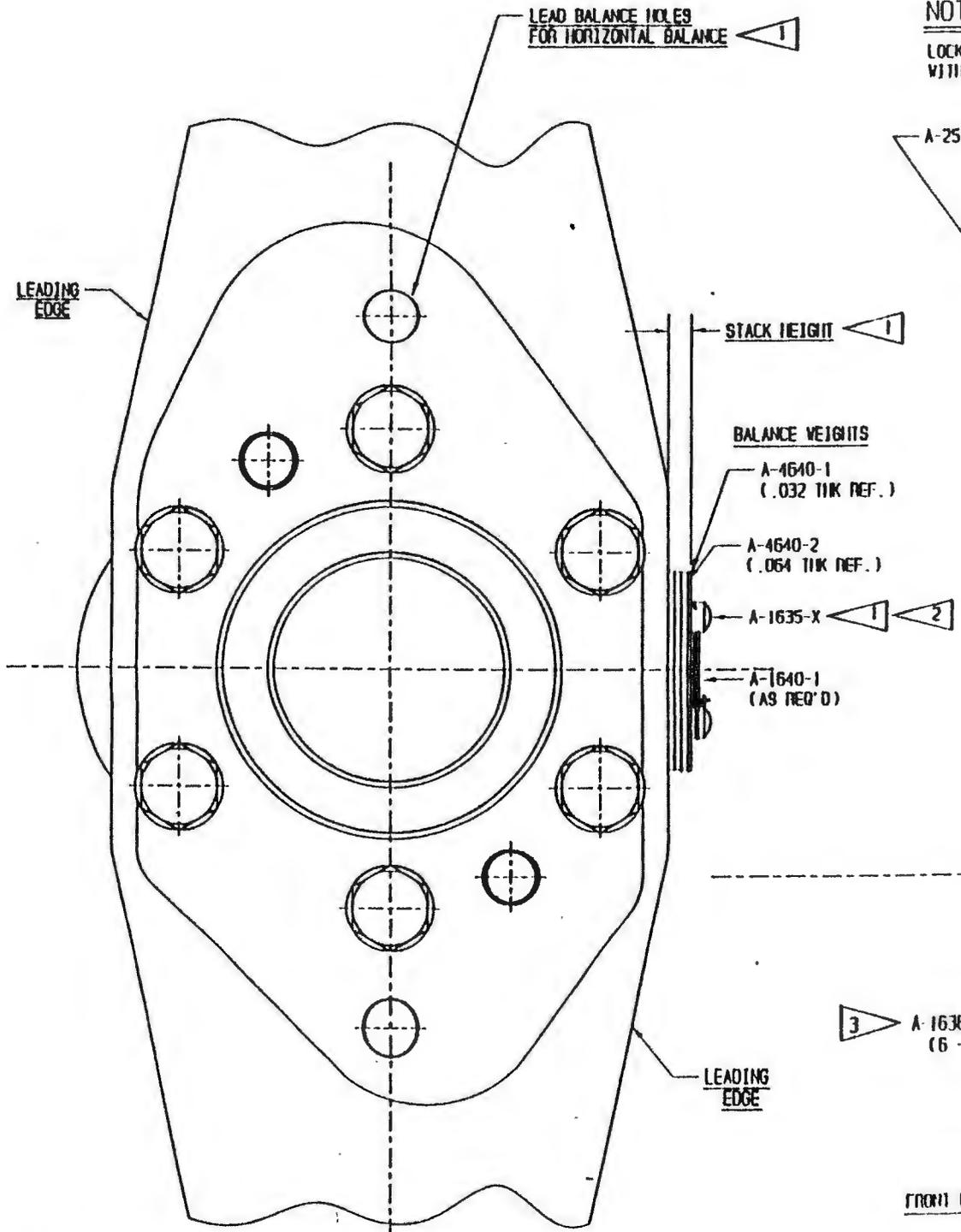
HUB PROCESS C-4021

UNLESS OTHERWISE SPECIFIED:			
DIMENSIONS IN INCHES & ANGLES			
DIMENSIONS IN MILLIMETERS			
DATE	TIME	BY	
4-19-80	5:23:00	JAY	
DRAWN	CHECKED	APPROVED	

MATERIAL	HEAT TREAT	PROT. TREATMENT	MAKE FROM	NO. REQ'D	NEXT ASS'Y	SCALE: FULL
McCAULEY ACCESSORY DIVISION WINDYBOLT, OHIO 45377						C-6626 DRAWING NUMBER
PROPELLER INSTALLATION 1A170/BMS PART NAME						

NOTE:
 LOCK ORILLER
 WITH A 1640

A-2513-39 00



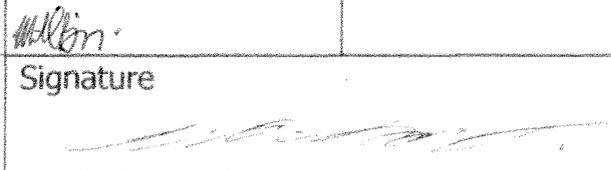
- 1** SEE DRAW C-4642 FOR BALANCE INFORMATION
- 2** TORQUE SCREWS 36 TO 30 I.B.-IN
- 3** THESE PARTS ARE INCLUDED IN BOLT KIT B-4622-24
- 4** BOLTS TO BE TIGHTENED WITH TORQUE WRENCH TO 55-65 LB.FT.
- 5** USE PAINT SCHEME D-6629

3 A-1638-2 WASHS (6 - REQ'D)

FRONT BULKHEAD

MODEL 1A1704

Aircraft Weight and Balance Revision

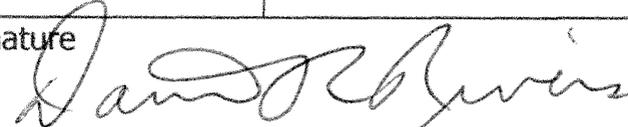
Tail Number: N1185X			Date: 10-05-2007		
Prepared by: QUALITY AVIONICS, INC. 40 OAKLEY-KELLY DR MERCER, PA 16137			Work Order No: 0708212		
			Type Certificate Data No:		
Aircraft Make: PIPER	Model: PA28-180	Serial No: 28-7505222	Time: 2112.75 TACH		
Registered Owner: A. VINCENT LOMBARDI		Address: 2552 BLOSSOM LANE NEW CASTLE, PA 16105			
Maximum Weight 2450		CG Range FWD		AFT	
As Received; Date of Previous Weight and Balance: 06-06-2005		Useful Load: 938.41	EW: 1511.59	EWCG: 80.08	Moment: 131051.80
Notes:					
			Weight	Arm	Moment
REMOVED GARMIN GNS430 GPS/ NAV/ COM			-5.0	57	-285.00
REMOVED GA56 GPS ANTENNA			-0.25	118	-29.50
INSTALLED GARMIN GNS430W WAAS/ NAV/ COM			5.0	57	285.00
INSTALLED GARMIN GA35 WAAS ANTENNA			0.5	118	59.00
			0.00	0.00	0.00
			0.00	0.00	0.00
			0.00	0.00	0.00
			0.00	0.00	0.00
			0.00	0.00	0.00
			0.00	0.00	0.00
X As Calculated		Moment 131081.30	New Empty Weight CG		New Useful Load
As Weighed		Weight 1511.84	80.08 46.70		938.16
			Signature 		
			Repair Agency QU8R488X or License No:		

Aircraft Weight and Balance Revision

Tail Number: N1185X			Date: 06-06-2005		
Prepared by: QUALITY AVIONICS, INC. 40 OAKLEY-KELLY DR. MERCER, PA 16137			Work Order No: 0334791		
			Type Certificate Data No:		
Aircraft Make: PIPER	Model: PA28-180	Serial No: 28-7505222	Time: TACH 2057.7		
Registered Owner: A. VINCENT LOMBARDI		Address: 2552 BLOSSOM LANE NEW CASTLE, PA. 16105			
Maximum Weight	2450	CG Range	FWD	AFT	
As Received; Date of Previous Weight and Balance: 06-06-2005		Useful Load:	EW:	EWCG:	Moment:
		947.41	1502.59	86.86	120514.18
Notes:					
			Weight	Arm	Moment
INSTALLED GARMIN GNS 430 GPS/ NAV/ COM			6.5	57	370.50
INSTALLED GARMIN GA 56 GPS ANTENNA			0.25	118	29.50
INSTALLED GARMIN GI 106A C D I			1.5	61.0	91.50
INSTALLED LONE STAR CRB 6457 AVIONICS COOLING FAN			0.75	61.5	46.13
					0.00
					0.00
					0.00
					0.00
					0.00
					0.00
X As Calculated		Moment	New Empty Weight CG		New Useful Load
		121051.80	80.08 86.70		938.41
As Weighed		Weight	1511.59		
Signature <i>David R. Davis</i>					
Repair Agency QU8R488X or License No:					

SUPERSEDED 10-05-07

Aircraft Weight and Balance Revision

Tail Number: N1185X			Date: 06-06-2005		
Prepared by: QUALITY AVIONICS, INC. 40 OAKLEY-KELLY DR. MERCER, PA 16137			Work Order No: 0334791		
			Type Certificate Data No:		
Aircraft Make: PIPER	Model: PA28-180	Serial No: 28-7505222	Time: TACH 2057.7		
Registered Owner: A. VINCENT LOMBARDI		Address: 2552 BLOSSOM LANE NEW CASTLE, PA. 16205			
Maximum Weight	2450	CG Range		FWD	AFT
As Received; Date of Previous Weight and Balance: 07-29-1993		Useful Load: 931.01	EW: 1518.99	EWCG: 86.72	Moment: 131730.64
Notes:					
			Weight	Arm	Moment
REMOVED NARCO 890 DME			-3.9	57.5	-224.25
REMOVED NARCO 141 ADF INDICATOR			-1.3	61.0	-79.30
REMOVED NARCO 140-141-200 ADF LOOP ANTENNA			-1.6	162.0	-259.20
REMOVED NARCO 141 ADF RECEIVER			-2.5	57.5	-143.75
REMOVED ADF LOOP ANTENNA CABLE			-0.6	105.5	-63.30
REMOVED ADF SENSE ANTENNA AND CABLE			-0.4	147.5	-59.00
REMOVED NARCO UDA-3 DME ANTENNA			-0.25	120	-30.00
REMOVED NARCO NAV 11 NAV RECEIVER			-2.8	58.6	-164.08
REMOVED GARMIN GNC 250XL GPS/COM			-2.8	58.6	-164.08
REMOVED GARMIN GA 56 GPS ANTENNA			-0.25	118	-29.50
X As Calculated	Moment	130514.18	New Empty Weight CG		New Useful Load
As Weighed	Weight	1502.59	86.86		947.41
Signature 					
Repair Agency QU8R488X or License No:					

P.T. AERO SERVICE, INC.

Hanger 1, T.F. Green Airport

Warwick, R.I. 02886

401-737-7788

REVISED WEIGHT AND BALANCE DATA

Aircraft Make and Model: PA-28-180

Aircraft Serial Number: 28-7505222

Aircraft Registration Number: N1185X

Date of Revision: 7-29-93

Date of Previous Revision: 6-1-92

Aircraft	Weight	Arm	Moment
Removed;	1522.19	86.72	131742.80
Sensenich 76EM8S5-0-60 Propeller	- 39.2	3.8	-148.96
Installed;	36.0	3.8	136.8
McCauley 1A170/BMS Propeller			
New Empty Weight:	1518.99		
New Center of Gravity:		86.72	
New Moment:			131730.64

Gross Weight (Normal) 2450.00 lbs.
Gross Weight: (Utility) 1950.00 lbs.
New Useful Load: (Normal) 931.01 lbs.
(Utility) 431.01 lbs.

Note:

It is the responsibility of the pilot-in-command to insure that the aircraft is properly loaded using the above weight and balance data and the owners manual loading charts.

James A. Thistlewaite
A&P 2073585

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY

OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE Piper	MODEL PA28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK N1185X
2. OWNER	NAME (As shown on registration certificate) Halzel, Lenny	ADDRESS (As shown on registration certificate) 144 President Avenue Providence, RI 02906

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION				5. TYPE	
UNIT	MAKE	MODEL	SERIAL NO.	REPAIR	ALTERATION
AIRFRAME	***** (As described in item 1 above) *****				
POWERPLANT					
PROPELLER	McCauley	1A170/BMS	MF028		X
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS	B. KIND OF AGENCY	C. CERTIFICATE NO.
James A. Thistlewaite 5 Kennedy Drive Coventry, RI 02816	<input checked="" type="checkbox"/> U.S. CERTIFICATED MECHANIC	2073585
	<input type="checkbox"/> FOREIGN CERTIFICATED MECHANIC	
	<input type="checkbox"/> CERTIFICATED REPAIR STATION	
	<input type="checkbox"/> MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 7-29-93	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>James A. Thistlewaite</i>
-----------------	--

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA FLT. STANDARDS INSPECTOR	MANUFACTURER	<input checked="" type="checkbox"/>	INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	REPAIR STATION		CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE OF APPROVAL OR REJECTION 7-29-93	CERTIFICATE OR DESIGNATION NO. 2073585	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>James A. Thistlewaite</i>			

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed McCauley Model Number 1A170/BMS Serial Number MF028 propeller in accordance with U.S. Propeller Service of Connecticut Report No. 28181, dated August 27, 1990 or later FAA approved revisions and in accordance with STC #SA779NE. See weight and balance update in flight manual.

E N D

ADDITIONAL SHEETS ARE ATTACHED

United States of America
Department of Transportation Federal Aviation Administration
Supplemental Type Certificate

Number SA1194GL

This certificate, issued to Knots 2U, Inc.
1941 Highland Avenue
Wilmette, IL 60091

*certifies that the change in the type design for the following product with the limitations and conditions
therein as specified herein meets the airworthiness requirements of Part 3 of the Civil Air
Regulations. (See Type Certificate Data Sheet 2A13 for complete certification basis).*

Original Product — Type Certificate Number 2A13

Make Piper Aircraft Corporation

Model PA-28-140, PA-28-150, PA-28-160, PA-28-180,
PA-28-235, PA-28R-180, PA-28R-200, PA-28-151,
PA-28-161, PA-28-181, PA-28-201T, PA-28-236,
PA-28R-201, PA-28R-201T, PA-28RT-201, PA-28RT-201T

Description of Type Design Change.

Install fairings over flap hinges in accordance with Knots 2U, Inc. Flap Hinge
Fairing Installation Manual, issued 4/15/87, or later FAA approved revisions.

Limitations and Conditions.

The installer must determine the compatibility of this modification with previously
approved modifications.

*This certificate and the supporting data which is the basis for approval shall remain in effect until sus-
pended, suspended, revoked, or a termination date is otherwise established by the Administrator of the
Federal Aviation Administration.*

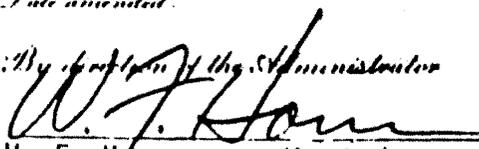
Date of application: April 29, 1987

Date issued:

Date of issuance: June 26, 1987

Date amended:



By  *Administrator*

W. F. Horn (Signature)
Manager, Chicago Aircraft Certification Office,
ACE-115C, Central Region, FAA

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA1216GL

This certificate, issued to Knots 2U, Inc.
1941 Highland Avenue
Wilmette, IL 60091

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 3 of the Civil Air Regulations. (See Type Certificate Data Sheet 2A13 for complete certification basis).

Original Product — Type Certificate Number: 2A13

Make: Piper Aircraft Corporation

Model: PA-28-140, PA-28-150, PA-28-160, PA-28-180,
PA-28-235, PA-28R-180, PA-28R-200, PA-28-151,
PA-28-161, PA-28-181, PA-28-201T, PA-28-236,
PA-28R-201, PA-28R-201T, PA-28RT-201, PA-28RT-201T

Description of Type Design Change:

Install Wing Root Fairings in accordance with Knots 2U, Inc. Wing Root Fairing Installation Manual, issued 7/4/87, or later FAA approved revision.

Limitations and Conditions:

The installer must determine the compatibility of this modification with previously approved modifications.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: July 20, 1987

Date issued:

Date of issuance: August 5, 1987

Date amended:



By direction of the Administrator

W. F. Horn (Signature)

Manager, Chicago Aircraft Certification Office
ACE-115C, Central Region, FAA
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA603GL

This certificate, issued to Knots 2U, Inc.
P. O. Box 589
Harbor Springs, MI 49740

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 3 of the Civil Air Regulations. (See Type Certificate Data Sheet No. 2A13 for complete certification basis).

Original Product — Type Certificate Number: 2A13

Make: Piper

Model: PA-28-140, PA-28-150, PA-28-160,
PA-28-180, PA-28-235, PA-28R-180,
PA-28R-200

Description of Type Design Change:

Installation of Aileron, Flap, and Stabilator Gap Seals in accordance with the Knots 2U, Inc. PA-28 Straight Wing Models Gap Seals Installation Manual, issued May 5, 1984, revised May 30, 1990, or later FAA approved revisions.

Limitations and Conditions:

1. Any or all combinations of the above items are approved.
2. This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will introduce no adverse effect upon the airworthiness of that aircraft.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: February 24, 1982

Date issued: July 21, 1982

Date of issuance: April 12, 1982

Date amended: August 31, 1982;
November 26, 1984; October 11, 1990



By direction of the Administrator

Charles L Smalley
(Signature)

for Donald P. Michal, Manager
Chicago Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

KNOTS 2 U, I N C .*SECTION 8.0 GAP SEAL MAINTENANCE MANUAL*PART A. INSPECTION

1. Daily inspection at preflight to ensure there is no bending of the controls, bent gap seals, abrading of rivets or control surfaces, or broken parts.
2. When aircraft has been stored outside during snow or freezing conditions, a careful inspection should be made of areas behind and under the seals for ice accumulations. If ice is found, which cannot be removed by careful brushing with seal held slightly away from control *surface, the aircraft should be deiced or defrosted.
3. 100 hour inspections are suggested to check for abrading of the control surfaces and/or rivet heads, wear of the gap seals, and peeling of the Teflon coating. Check for loose rivets and/or other gap seal attachment hardware.

PART B. MAINTENANCE

1. There are no special tools required to maintain the seals. Any tools needed are basic hand tools.
2. Maintenance of the gap seals is to keep the seal surface clean of oil and dirt and the edge of the seal touching the control surface smoothly. If the gap seal appears to be abrading the control surface 3M 5490 Teflon tape, or equivalent, may be applied to gap seal to act as a wear surface. The Teflon tape should be applied before further flight to prevent control wear.
3. If upon installation, or through wear, there is a warp in the seal, or it lies unevenly, you may drill a #40 size hole and cut the seal to the hole in a direction 90 degrees to the trailing edge of the seal. Drill the hole in the center of the warp, 1/2 inch from the trailing edge of the surface to which the seal is attached. The cut in the seal should be trimmed to give a slot 1/16th inch wide, with parallel edges. The slots should be no closer than 6 inches, to each other, or the end of the seal. Refer to detail on next page.
4. When washing aircraft, care should be taken to brush along the length of the seal rather than perpendicular to, or across the seal.
5. When aircraft is painted care should be taken to prevent paint, paint remover or solvents from contacting the Teflon. If Teflon is damaged, use the maintenance procedures with 3M 5490 tape, or equivalent.

PART C. BALANCING

1. If any change is done to the horizontal stabilator; if Teflon tape is added, if slots are cut to relieve warping, or if the seals are painted, the stabilator must be re-balanced.

PART D. CRACKING, DEFECTS, LOOSE RIVETS

1. If cracks are found in a gap seal, stop drill the crack. If there are more than 3 cracks in a gap seal, the seal must be replaced.
2. If the Teflon peels 3M 5490 Teflon tape, or equivalent, may be applied.
3. If there are excessive bends or kinks in the seal, and the air flow over the control surface is disturbed, the seal must be replaced.
4. If seal rivets become loose you may drill the rivets and replace with the next size rivet.

F. A. A

APPROVED

OCT 11 1990
57m AL120C
CHICAGO AIRCRAFT
CERTIFICATION OFFICE
CENTRAL REGION

SECTION 8. =MAINTENANCE MANUAL=

ART A. INSPECTION

1. There are no special tools required to maintain the fairings. Any tools needed are basic hand tools.
2. Daily inspection at preflight to check security of Flap Hinge Fairing to aircraft attachment. Check for binding during flap operation. Also, check for cracks in the Flap Hinge Fairing.
3. Remove Flap Hinge Fairing, during annual or 100 hour inspections, to inspect flap hinge.

ART B. MAINTENANCE

1. Keep slot of Hinge Fairing clear of obstructions. If the aircraft is painted and paint stripper is used the fairings must not be permitted to contact the stripper. Removing the fairings during stripping may be the simplest way to protect them from damage.

ART C. CRACKING OR DEFECTS

1. If any cracks are found on Flap Hinge Fairing, stop drill the crack.
2. If any crack exceeds 1/2 inch in length; or, if a crack runs from an attachment hole to the outer edge of the flange on the Flap Hinge Fairing, the Fairing must be replaced.

K N O T S 2 U , I N C .

E A A
A P P R O V E D

NOV 2 1988
SPM ACE-120C
CHICAGO AIRCRAFT
CERTIFICATION OFFICE
CENTRAL REGION

SECTION 12. =MAINTENANCE MANUAL=

KNOTS 2 U , I N C .

WING ROOT FAIRINGS
P I P E R P A - 2 8

M A I N T E N A N C E M A N U A L

1. Before each flight, visually inspect the fairings for loose screws and proper fit.
2. Upon each annual or 100 hour inspection, inspect the fairings for proper fit, loose screws, cracks or separated plies of the fiberglass.
3. If any of the above discrepancies are found; repair, using proper maintenance techniques and procedures, as outlined in FAR 43.13-1A & 2

E A A

A P P R O V E D

AUG 0 1 1988

8/1/88 ACE-1200
CHICAGO AIRCRAFT

**CERTIFICATION OFFICE
CENTRAL REGION**

P.T. AERO SERVICE, INC.

Hanger 1, T.F. Green Airport

Warwick, R.I. 02886

401-737-7788

REVISED WEIGHT AND BALANCE DATA

Aircraft Make and Model: PA-28-180

Aircraft Serial Number: 28-7505222

Aircraft Registration Number: N 1185X

Date of Revision: 6 January 92

Date of Previous Revision: 12 March 90

Aircraft:	Weight	Arm	Moment
	1515.19	86.56	131156.97
Installed:	1518.93	86.5	131389.01
1. Flap Hinge Fairing	1.6	133	212.80
2. Aileron gap seals	0.22	129	28.38
3. Flap Gap Seals	0.27	129	34.83
4. Wing Root Fairings	1.14	64.65	73.70
5. Flap/Fuselage Gap Seals	0.03	136	4.08
<hr/>			
New Empty Weight:	1522.19		
New Center of Gravity:		86.54	
New Moment ⁰			131742.80

Gross Weight: (Normal) 2450.00 lbs.

Gross Weight: (Utility) 1950.00 lbs.

New Useful Load: (Normal) 927.81 lbs.

(Utility) 427.81 lbs.

Note:

It is the responsibility of the pilot-in-command to insure that the aircraft is properly loaded using the above weight and balance data and the owners manual loading charts.

William Brearton

William Brearton A & P 36482340

*Superseded
7-29-93*

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY

OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE Piper	MODEL PA-28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK N 1185X
2. OWNER	NAME (As shown on registration certificate) Halzel, Lenny	ADDRESS (As shown on registration certificate) 144 President Ave. Providence, RI 02906

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION				5. TYPE	
UNIT	MAKE	MODEL	SERIAL NO.	REPAIR	ALTERATION
AIRFRAME (As described in item 1 above)				X
POWERPLANT					
PROPELLER					
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS	B. KIND OF AGENCY	C. CERTIFICATE NO.
William Brearton 52 Benedict Rd. Warwick, RI 02888	<input checked="" type="checkbox"/> U.S. CERTIFICATED MECHANIC	36482340 A & P
	<input type="checkbox"/> FOREIGN CERTIFICATED MECHANIC	
	<input type="checkbox"/> CERTIFICATED REPAIR STATION	
	<input type="checkbox"/> MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 6 January, 1992	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>William Brearton</i> William Brearton
-------------------------	---

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA FLT. STANDARDS INSPECTOR	MANUFACTURER	<input checked="" type="checkbox"/> INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE OF APPROVAL OR REJECTION 6 Jan 92	CERTIFICATE OR DESIGNATION NO. 1522270	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>Harold Thistlewaite</i> Harold Thistlewaite		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed on N1185X:

6 JAN 92

1. Left and right aileron and flap gap seals in accordance with STC # SA603GL and installation manual provided by knots 2U, Inc.
2. Left and right flap hinge fairings in accordance with STC # SA1216GL and installation manual provided by knots 2U, Inc.
3. Left and right flap hinge fairings in accordance with STC # SA119GL and installation manual supplied by knots 2U, Inc.

Weight and Balance Revised.

END

ADDITIONAL SHEETS ARE ATTACHED

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY

OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE PIPER	MODEL PA-28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK USA N1185X
2. OWNER	NAME (As shown on registration certificate) LENNY HALZEL	ADDRESS (As shown on registration certificate) 144 PRESIDENT AVE PROVIDENCE, RI 02906

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION

5. TYPE

UNIT	MAKE	MODEL	SERIAL NO.	5. TYPE	
				REPAIR	ALTERATION
AIRFRAME	As described in item 1 above				XX
POWERPLANT					
PROPELLER					
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS CARROLL AVIONICS, INC. Hangar South Complex Tom Harvey Rd. Westerly, RI. 02891	B. KIND OF AGENCY	C. CERTIFICATE NO. UQ5R258N
	U.S. CERTIFICATED MECHANIC	
	FOREIGN CERTIFICATED MECHANIC	
	CERTIFICATED REPAIR STATION	
	MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 3-12-90	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>[Signature]</i>
------------------------	--

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA FLT. STANDARDS INSPECTOR	MANUFACTURER	INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	

DATE OF APPROVAL OR REJECTION 3-12-90	CERTIFICATE OR DESIGNATION NO. 2377051	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>[Signature]</i>
---	--	--

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

- 1) Installed of Apollo 604TCA Loran C system according to manufacturers instructions # 560-0023 Rev 03. Power to unit is supplied through a 20-gauge Tefzel wire, protected by a P&B M581C4C4C12A-5 amp Circuit Breaker and connected to the avionics buss. Unit is panel mounted in R/M radio stack.
- 2) All interface wiring is Tefzel, and all mounting hardware is stainless-steel.
- 3) Structural mounting of all units is sufficient to ensure the restraint of the equipment when subjected to the emergency landing load appropriate to the aircraft category. Refer to updated "Weight and Balance Data and Supplemental Equipment List" this date for station locations of installed equipment.
- 4) Full electric load does not exceed 80% of the electrical system capacity using calculated method referenced in AC 43.13-1A Chap 11, Section 2, Para 428.
- 5) Installation performed in accordance with AC 43.13-1A, Chapter 15, Sections 1 thru 6, Paragraphs 747 thru 853 and AC 43.13-2A, Chapters 2 & 3, paragraphs 21 thru 44.
- 6) Ground and flight tests performed in accordance with AC 20-121, Section 6, Part A, Paragraph 1.
- 7) No flight is predicated upon the use of this Loran C.
- 8) Aircraft is placarded "WFR USE ONLY OF LORAN C".
- 9) Manufacturer's operating instructions provided to aircraft owner.
- 10) Aircraft "Weight and Balance Data, and Supplemental Equipment List" revised.

>>>***** END *****<<<

ADDITIONAL SHEETS ARE ATTACHED

CARROLL AVIONICS, INC.

HANGAR SOUTH COMPLEX TOM HARVEY RD. WESTERLY, RI. 02891 (401) 348-1080 FAA# UQ5R258N

**REVISED WEIGHT AND BALANCE DATA
AND SUPPLEMENTAL EQUIPMENT LIST**

AIRCRAFT MAKE AND MODEL: PIPER PA-28-180
Aircraft Serial Number: 28-7505222
Aircraft Registration Number: N1185X
Date of this Revision: 03-12-90
Date of Previous Revision: 04-27-88

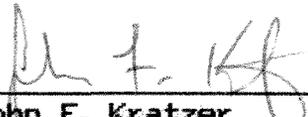
	WEIGHT	ARM	MOMENT
Aircraft:	1515.19	86.56	131156.97
Installed:			
1) APOLLO 604TCA LORAN C	3.18	58.00	184.44
2) APOLLO A-16 ANTENNA	0.56	83.00	47.60

New Empty Weight: 1518.93
 New Center of Gravity: 86.50
 New Moments: 131389.01

Gross Weight: (Normal) 2450.00 lbs.
 Gross Weight: (Utility) 1950.00 lbs.
 New Useful Load: (Normal) 931.07 lbs.
 (Utility) 431.07 lbs.

NOTE:

It is the responsibility of the pilot-in-command to insure that the aircraft is properly loaded using the above Weight and Balance data and the owners manual loading charts.



 John F. Kratzer

2377051

Inspector for CARROLL AVIONICS, INC.
F.A.A. Certified Repair Station # UQ5R258N

SUPERSEDED
6-1-90



U.S. Department of Transportation
Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020

For FAA Use Only
Office Identification

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Make PIPER	Model PA-28-180
	Serial No. 28-7505222	Nationality and Registration Mark U.S.A. N1185X
2. Owner	Name (As shown on registration certificate) LENNY HALZEL	Address (As shown on registration certificate) 144 PRESIDENT AVE. PROVIDENCE, R.I. 02906

3. For FAA Use Only

(This section is for FAA use only. Do not enter information here.)

4. Unit Identification

Unit	Make	Model	Serial No.	5. Type	
				Repair	Alteration
AIRFRAME	~~~~~ (As described in Item 1 above) ~~~~~				XX
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address CARROLL AVIONICS, INC. Hangar South Complex Tom Harvey Rd. Westerly, RI. 02891	B. Kind of Agency <input type="checkbox"/> U.S. Certificated Mechanic <input type="checkbox"/> Foreign Certificated Mechanic <input checked="" type="checkbox"/> Certificated Repair Station <input type="checkbox"/> Manufacturer	C. Certificate No. E61-29
---	--	-------------------------------------

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date 4-27-88	Signature of Authorized Individual <i>Ernest J. Carroll</i>
------------------------	--

7. Approval for Return To Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA Fit. Standards Inspector	Manufacturer	Inspection Authorization	Other (Specify)
	FAA Designee <input checked="" type="checkbox"/>	Repair Station	Person Approved by Transport Canada Airworthiness Group	

Date of Approval or Rejection 4-27-88	Certificate or Designation No. E61-29	Signature of Authorized Individual <i>Ernest J. Carroll</i>
---	---	--

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

- 1) Installed 3M WX-8 Weather Mapping System in accordance with manufacturer's instructions (78-8041-7921-2).

- 2) Installation performed in accordance with AC 43.13-1A, Chapter 15, Sections 1 thru 6, and AC 43.13-2A, Chapters 2 & 3.

- 3) Aircraft "Weight and Balance Data, and Supplemental Equipment List" revised.

=====END=====

CARROLL AVIONICS, INC.
 HANGAR SOUTH COMPLEX TOM HARVEY RD. WESTERLY, RI. 02891 (401) 348-1080 FAA# E61-29

**REVISED WEIGHT AND BALANCE DATA
 AND SUPPLEMENTAL EQUIPMENT LIST**

AIRCRAFT MAKE AND MODEL: PIPER PA-28-180
 Aircraft Serial Number: 28-7505222
 Aircraft Registration Number: N 1185X
 Date of this Revision: 04-27-88
 Date of Previous Revision: 01-29-88

	WEIGHT	ARM	MOMENT
Aircraft:	1511.19	86.49	130704.97

Installed:

1) 3M WX-B DISPLAY	2.00	58.00	116.00
2) 3M WX-B ANTENNA	2.00	168.00	336.00

New Empty Weight:	1515.19		
New Center of Gravity:		86.56	
New Moment:			131156.97

Gross Weight: (Normal) 2450.00 lbs.
 Gross Weight: (Utility) 1950.00 lbs.
 New Useful Load: (Normal) 934.81 lbs.
 (Utility) 434.81 lbs.

NOTE:
 It is the responsibility of the pilot-in-command to insure that the aircraft is properly loaded using the above Weight and Balance data and the owners manual loading charts.

Ernest J. Carroll

Ernest J. Carroll 2377804 Inspector for CARROLL AVIONICS, INC.
 F.A.A. Certified Repair Station # E61-29

*SUPERSEDED
 4-27-88*

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY
OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE PIPER	MODEL PA-28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK U.S.A. N 1185X
2. OWNER	NAME (As shown on registration certificate) LENNARD HALZEL	ADDRESS (As shown on registration certificate) 144 PRESIDENT AVE PROVIDENCE, RI 02906

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION				5. TYPE	
UNIT	MAKE	MODEL	SERIAL NO.	REPAIR	ALTERATION
AIRFRAME	***** (As described in item 1 above) *****				XX
POWERPLANT					
PROPELLER					
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS CARROLL AVIONICS, INC. Hangar South Complex Tom Harvey Rd. Westerly, RI. 02891	B. KIND OF AGENCY	C. CERTIFICATE NO. E61-29
	U.S. CERTIFICATED MECHANIC	
	FOREIGN CERTIFICATED MECHANIC	
	XX CERTIFICATED REPAIR STATION	
	MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 1-29-88	SIGNATURE OF AUTHORIZED INDIVIDUAL Ernest J. Carroll
-----------------	---

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA PLY. STANDARDS INSPECTOR	MANUFACTURER	INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	X REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE OF APPROVAL OR REJECTION 1-29-88	CERTIFICATE OR DESIGNATION NO. E61-29	SIGNATURE OF AUTHORIZED INDIVIDUAL Ernest J. Carroll		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

1) Installed S-TEC system 50 autopilot according to STC # SA51945W-D dated 04-03-85 REV 2 and S-TEC installation instructions ST-174-50 dated 03-12-85

2) Aircraft flight manual revised with FAA approved flight manual supplement P/N 89487 dated 05-31-84.

3) Installation performed in accordance with AC 43.13-1A, Chapter 15, Sections 1 thru 6, and AC 43.13-2A, Chapters 2 & 3.

4) Aircraft "Weight and Balance Data, and Supplemental Equipment List" revised.

END

ADDITIONAL SHEETS ARE ATTACHED

CARROLL-KRATZER AVIONICS, INC.

HANGAR SOUTH COMPLEX TOM HARVEY RD. WESTERLY, RI. 02891 (401) 348-1080 FAA# E61-29

REVISED WEIGHT AND BALANCE DATA
AND SUPPLEMENTAL EQUIPMENT LIST

AIRCRAFT MAKE AND MODEL: PIPER PA-28-180
 Aircraft Serial Number: 28-7505222
 Aircraft Registration Number: N 1185X
 Date of this Revision: 01-29-88
 Date of Previous Revision: -1-23-85

	WEIGHT	ARM	MOMENT
Aircraft:	1502.40	86.40	129831.00
Removed:			
1) PIPER D.G.	2.60	59.70	155.00
2) PIPER T & B	2.60	59.70	155.00
Installed:			
1) S-TEC 0131 PROGRAMMER	2.80	59.30	166.04
2) S-TEC 0105 ROLL SERVO	2.90	122.20	354.38
3) S-TEC 0107 PITCH SERVO	2.90	119.50	346.55
4) S-TEC 0111 TRANSDUCER	0.19	56.00	10.64
5) S-TEC 6406 D.G.	3.40	58.50	198.90
6) S-TEC 6405 T & B	1.80	59.70	107.46

New Empty Weight: 1511.19
 New Center of Gravity: 86.49
 New Moment: 130704.97

Gross Weight: (Normal) 2450.00 lbs.
 Gross Weight: (Utility) 1950.00 lbs.
 New Useful Load: (Normal) 938.81 lbs.
 (Utility) 438.81 lbs.

*SUPERSEDED
1-29-88*

Ernest J. Carroll Inspector for CARROLL-KRATZER AVIONICS, INC.
 Ernest J. Carroll 2377804 F.A.A. Certified Repair Station # E61-29

WEIGHT AND BALANCE

FOR

CHEROKEE ARCHER

APPLICABLE TO SERIAL NUMBERS 28-7405001 THROUGH 28-7505259

WARNING

EXTREME CARE MUST BE EXERCISED TO LIMIT THE USE OF THIS REPORT TO APPLICABLE AIRCRAFT. THIS REPORT REVISED AS INDICATED BELOW OR SUBSEQUENTLY REVISED IS VALID FOR USE WITH THE AIRPLANE IDENTIFIED BELOW WHEN APPROVED BY PIPER AIRCRAFT CORPORATION. SUBSEQUENT REVISIONS SUPPLIED BY PIPER AIRCRAFT CORPORATION MUST BE PROPERLY INSERTED.

MODEL PA-28-180

AIRCRAFT SERIAL NO. 28-7505222 REGISTRATION NO. N 1185X

WEIGHT AND BALANCE, REPORT NUMBER VB-547 REVISION _____

PIPER AIRCRAFT CORPORATION

APPROVAL SIGNATURE AND STAMP _____

ISSUED: MAY 14, 1973
REVISED: NOVEMBER 21, 1975

REPORT: VB-547
MODEL: PA-28-180

WEIGHT AND BALANCE

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WEIGHT AND BALANCE LOG OF REVISIONS

Revision	Revised Pages	Description and Revision	Approved Date
1	5-11	Revised Propeller Weight and Moment;	Nov. 6, 1973
		Revised Spinner Weight, Arm and Moment.	
	5-13	Added Arm and Moment and Removed Dwg. No. for Nose Wheel.	
	5-14	Revised Voltage Regulator and Battery Weights and Moments.	
	5-16	Revised Toe Brakes Weight and Moment.	
	5-18	Revised Battery Weights and Moment.	
	5-23	Revised Selector Panel and Marker Beacon Weights, Arms and Moments.	
	5-25	Revised Microphones Weights, Arms and Moments.	
1	5-26	Revised Main and Nose Wheel Fairings Weights, Arms and Moments; Revised Left and Right Vert. Adj. Front Seats Weights, Arms and Moments; Added Overhead Vent System and revised Weight and Moment for Ground Vent Blower; Revised Overhead Vent System Weight, Arm and Moment.	<i>W. J. Jernard</i>
	5-27	Added Corrosive Resistant Kit.	
2	Title	Added PAC Approval Form. (NOTE: AIRCRAFT DELIVERED WITH MANUALS PRIOR TO THIS REVISION DO NOT REQUIRE THIS REVISION.)	May 30, 1974 <i>W. J. Jernard</i>
3	5-12	Added Oil Filters; added footnote.	June 17, 1974
	5-14	Added Annunciator Lights and footnote.	
	5-16	Revised Inertia Safety Belts Weights, Moment and Part No.	
	5-17	Added Oil Filter - Lycoming, Vacuum Pump - Airborne, Low Vacuum Regulator Light, and Vacuum Regulator - Airborne #2H3-19; revised Weight and Moment of Vacuum Regulator - Airborne #133A4; added footnotes.	
	5-20	Added Encoding Altimeter and footnote.	
	5-21	Added AutoControl IIIB and footnotes.	
	5-22	Revised King VHF Transceivers and added footnotes.	

WEIGHT AND BALANCE LOG OF REVISIONS (cont)

Revision	Revised Pages	Description and Revision	Approved Date
3 (cont)	5-23	Added footnote.	<i>R. Hamlin</i>
	5-24	Revised UGR-2 Glide Slope; added footnote.	
	5-25	Revised Narco Transponder and King Audio Panel.	
	5-25a	Added Page.	
	5-25b	Added Page.	
	5-26	Revised Inertia Safety Belts Part No.; added Assist Strap (79455)	
4	5-5	Revised equations.	<i>C. R. Rikli</i> Jan. 20, 1975
	5-12	Added O-360-A4M engine.	
	5-15	Revised Engine Clusters dash nos.	
	5-22	Revised King KNI-520 VOR/ILS Indicators' weights and moments.	
	5-27	Added Overhead Vent System (76304-9); added Overhead Vent System with Ground Vent. Blower (76304-10).	
5	Title	Added Applicable Serial Nos.	<i>C. R. Rikli</i> Nov. 21, 1975
	5-12	Removed Ser. Nos. from Engine - Lycoming Model O-360-A4A and Engine - Lycoming Model O-360-A4M; deleted Chrysler alternator.	
	5-18	Revised Rotating Beacon; revised Dwg. No. of Piper Pitch Trim; added footnote.	
	5-26	Added 79591-0 (Left) Vert. Adj. Front Seats; added 79591-1 (Right) Vert. Adj. Front Seats; relocated Overhead Vent Systems (2) to page 5-27.	
	5-27	Added Overhead Vent Systems (2) relocated from page 5-26; added 79337-18 Headrest (Front); added 79337-18 Headrest (Rear); added Stainless Steel Control Cables; relocated Exterior Finish to page 5-28.	
	5-28	Added Exterior Finish from page 5-27.	
6	5-1	Revised introduction.	<i>Hal Fletcher</i> March 26, 1979

WEIGHT AND BALANCE LOG OF REVISIONS (cont)

Revision	Revised Pages	Description and Revision	Approved Date
7	5-1 5-3 5-7	Revised text info. Revised text info. Revised Weight and Balance Data.	<i>Ward Evans</i> March 16, 1984

WEIGHT AND BALANCE

In order to achieve the performance and flying characteristics which are designed into the airplane, it must be flown with the weight and center of gravity (C.G.) position within the approved envelope. The aircraft offers flexibility in loading. However, you cannot fill the airplane, with the maximum number of adult passengers, full fuel tanks and maximum baggage. With the flexibility comes responsibility. The pilot must insure that the airplane is loaded within the loading envelope before he makes a takeoff.

Misloading carries consequences for any aircraft. An overloaded airplane will not take off, climb or cruise as well as a properly loaded one. The heavier the airplane is loaded, the less climb performance it will have.

Center of gravity is a determining factor in flight characteristics. If the C.G. is too far forward in any airplane, it may be difficult to rotate for takeoff or landing. If the C.G. is too far aft, the airplane may rotate prematurely on takeoff or try to pitch up during climb. Longitudinal stability will be reduced. This can lead to inadvertent stalls and even spins; and spin recovery becomes more difficult as the center of gravity moves aft of the approved limit.

A properly loaded aircraft, however, will perform as intended. Before the airplane is licensed, a basic weight and C.G. location is computed. (Basic weight consists of the empty weight of the aircraft plus the unusable fuel and full oil capacity.) Using the basic weight and C.G. location, the pilot can easily determine the weight and C.G. position for the loaded airplane by computing the total weight and moment and then determining whether they are within the approved envelope.

The basic weight and C.G. location for a particular airplane are recorded in the Weight and Balance section of the Airplane Flight Manual. The current values should always be used. Whenever new equipment is added or any modification work is done, the mechanic responsible for the work is required to compute a new basic weight and basic C.G. position and to write these in the aircraft log book. The owner should make sure that it is done.

A weight and balance calculation is necessary in determining how much fuel or baggage can be boarded so as to keep within allowable limits. Check calculations prior to adding fuel to insure against improper loading.

The following pages are forms used in weighing an airplane in production and in computing basic weight, basic C.G. position, and useful load. Note that the useful load includes fuel, oil, baggage, cargo and passengers. Following this is the method for computing takeoff weight and C.G.

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WEIGHT AND BALANCE DATA

WEIGHING PROCEDURE

At the time of licensing, Piper Aircraft Corporation provides each airplane with the licensed empty weight and center of gravity location. This data is on Page 5-7.

The removal or addition of an excessive amount of equipment or excessive airplane modifications can affect the licensed empty weight and empty weight center of gravity. The following is a weighing procedure to determine this licensed empty weight and center of gravity location:

1. PREPARATION

- a. Be certain that all items checked in the airplane equipment list are installed in the proper location in the airplane.
- b. Remove excessive dirt, grease, moisture, foreign items such as rags and tools from the airplane before weighing.
- c. Defuel airplane. Then open all fuel drains until all remaining fuel is drained. Operate engine on each tank until all undrainable fuel is used and engine stops.
- d. Drain all oil from the engine, by means of the oil drain, with the airplane in ground attitude. This will leave the undrainable oil still in the system. Engine oil temperature should be in the normal operating range before draining.
- e. Place pilot and copilot seats in fourth (4th) notch, aft of forward position. Put flaps in the fully retracted position and all control surfaces in the neutral position. Tow bar should be in the proper location and all entrance and baggage doors closed.
- f. Weigh the airplane inside a closed building to prevent errors in scale readings due to wind.

2. LEVELING

- a. With airplane on scales, block main gear oleo pistons in the fully extended position.
- b. Level airplane (see diagram) deflating nose wheel tire, to center bubble on level.

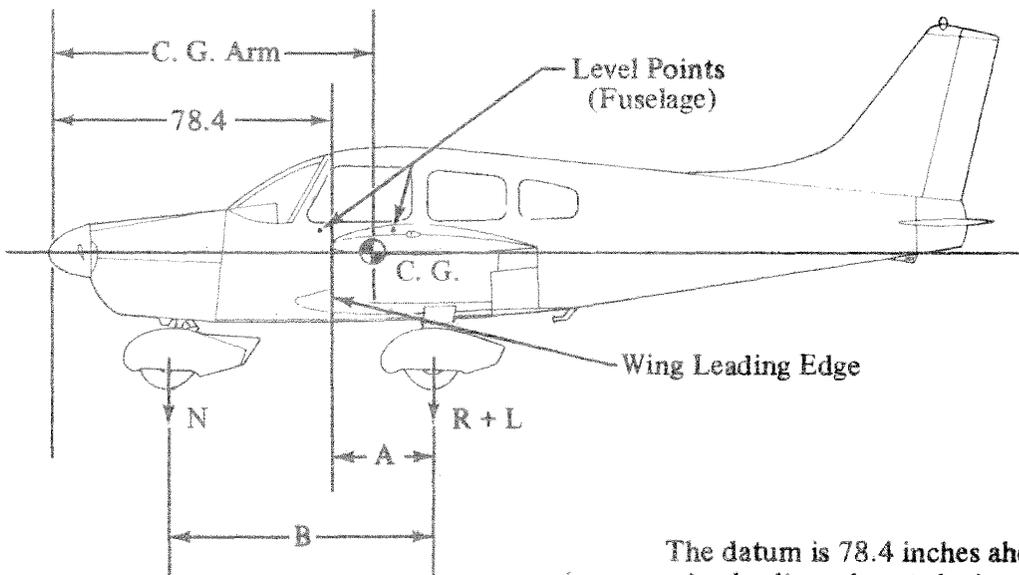
3. WEIGHING - AIRPLANE EMPTY WEIGHT

- a. With the airplane level and brakes released, record the weight shown on each scale. Deduct the tare, if any, from each reading.

Scale Position and Symbol	Scale Reading	Tare	Net Weight
Nose Wheel (N)			
Right Main Wheel (R)			
Left Main Wheel (L)			
Airplane Empty Weight, as Weighed (T)			

4. EMPTY WEIGHT CENTER OF GRAVITY

- a. The following geometry applies to the PA-28-180 airplane when airplane is level (See Item 2).



The datum is 78.4 inches ahead of the wing leading edge at the intersection of the straight and tapered section.

A =

B =

- b. Obtain measurement "A" by measuring from a plumb bob dropped from one wing leading edge, at the intersection of the straight and tapered section, horizontally and parallel to the airplane centerline, to the main wheel centerline.
- c. Obtain measurement "B" by measuring the distance from the main wheel centerline, horizontally and parallel to the airplane centerline, to each side of the nose wheel axle. Then average the measurements.
- d. The empty weight center of gravity (as weighed including optional equipment and undrainable oil) can be determined by the following formula:

$$C.G. \text{ Arm} = 78.4 + A - \frac{B(N)}{T}$$

$$C. G. \text{ Arm} = 78.4 + (\quad) - \frac{(\quad) (\quad)}{(\quad)} = \quad \text{inches}$$

5. LICENSED EMPTY WEIGHT AND EMPTY WEIGHT CENTER OF GRAVITY

	Weight	Arm	Moment
Empty Weight (as weighed)			
Unusable Fuel (13-1/3 pints)	+10.0	103.0	+1030
Licensed Empty Weight			

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WEIGHT AND BALANCE DATA

MODEL PA-28-180 CHEROKEE

Airplane Serial Number _____

Registration Number _____

Date _____

AIRPLANE EMPTY WEIGHT

Item	Weight (Lbs)		× C. G. Arm (Inches Aft of Datum)	= Moment (In-Lbs)
	Actual	Computed		
*Empty Weight				
Unusable Fuel (13-1/3 pints)	10.0		103.0	1030
Standard Empty Weight				
Optional Equipment				
Licensed Empty Weight				

*Empty weight is defined as dry empty weight (including paint and hydraulic fluid) plus 1.8 lbs undrainable engine oil.

AIRPLANE USEFUL LOAD

$$(\text{Gross Weight}) - (\text{Licensed Empty Weight}) = \text{Useful Load}$$

Normal Category: (2450 lbs) - (lbs) = lbs

Utility Category: (1950 lbs) - (lbs) = lbs

THIS LICENSED EMPTY WEIGHT, C.G. AND USEFUL LOAD FOR THE AIRPLANE AS LICENSED AT THE FACTORY. REFER TO APPROPRIATE AIRCRAFT RECORD WHEN ALTERATIONS HAVE BEEN MADE.

C. G. RANGE AND WEIGHT INSTRUCTIONS

1. Add the weight of all items to be loaded to the licensed empty weight.
2. Use the loading graph to determine the moment of all items to be carried in the airplane.
3. Add the moment of all items to be loaded to the licensed empty weight moment.
4. Divide the total moment by the total weight to determine the C.G. location.
5. By using the figures of Item 1 and Item 4, locate a point on the C.G. range and weight graph. If the point falls within the C.G. envelope, the loading meets the weight and balance requirements.

SAMPLE LOADING PROBLEM (Normal Category)

	Weight (Lbs)	Arm Aft Datum (Inches)	Moment (In-Lbs)
Licensed Empty Weight			
Oil (8 quarts)	15	27.5	413
Pilot and Front Passenger	340	80.5	27370
Passengers, Aft* (Rear Seat)	340	118.1	40154
Fuel (48 Gal. Maximum)		95.0	
Baggage*		142.8	
Total Loaded Airplane			

The center of gravity (C.G.) of this sample loading problem is at _____ inches aft of the datum line. Locate this point () on the C.G. range and weight graph. Since this point falls within the weight - C.G. envelope, this loading meets the weight and balance requirements.

IT IS THE RESPONSIBILITY OF THE PILOT AND AIRCRAFT OWNER TO INSURE THAT THE AIRPLANE IS LOADED PROPERLY.

*Utility Category Operation - No baggage or aft passengers allowed.



WAUKEGAN MEMORIAL AIRPORT
 3550 N. McAree
 Waukegan, Illinois 60087

312-336-6075

MINOR ALTERATION AND EQUIPMENT LIST REVISION

AIRCRAFT MAKE PIPER N- 1185X DATE 1/23/85
 MODEL PA-28-180 SERIAL NO. 28-7505222

DESCRIPTION OF WORK ACCOMPLISHED:

INSTALLED NARCO AR-500 ENCODER. SEE FAA 337 THIS DATE FOR APPROVAL INFORMATION.

WEIGHT AND BALANCE - EQUIPMENT LIST REVISION

AIRCRAFT EMPTY WEIGHT	1501.4	86.4	129779
INSTALL AR-500	<u>1.0</u>	52.0	<u>52</u>
	1502.4		129831

*SUPERSEDED
1-23-85*

NEW EMPTY WEIGHT 1502.4 LBS.

NEW USEFUL LOAD 947.6 LBS.

NEW EMPTY WEIGHT C.G. 86.4 INCHES

Shoy French

 AUTHORIZED SIGNATURE
 CRS C.O. 3-46

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY
OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE PIPER	MODEL PA-28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK USA N 1185X
2. OWNER	NAME (As shown on registration certificate) Howell, Willis T.	ADDRESS (As shown on registration certificate) 13202 S. Cicero Midlothian, IL 60445

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION

UNIT	MAKE	MODEL	SERIAL NO.	5. TYPE	
				REPAIR	ALTERATION
AIRFRAME	***** (As described in item 1 above) *****				X
POWERPLANT					
PROPELLER					
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS Waukegan Avionics 3550 N. McAree Waukegan, IL 60087	B. KIND OF AGENCY		C. CERTIFICATE NO. C03-46
	<input type="checkbox"/>	U.S. CERTIFICATED MECHANIC	
	<input type="checkbox"/>	FOREIGN CERTIFICATED MECHANIC	
	<input checked="" type="checkbox"/>	CERTIFICATED REPAIR STATION	
	<input type="checkbox"/>	MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 1/23/85	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>Serge French</i>
-----------------	---

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA FLT. STANDARDS INSPECTOR	MANUFACTURER	INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	<input checked="" type="checkbox"/> REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE OF APPROVAL OR REJECTION 1/23/85	CERTIFICATE OR DESIGNATION NO.	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>Serge French</i>		

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

INSTALLED NARCO AR-500 ENCODER, TSOC88 MATED WITH NARCO AT50A,
TSOC74C CLASS I.

FULLY CHECKED ALTITUDE REPORTING SYSTEM IN ACCORDANCE WITH
AC 43-6A APPENDIX 1. SEE LOGBOOK ENTRY DATED 1/23/85 FOR COM-
PLIANCE WITH FAR 91.36, 91.171, and 91.172.

-----END-----

ADDITIONAL SHEETS ARE ATTACHED

ELGIN PHONE
741-6510
CHICAGO PHONE
626-6436
Area Code 312



P. O. BOX 557 - ELGIN, ILLINOIS 60120

COMPLETE
AIR CHARTER SERVICE
AIRFRAME & POWERPLANT SERVICE
AIRPORT FACILITIES

MINOR ALTERATION AND EQUIPMENT LIST REVISION

AIRCRAFT MAKE Piper N- 1185X DATE 9/4/81
MODEL PA28-180 SERIAL NO. 28-7505222

DESCRIPTION OF WORK ACCOMPLISHED: Installed Narco Com-810, Nav-825, ADF-141, CP-136M & DME-890, cables & antennas. All work done in accordance with manufactures specifications, drawings and A.C. 43.13-1A & A.C. 43.13-2. Electrical load does not exceed electrical generating capabilities.

WEIGHT AND BALANCE - EQUIPMENT LIST REVISION

AIRCRAFT EMPTY WEIGHT	1480.5	86.6	128262
INSTALLED			
Com-810	3.5	57.5	201
Nav-825	3.3	57.5	190
ADF-141	2.5	58.5	146
DME-890	3.9	57.5	224
ID-825	1.0	60.0	60
CP-136M	2.0	59.0	118
ADF-141 Indicator	1.3	61.0	79
Loop Antenna	1.6	162.0	259
Loop Cable	.6	105.5	63
Sense Antenna & Cable	.4	147.5	59
#2 VHF Com Antenna	.3	192.8	58
#2 Cable	.5	120.9	60
	<u>1501.4</u>		<u>129779</u>

*Superseded
1/23/85*

NEW EMPTY WEIGHT 1501.4 LBS.
NEW USEFUL LOAD 948.6 LBS.
NEW EMPTY WEIGHT C.G. 86.4 INCHES.

George French
AUTHORIZED SIGNATURE
CRS 3250

WEIGHT AND BALANCE DATA

MODEL PA-28-180 CHEROKEE

Airplane Serial Number 28-7505222

Registration Number N1185X

Date 5/23/75

AIRPLANE EMPTY WEIGHT

Item	Weight (Lbs)	×	C. G. Arm (Inches Aft of Datum)	=	Moment (In-Lbs)
*Empty Weight XXXXX Actual Computed	1394.0		85.7		119459
Unusable Fuel (13-1/3 pints)	10.0		103.0		1030
Standard Empty Weight	1404.0		85.8		120489
Optional Equipment	76.5		101.6		7773
Licensed Empty Weight	1480.5		86.6		128262

*Empty weight is defined as dry empty weight (including paint and hydraulic fluid) plus 1.8 lbs undrainable engine oil.

AIRPLANE USEFUL LOAD

(Gross Weight) - (Licensed Empty Weight) = Useful Load

Normal Category: (2450 lbs) - (1480.5 lbs) = 969.5 lbs

Utility Category: (1950 lbs) - (1480.5 lbs) = 469.5 lbs

*Superseded
9/4/81*

THIS LICENSED EMPTY WEIGHT, C. G. AND USEFUL LOAD FOR THE AIRPLANE AS DELIVERED FROM THE FACTORY. REFER TO APPROPRIATE AIRCRAFT RECORD WHEN ALTERATIONS HAVE BEEN MADE.

C. G. RANGE AND WEIGHT INSTRUCTIONS

1. Add the weight of all items to be loaded to the licensed empty weight.
2. Use the loading graph to determine the moment of all items to be carried in the airplane.
3. Add the moment of all items to be loaded to the licensed empty weight moment.
4. Divide the total moment by the total weight to determine the C.G. location.
5. By using the figures of Item 1 and Item 4, locate a point on the C.G. range and weight graph. If the point falls within the C.G. envelope, the loading meets the weight and balance requirements.

SAMPLE LOADING PROBLEM (Normal Category)

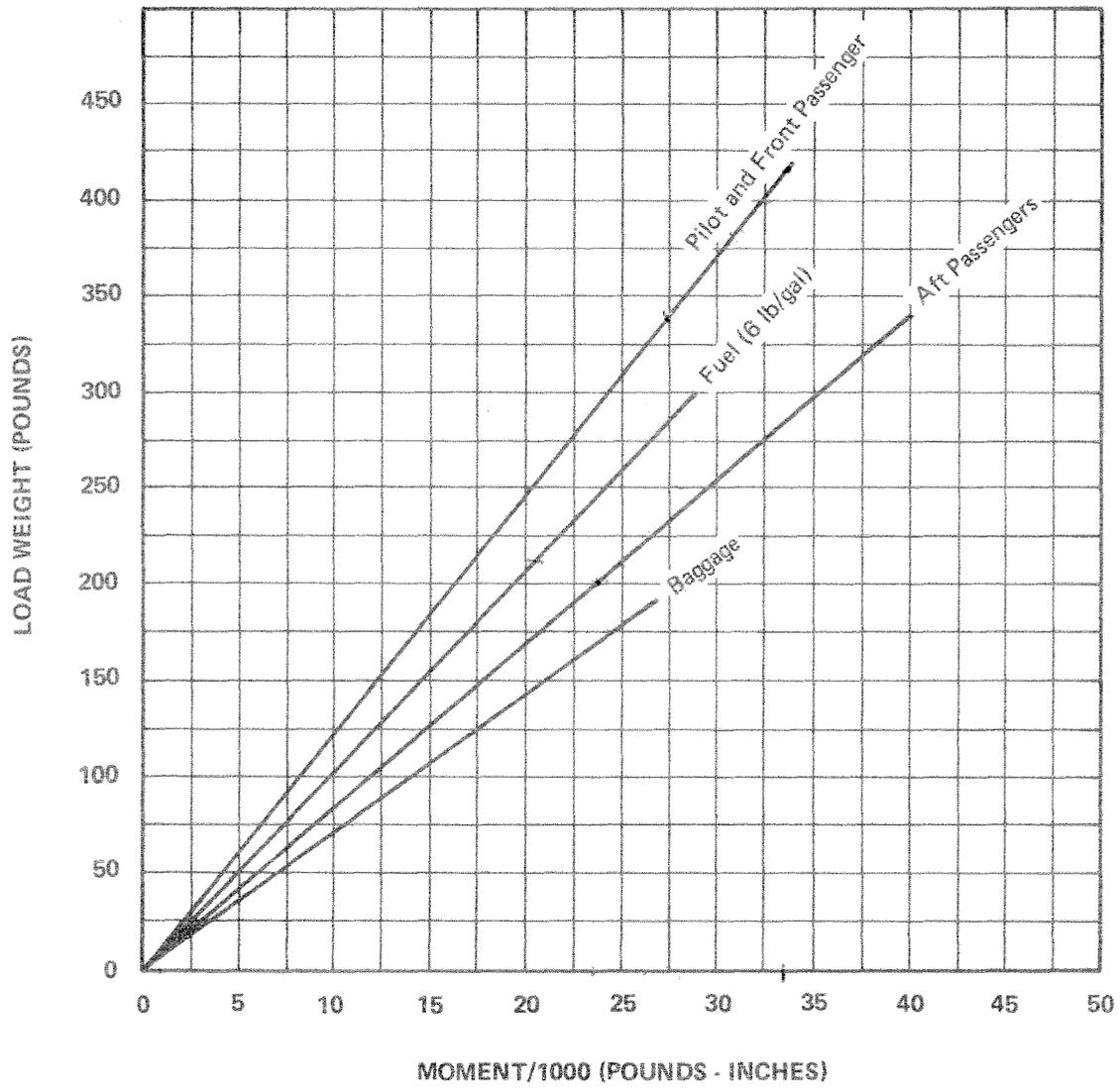
	Weight (Lbs)	Arm Aft Datum (Inches)	Moment (In-Lbs)
Licensed Empty Weight	1480.5	86.6	128262
Oil (8 quarts)	15	27.5	413
Pilot and Front Passenger	340	80.5	27370
Passengers, Aft* (Rear Seat)	340	118.1	40154
Fuel (48 Gal. Maximum)	274.5	95.0	26078
Baggage*		142.8	
Total Loaded Airplane	2450	90.7	222277

The center of gravity (C.G.) of this sample loading problem is at 90.7 inches aft of the datum line. Locate this point (90.7) on the C.G. range and weight graph. Since this point falls within the weight - C.G. envelope, this loading meets the weight and balance requirements.

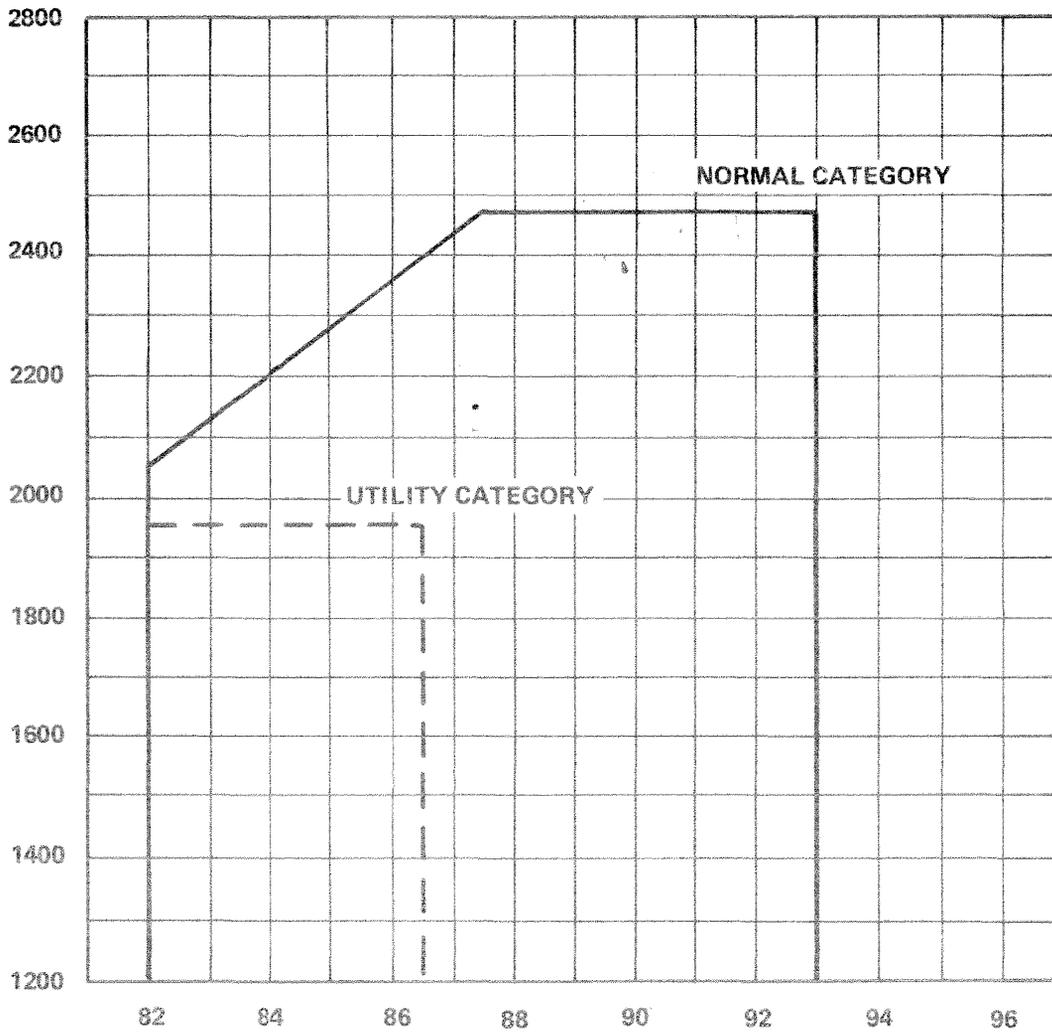
IT IS THE RESPONSIBILITY OF THE PILOT AND AIRCRAFT OWNER TO INSURE THAT THE AIRPLANE IS LOADED PROPERLY.

*Utility Category Operation - No baggage or aft passengers allowed.

LOADING GRAPH



C. G. RANGE AND WEIGHT



EQUIPMENT LIST

The following is a list of equipment which may be installed in the PA-28-180. Items marked with an "X" are items installed when the airplane was delivered by the manufacturer.

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
A. Propeller and Propeller Accessories					
<u>X</u>	Propeller, Sensenich 76EM8S5-0-60 Piper Spec. PS50077-8	39.2	3.8	149	TC P4EA
<u>X</u>	Spinner and Attachment Plates Piper Dwg. 65803, 65804, 65805	4.5	3.4	15	TC 2A13
<u>X</u>	PROPPELLER, McCAULEY 1A170/BAS STC SA 779 NE	36.0	3.8	136.8	

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
B. Engine and Engine Accessories					
_____	Engine - Lycoming Model O-360-A4A Piper Dwg. 62941-14 & -15	288.9	21.1	6096	TC P286
_____	Engine - Lycoming Model O-360-A4M Piper Dwg. 62941-16	285.0	20.9	5964	TC P286
_____	Fuel Pump, Electric Auxiliary, Bendix Model 478360	1.8	36.8	66	TC 2A13
_____	Fuel Pump, Engine Driven, Lycoming Dwg. No. 73297, 74082, 75148 or 75246	1.7	36.3	62	TC P286
_____	Oil Cooler, Piper Dwg. 18622 Harrison *C-8526250	1.9	41.3	78	TC 2A13
_____	Air Filter, Fram Model CA-161 PL or Purolator AFP-2	.9	29.5	27	TC 2A13
_____	Starter - Lycoming 76211 (Prestolite MZ4206)	* 18.0	14.5	261	TC P286
_____	Oil Filter - Lycoming No. 75528 (AC *OF5578770)**	3.3	35.5	117	TC 2A13
_____	Oil Filter - Lycoming *LW-13743 (Champion *CH-48110)**	2.8	35.5	99	TC 2A13

*Included in Engine Weight.

**Serial nos. 7505001 and up

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
C. Landing Gear and Brakes					
✓	Two Main Wheel Assemblies Piper Dwg. 63370-0 & -1	32.3	109.6	3540	TC 2A13
	(a) Cleveland Aircraft Products Wheel Assembly No. 40-86 Brake Assembly No. 30-55				
	(b) Two Main 4-Ply Rating Tires 6.00-6 with Regular Tubes				
✓	One Nose Wheel 6.00-6	12.8	31.0	397	TC 2A13
	(a) Cleveland Aircraft Products Wheel Assembly No. 40-76B (Less Brake Drum)				
	(b) One Nose Wheel 4-Ply Rating Tire 6.00-6 with Regular Tube				

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
	D. Electrical Equipment				
_____	✕ Stall Warning Device, Safe Flight Instrument Corporation No. C52207-4	.2	80.2	16	TSO C54
_____	✕ Voltage Regulator, Wico Electric No. X-16300B	.9	51.9	47	TC 2A13
_____	Battery 12V, 25 A.H., Rebat Model S-25	21.9	168.0	3679	TC 2A13
_____	✕ Overvoltage Relay, Wico Electric No. X16799	.5	55.4	28	TC 2A13
_____	✕ Annunciator Lights *	.9	55.5	50	TC 2A13

*Serial nos. 7505001 and up

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
E. Instruments					
_____ X	Compass - Piper Dwg. 67462	.9	59.9	54	TSO C7c
_____	Airspeed Indicator, Piper Dwg. 63205-2	.6	61.8	37	TSO C2b
_____ X	Tachometer, Piper Dwg. 62177-3	.7	61.2	43	TC 2A13
_____	Altimeter, Piper PS50008-2 or -3	1.0	60.9	61	TSO C10b
_____	Engine Cluster, Piper Dwg. 95241-14	.8	62.4	50	TC 2A13
_____ X	Engine Cluster, Piper Dwg. 95241-11	.8	62.4	50	TC 2A13

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
F. Miscellaneous					
X	Forward Seat Belts (2) .75 lbs. each Piper Spec. PS50039-4-2A	1.5	81.9	123	TSO C22
X	Inertia Safety Belts (2) 0.75 lbs. each Piper Spec. PS50039-4-17	1.5	119.6	179	TC 2A13
X	Rear Seat Belts (2) .70 lbs. each Piper Spec. PS50039-4-3	1.4	123.0	172	TSO C22
X	Rear Seats (2) Piper Dwg. 76170	22.8	124.2	2832	TC 2A13
	Flight Manual	2.6	--	--	TC 2A13
X	Tow Bar, Piper Dwg. 99458	1.3	161.8	210	TC 2A13
X	Toe Brakes (Dual) Piper Dwg. 63473	11.0	49.6	546	TC 2A13

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
	G. Engine and Engine Accessories (Optional Equipment)				
_____	Vacuum Pump, Airborne Mfg. Co., Model No. 10-113A1, 113A5, or 200cc and Drive	5.0	32.0	160	TC 2A13
_____	Oil Filter - Lycoming No. 75528 (AC*OF5578770)*	3.3	35.5	117	TC 2A13
_____	Oil Filter - Lycoming *LW-13743 (Champion *CH-48110)*	2.8	35.5	99	TC 2A13
_____	Vacuum Regulator, Airborne* Mfg. Co., *133A4	.6	52.0	31	TC 2A13
_____	Vacuum Filter, Airborne Mfg. Co., *1J7-1 Piper Dwg. 66673	.3	52.0	16	TC 2A13
_____	Vacuum Pump, Airborne Mfg. Co., Model 211cc and Drive, PAC 79399-0	3.2	32.0	103	TC 2A13
_____	Low Vacuum Annunciator Light**	Neglect			TC 2A13
_____	Vacuum Regulator, Airborne** Mfg. Co. *2H3-19	.5	52.0	26	TC 2A13

*Serial nos. 7405001 through 7405290
**Serial nos. 7505001 and up

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
	H. Electrical Equipment (Optional Equipment)				
x	Rotating Beacon <i>Removed When HC C 10 Aug 09</i>	1.5	263.4	395	TC 2A13
	Landing Light, G.E. Model 4509	.5	13.1	7	TC 2A13
	Navigation Lights (2) Grimes Model A1285 (Red and Green)	.4	106.6	43	TSO C30b
	Navigation Light (Rear) (1) Grimes Model 2064 (White)	.2	281.0	56	TSO C30b
	Battery 12V, 35 A.H., Rebat R-35 (Weight 27.2 lbs.)	5.3	* 168.0	890	TC 2A13
	Cabin Light, Piper Dwgs. 66632-0 & 95229-0	.3	99.0	30	TC 2A13
	Cabin Speaker SB-15052 or 6EU 1937, Quincy Speaker Co., Oakton, Indiana	.8	99.0	79	TC 2A13
	Auxiliary Power Receptacle, Piper Dwg. 65647	2.7	178.5	482	TC 2A13
	External Power Cable 62355-2	4.6	142.8	657	TC 2A13
	Piper Pitch Trim, Piper Dwg. 67496-2 or 67496-3**	4.3	155.3	668	TC 2A13
	Heated Pitot Head Piper Dwg. 69041-5	.4	100.0	40	TC 2A13

*Weight and moment difference between standard and optional equipment.

**Serial numbers 28-7505080 through 28-7505259.

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
H. Electrical Equipment (Optional Equipment) (cont)					
Red Strobe Light, Whelen Engineering Co. Piper Dwg. 99033					
_____	Power Supply, Whelen Model HS *A4-12A-14 with Fin Light only	2.3	198.0	455	TC 2A13
_____	Light (Fin Tip)	.4	263.4	105	TC 2A13
_____	Cable	.4	230.7	92	TC 2A13
Red/White Strobe Light, Whelen Engineering Co.					
_____	Power Supply, Whelen Model HD, T3	3.0	198.0	594	TC 2A13
_____	Light (Fin Tip)	.4	263.4	105	TC 2A13
_____	Cable	.4	230.7	92	TC 2A13
_____	Lights (Wing Tip) (2)	.3	106.6	32	TSO C30b
_____	Cables	2.0	115.6	231	TC 2A13

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
I. Instruments (Optional Equipment)					
_____ X	Suction Gauge, Piper Dwg. 99480-0 or -2	.5	62.2	31	TC 2A13
_____ X	Vertical Speed, Piper Dwg. 99010-2, -4 or -5	1.0	60.9	61	TSO C8b
_____ X	Attitude Gyro, Piper Dwg. 99002-2, -3, -4 or -5	2.2	59.4	131	TSO C4c
_____ Δ	Directional Gyro, Piper Dwg. 99003-2, -3, -4 or -5	2.6	59.7	155	TSO C5c
_____ X	Air Temperature Gauge, Piper Dwg. 79316	.2	72.6	15	TC 2A13
_____ X	Clock, Piper Dwg. 99478	.4	62.4	25	TC 2A13
_____ V	Tru-Speed Indicator, Piper Dwg. 62143-2 or -13	(same as Standard Equipment)			TC 2A13
_____ ∞	Turn and Slip Indicator, Piper PS50030-2 or -3	2.6	59.7	155	TSO C3b
_____	Manifold Pressure Gauge, Piper PS50031-3 or -4	.9	60.8	55	TSO C45
_____	Exhaust Gas Temperature, Piper Dwg. 99026	.7	55.4	39	TC 2A13
_____	Encoding Altimeter Piper PS50008-6 or -7	* .9	60.3	54	TSO C10b C88

*Weight and moment difference between standard and optional equipment.

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
J. Autopilots (Optional Equipment)					
	AutoControl III *				
_____	Roll Servo, *1C363-1-183R	2.5	122.2	306	STC SA1406SW
_____	Console, *1C338 (thru S/N 9999)	1.2	60.1	72	STC SA1406SW
_____	Cables	.7	95.5	67	STC SA1406SW
_____	Attitude Gyro, *52D66	2.3	59.4	137	STC SA1406SW
_____	Directional Gyro, *52D54	3.2	59.0	189	STC SA1406SW
_____	Omni Coupler, *1C388	.9	59.3	53	STC SA1406SW
	AutoFlite II				
_____	Roll Servo, *1C363-1-183R	2.5	122.2	306	STC SA1406SW
_____	Cable	.7	93.4	65	STC SA1406SW
_____	Panel Unit, *52D75-3 or -4	2.4	59.4	143	STC SA1406SW
	AutoControl III B **				
_____	Roll Servo *1C363-1-183R	2.5	122.2	306	STC SA1406SW
_____	Console, *1C338 (S/N 10000 & up)	1.0	60.1	60	STC SA1406SW
_____	Cables	.5	95.5	48	STC SA1406SW
_____	Attitude Gyro, *52D66	2.7	59.4	160	STC SA1406SW
_____	Directional Gyro, *52D54	2.9	59.0	171	STC SA1406SW
_____	Omni Coupler, *1C388	1.0	59.3	59	STC SA1406SW

*Serial nos. 7405001 through 7405290

**Serial nos. 7505001 and up

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
K. Radio Equipment (Optional Equipment)					
_____	Narco Mark 16 (VHF Comm/Nav)* Transceiver, Single	7.5	56.9	427	TC 2A13
_____	Transceiver, Dual	15.0	56.9	854	TC 2A13
_____	Narco VOA-50M Omni Converter *	2.1	59.9	126	TC 2A13
_____	Narco VOA-40M Omni Converter *	1.9	59.9	114	TC 2A13
_____	Narco VOA-40 Omni Converter*	1.9	59.9	114	TC 2A13
_____	Narco Comm 10A VHF Transceiver	3.9	57.4	224	TC 2A13
_____	X Narco Comm 11A VHF Transceiver	3.6	57.4	207	TC 2A13
_____	Narco Dual Comm 11A VHF Transceiver	7.1	57.4	408	TC 2A13
_____	Narco Nav 10 VHF Receiver	1.9	58.6	111	TC 2A13
_____	X Narco Nav 11 VHF Receiver	2.8	58.6	164	TC 2A13
_____	Narco Nav 12 VHF Receiver	3.4	58.6	199	TC 2A13
_____	Narco Dual Nav 11 VHF Receiver	5.6	58.6	328	TC 2A13
_____	King KX-175 () VHF Transceiver	9.4	56.6	532	TC 2A13
_____	King KN-73 Glide Slope Receiver	3.2	184.3	590	TC 2 13
_____	King KN-77 VOR/LOC Converter	3.6	183.6	661	TC 2A13
_____	King KNI-520 VOR/ILS Indicator	2.8	60.5	169	TC 2A13
_____	King KX-175 () VHF Transceiver (2nd)	8.6	56.6	487	TC 2A13
_____	King KN-77 VOR/LOC Converter	4.2	183.6	771	TC 2A13
_____	King KNI-520 VOR/ILS Indicator	2.8	60.5	169	TC 2A13

*Serial nos. 7405001 through 7405290

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
	K. Radio Equipment (Optional Equipment) (cont)				
_____	Narco CP-25B/125 Selector Panel *	1.2	55.0	66	TC 2A13
_____	Narco MBT-12-R Marker Beacon	3.1	69.1	214	TC 2A13
_____	Narco Comm 110*	3.0	57.4	172	TC 2A13
_____	Narco Comm 111	3.0	57.4	172	TC 2A13
_____	Narco Nav 110*	1.7	58.6	100	TC 2A13
_____	Narco Nav 111	2.5	58.6	147	TC 2A13
_____	Narco Nav 112	3.3	58.6	193	TC 2A13
	King KX-170/175 () (VHF Comm/Nav)				
_____	Transceiver, Single	7.5	56.6	425	TC 2A13
_____	Transceiver, Dual	15.0	56.6	849	TC 2A13
_____	King KI-201 () VOR/LOC Ind.	2.5	59.6	149	TC 2A13
_____	King Dual KI-201 () VOR/LOC Ind.	5.0	59.9	300	TC 2A13
_____	King KI-214 () VOR/LOC/GS Ind.	3.3	59.9	198	TC 2A13
_____	X Nav Receiving Antenna	.5	265.0	133	TC 2A13
_____	X Cable, Nav Antenna	.9	157.0	141	TC 2A13
_____	X *1 VHF Comm Antenna	.3	157.8	47	TC 2A13
_____	X Cable, Antenna *1 VHF	.4	103.4	41	TC 2A13
_____	X *2 VHF Comm Antenna	.3	192.8	58	TC 2A13
_____	X Cable, Antenna *2 VHF	.5	120.9	60	TC 2A13

*Serial nos. 7405001 through 7405290

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
K. Radio Equipment (Optional Equipment) (cont)					
Anti Static Kit					
_____	1 VHF Comm Antenna	1.0	160.8	161	TC 2A13
_____	Cable * 1 VHF Antenna	0.4	103.4	41	TC 2A13
_____	*2 VHF Comm Antenna	1.0	195.8	196	TC 2A13
_____	Cable * 2 VHF Comm Antenna	0.5	120.9	60	TC 2A13
_____	Low Frequency Antenna	0.5	147.5	74	TC 2A13
_____	Static Wicks	—	—	—	TC 2A13
Narco ADF-31*					
_____	Panel Unit	5.0	58.5	293	TC 2A13
_____	Sensor Unit	2.5	162.7	407	TC 2A13
_____	Sensor Cable	2.3	100.6	231	TC 2A13
_____	Sense Antenna and Cable	.4	150.0	60	TC 2A13
Bendix ADF-T-12C* or Bendix ADF-T-12D*					
_____	Receiver	3.5	59.4	208	TC 2A13
_____	Audio Amplifier	.8	52.4	42	TC 2A13
_____	Servo Indicator	1.7	60.9	104	TC 2A13
_____	Loop Antenna	1.3	160.8	209	TC 2A13
_____	Cable, Interconnecting	2.3	108.0	248	TC 2A13
_____	Sense Antenna and Cable	.4	150.0	60	TC 2A13
King KR-85					
_____	Receiver	4.3	59.4	255	TC 2A13
_____	Servo Indicator	1.2	61.3	74	TC 2A13
_____	Loop Antenna	1.3	161.5	210	TC 2A13
_____	Loop Cable	1.8	108.0	194	TC 2A13
_____	Audio Amplifier	.8	51.0	41	TC 2A13
_____	Sense Antenna and Cable	.4	150.0	60	TC 2A13
PM-1 Marker Beaco.					
_____	Receiver	1.1	121.3	133	TC 2A13
_____	Remote Unit	.3	128.4	39	TC 2A13
_____	Cable	.3	80.0	24	TC 2A13
UGR-2A Glide Slope					
_____	Receiver	2.4	173.8	417	TC 2A13
_____	Cable	1.8	128.0	230	TC 2A13
_____	Antenna	.4	87.4	35	TC 2A13
_____	Cable, Antenna	.5	145.0	73	TC 2A13

*Serial nos. 7405001 through 7405290

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
	K. Radio Equipment (Optional Equipment) (cont)				
X	Narco AT-50A Transponder Panel Unit	* 3.0	57.3	172	TC 2A13
	King KT76/78 Transponder Panel Unit	3.1	58.1	180	TC 2A13
	Antenna and Cable	—	—	—	TC 2A13
	King KMA-20 () Audio Panel	2.8	60.2	169	TC 2A13
	Antenna	.5	116.3	58	TC 2A13
	Cable	.4	87.5	35	TC 2A13
	King KN60C DME Receiver	6.8	56.7	386	TC 2A13
	Antenna	.2	107.1	21	TC 2A13
	Cable, Antenna	0.3	80.6	24	TC 2A13
X	Piper Automatic Locator, Piper Dwg. 99890 Transmitter	1.7	236.2	402	TC 2A13
X	Antenna and Cable	.2	224.4	45	TC 2A13
X	Shelf and Access Plate	.3	235.4	71	TC 2A13
	Audio Selector Panel,** Piper Dwg. 99395-0, -2 or -3	.7	61.3	43	TC 2A13
	Microphone (Dynamic) Piper Dwg. 68856-12	.3	64.9	19	TC 2A13
	Microphone (Carbon), Piper Dwg. 68856-10	.3	64.9	19	TC 2A13
X	Headset, Piper Dwg. 68856-10	.5	60.0	30	TC 2A13
X	Narco DME-890	3.9	57.5	224	
X	Narco CP-136M	2.0	59.0	118	

*Weight includes Antenna and Cable.

**Serial nos. 7405001 through 7405290

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
K. Radio Equipment (Optional Equipment) (cont)					
_____	King KI-213 VOR/LOC/GS Indicator *	2.5	60.4	151	TC 2A13
King KR-86 ADF *					
_____	Receiver	3.9	59.4	232	TC 2A13
_____	Loop Antenna	1.5	161.5	242	TC 2A13
_____	Loop Cable	1.3	108.0	140	TC 2A13
_____	Audio Amplifier	0.8	51.0	41	TC 2A13
_____	Sense Antenna & Cable	0.4	150.0	60	TC 2A13
King KR-86 ADF (2nd)*					
_____	Receiver	3.9	59.4	232	TC 2A13
_____	Loop Antenna	1.5	150.7	226	TC 2A13
_____	Loop Cable	1.3	105.0	137	TC 2A13
_____	Sense Antenna & Cable	3.0	147.5	443	TC 2A13
King KN-65 DME *					
_____	Receiver	7.6	201.6	1532	TC 2A13
_____	Antenna	0.2	107.1	21	TC 2A13
_____	Cable, Antenna	0.3	157.1	47	TC 2A13
_____	Indicator	1.0	62.4	62	TC 2A13
King KN-74 R-Nav *					
_____	Computer	3.7	57.6	213	TC 2A13
_____	Cable Assy.	1.0	53.0	53	TC 2A13
_____	Narco Comm 11B VHF Transceiver *	3.9	57.4	224	TC 2A13
_____	Narco Dual Comm 11B VHF * Transceiver	7.8	57.4	448	TC 2A13
_____	Narco Dual Comm 111 VHF * Transceiver	6.0	57.4	344	TC 2A13
_____	Narco Comm 111B VHF * Transceiver	3.9	57.4	224	TC 2A13
X	Narco Com-810	3.5	57.5	201	
X	Narco Nav-825	3.3	57.5	190	
X	Narco ID-825	1.0	60.0	60	
X	Narco ADF-141	2.5	58.5	146	
X	Narco ADF-141 Ind.	1.3	61.0	79	

*Serial nos. 7505001 and up

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
K. Radio Equipment (Optional Equipment) (cont)					
_____	Narco Dual Comm 111B VHF Transceiver *	7.8	57.4	448	TC 2A13
_____	Narco Nav 14 VHF Receiver *	2.5	57.4	144	TC 2A13
_____	Narco Nav 114 VHF Receiver *	2.5	57.4	144	TC 2A13
_____	Narco UGR-3 Glide Slope * Receiver	2.4	173.8	417	TC 2A13
_____	Cable	1.8	128.0	230	TC 2A13
_____	Antenna	0.4	87.4	35	TC 2A13
_____	Cable, Antenna	0.5	145.0	73	TC 2A13
_____	Narco CP-125 Audio Selector Panel *	2.2	55.0	121	TC 2A13
_____	Narco ADF-140 * Receiver	2.5	58.3	146	TC 2A13
_____	Servo Indicator	1.3	61.0	79	TC 2A13
_____	Loop Antenna	1.6	162.0	259	TC 2A13
_____	Cable, Loop	0.6	105.5	63	TC 2A13
_____	Sense Antenna and Cable	0.4	147.5	59	TC 2A13
_____	Narco Dual ADF-140 * Receivers	5.0	58.3	292	TC 2A13
_____	Dual Needle Indicator	3.5	61.0	214	TC 2A13
_____	Loop Antenna *1	1.6	162.0	259	TC 2A13
_____	Cable, Loop *1	0.6	105.5	63	TC 2A13
_____	Sense Antenna and Cable *1	0.4	143.8	58	TC 2A13
_____	Loop Antenna *2	1.6	150.0	240	TC 2A13
_____	Cable, Loop *2	0.6	93.5	56	TC 2A13
_____	Sense Antenna and Cable *2	3.0	143.8	431	TC 2A13
_____	Remote for Dual Ind.	2.0	185.5	371	TC 2A13
_____	Narco DME-190 * Receiver	5.2	61.8	321	TC 2A13
_____	Antenna	0.3	113.9	34	TC 2A13
_____	Cable, Antenna	0.4	85.6	34	TC 2A13
_____	Microphone (Dynamic)* Piper Dwg. 68856-11	0.6	69.9	42	TC 2A13

*Serial nos. 7505001 and up

CHEROKEE ARCHER

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
L. Miscellaneous (Optional Equipment)					
_____	Fire Extinguisher, Scott Aviation * 42211-00, Piper Dwg. 76167-2	4.6	71.0	327	TC 2A13
_____	Assist Step Piper Dwg. 65384-0	1.8	156.0	281	TC 2A13
_____	Inertia Safety Belts (Rear) (2) 0.8 lbs. each Piper Spec. PS50039-4-14	1.6	140.3	224	TC 2A13
_____	Nose Wheel Fairing Piper Dwg. 65348-2	3.6	36.3	131	TC 2A13
_____	Main Wheel Fairings Piper Dwg. 65237	7.6	113.6	863	TC 2A13
_____	Vert. Adj. Front Seats (Left) Piper Dwg. 76340-0	* 6.6	80.7	533	TC 2A13
_____	(Left) Piper Dwg. 79591-0	* 6.6	80.3	530	TC 2A13
_____	Vert. Adj. Front Seat (Right) Piper Dwg. 76340-1	* 6.8	80.0	544	TC 2A13
_____	(Right) Piper Dwg. 79591-1	* 6.6	79.6	525	TC 2A13
_____	Super Cabin Sound Proofing Piper Dwg. 79030-3	18.1	86.8	1571	TC 2A13
_____	Lighter, 12V Universal *200462	.2	62.9	13	TC 2A13
_____	Assist Strap and Coat Hook Piper Dwg. 62353-5	.2	109.5	22	TC 2A13
_____	Assist Strap, Piper Dwg. 79455	.2	109.5	22	TC 2A13

*Weight and moment difference between standard and optional equipment.

Item	Item	Weight Lbs.	Arm Aft Datum	Moment	Cert. Basis
L. Miscellaneous (Optional Equipment) (cont)					
_____	Overhead Vent System with Ground Ventilating Blower Piper Dwg. 76304-2	13.2	170.4	2249	TC 2A13
_____	Overhead Vent System Piper Dwg. 76304-0	5.6	157.3	881	TC 2A13
_____	Overhead Vent System, Piper Dwg. 76304-9	6.4	159.6	1022	TC 2A13
_____	Overhead Vent System with Ground Ventilating Blower, Piper Dwg. 76304-10	14.0	170.7	2390	TC 2A13
_____	Alternate Static Source	.4	61.0	24	TC 2A13
Calibrated Alternate Static Source					
Placard Required: Yes _____ No _____					
_____	Headrest (2) (Front) Piper Dwg. 99255-3 or 79337-18	2.2	94.5	208	TC 2A13
_____	Headrest (2) (Rear) Piper Dwg. 99255-3 or 79337-18	2.2	132.1	291	TC 2A13
_____	Air Conditioning Installation 99575-0	67.4	102.8	6929	TC 2A13
_____	Zinc Chromate Finish Piper Dwg. 65665	5.0	158.0	790	TC 2A13
_____	Corrosive Resistant Kit	3.0	106.0	318	TC 2A13
_____	Stainless Steel Control Cables	—	—	—	TC 2A13
TOTAL OPTIONAL EQUIPMENT		_____	_____	_____	

CHEROKEE ARCHER

EXTERIOR FINISH

Base Color _____

Registration No. Color _____

Trim Color _____

Type Finish _____

Accent Color _____

LOADING INSTRUCTIONS

**THIS SECTION IS NOT
APPLICABLE TO THIS AIRPLANE**

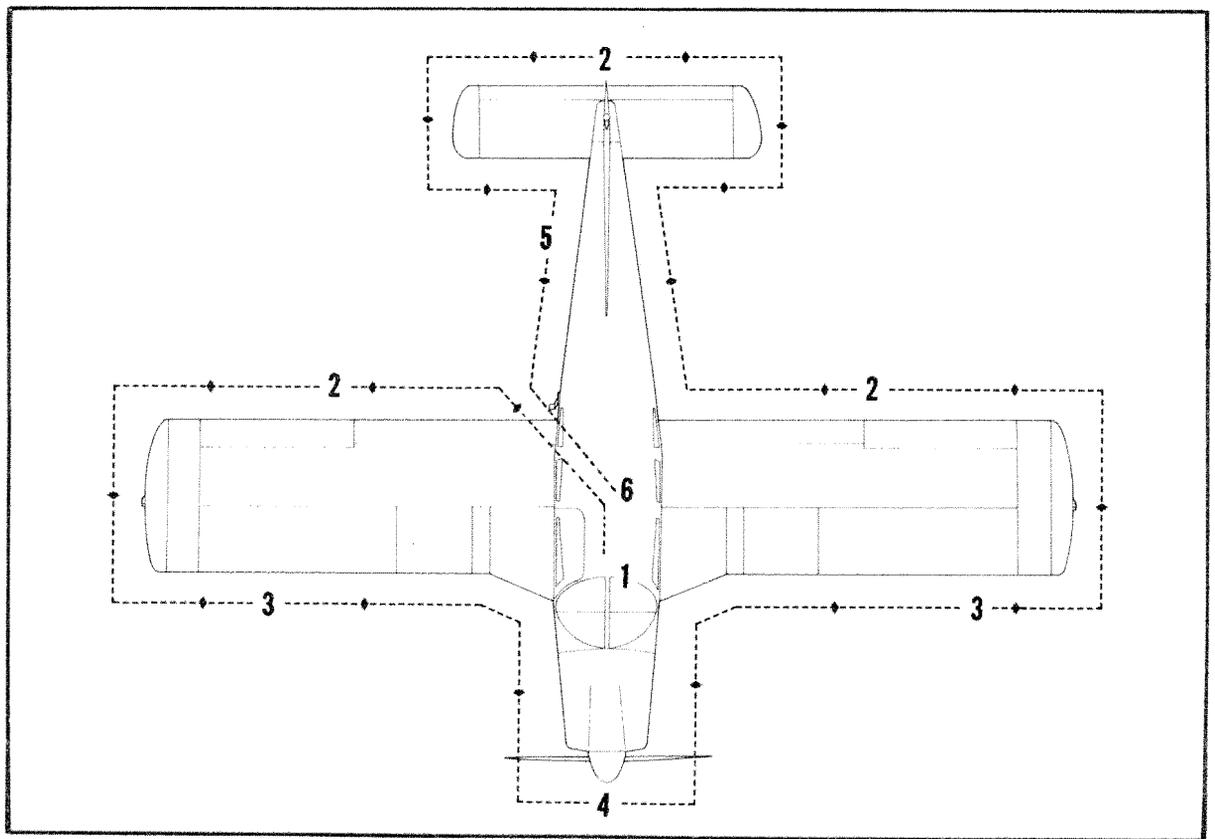
OPERATING INSTRUCTIONS

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OPERATING INSTRUCTIONS

PREFLIGHT

1.
 - a. Release seat belt securing controls.
 - b. Master switch ON.
 - c. Check fuel quantity indicators.
 - d. Master switch and ignition OFF.
2.
 - a. Check for external damage and operational interference of control surfaces or hinges.
 - b. Insure that wings and control surfaces are free of snow, ice or frost.
3.
 - a. Visually check fuel supply and secure caps.
 - b. Drain fuel tank sumps (two), check for water, sediment and proper fuel.
 - c. Drain fuel system sump (left side of aircraft), check for water, sediment and proper fuel.
 - d. Check that fuel system vents are open.
 - e. Check main landing gear shock struts for proper inflation (approximately 4.50 inches showing).
 - f. Check tires for cuts, wear and proper inflation.
 - g. Check brake blocks and discs for wear and damage.



4.
 - a. Check windshield for cleanliness.
 - b. Check propeller and spinner for defects or nicks.
 - c. Check for obvious fuel or oil leaks.
 - d. Check oil level (Insure dipstick is properly seated.)
 - e. Check cowling and inspection covers for security.
 - f. Check nose wheel tire for inflation and wear.
 - g. Check nose gear shock strut for proper inflation (approximately 3.25 inches showing).
 - h. Check for foreign matter in air inlets.
5.
 - a. Stow tow bar and control locks if used.
 - b. Check baggage for storage and security.
 - c. Close and secure the baggage compartment door.
6.
 - a. Upon entering airplane remove seat belt securing control wheel. Check that all primary flight controls operate properly.
 - b. Close and secure cabin door.
 - c. Check that required papers are in order and in the airplane.
 - d. Fasten seat belts and shoulder harness. Check function of inertia reel.

STARTING ENGINE

1. Set parking brake ON.
2. Set the carburetor heat control in the full COLD position.
3. Select the desired tank with fuel selector valve.

STARTING ENGINE WHEN COLD

1. Open throttle approximately 1/4 inch.
2. Turn the master switch ON.
3. Turn the electric fuel pump ON.
4. Move the mixture control to FULL RICH.
5. Engage the starter by rotating magneto switch clockwise and pressing in.
6. When the engine fires, advance throttle to desired setting. If the engine does not fire within five to ten seconds, disengage starter and prime with one to three strokes of the priming pump. Repeat starting procedure.

STARTING ENGINE WHEN HOT

1. Open the throttle approximately 1/2 inch.
2. Turn the master switch ON.
3. Turn the electric fuel pump ON.
4. Put mixture control in FULL RICH.
5. Engage the starter by rotating magneto switch clockwise and pressing in. When the engine fires, move the throttle to desired setting.

STARTING ENGINE WHEN FLOODED

1. Open the throttle full.
2. Turn the master switch ON.
3. Turn the electric fuel pump OFF.
4. Put mixture control in IDLE CUT-OFF.
5. Engage the starter by rotating magneto switch clockwise and pressing in. When the engine fires, advance the mixture control and retard the throttle.

When the engine is firing evenly, advance the throttle to 800 RPM. If oil pressure is not indicated within thirty seconds, stop the engine and determine the trouble. In cold weather it will take a few seconds longer to get an oil pressure indication. If the engine has failed to start, refer to the "Lycoming Operating Handbook, Engine Troubles and Their Remedies."

Starter manufacturers recommend that cranking periods be limited to thirty seconds with a two minute rest between cranking periods. Longer cranking will shorten the life of the starter.

STARTING ENGINE WITH EXTERNAL POWER SOURCE*

An optional feature known as Piper External Power (PEP) allows the operator to use an external battery to crank the engine without having to gain access to the aircraft battery.

The procedure is as follows:

1. Turn aircraft MASTER SWITCH to OFF.
2. Connect RED lead of PEP kit jumper cable to POSITIVE (+) terminal of external 12-volt battery and BLACK lead to NEGATIVE (-) terminal.
3. Insert plug of jumper cable into socket located on aircraft fuselage.
4. Turn aircraft MASTER SWITCH to ON and proceed with NORMAL engine starting technique.
5. After engine has been started, turn MASTER SWITCH to OFF and remove jumper cable plug from aircraft.
6. Turn aircraft MASTER SWITCH to ON and check alternator ammeter for indication of output. DO NOT ATTEMPT FLIGHT IF THERE IS NO INDICATION OF ALTERNATOR OUTPUT.

WARM-UP AND GROUND CHECK

Warm-up the engine at 800 to 1200 RPM for not more than two minutes in warm weather, four minutes in cold weather. Avoid prolonged idling at low RPM as this practice may result in fouled spark plugs. If necessary to hold before takeoff, it is recommended that the engine be idled at 1200 RPM.

The magnetos should be checked at 2000 RPM. The drop on either magneto should not exceed 175 RPM and should be within 50 RPM of the other. Prolonged operation on one magneto should be avoided.

Check vacuum gauge; indicator should read 5" \pm .1" Hg at 2000 RPM.

*Optional equipment

CHEROKEE ARCHER

Check both the oil temperature and pressure. The temperature may be low for some time if the engine is being run for the first time of the day, but as long as the pressure is within limits, the engine is ready for takeoff. Release the parking brake prior to taxiing.

Check the annunciator panel lights with the press-to-test button.*

Carburetor heat should also be checked prior to takeoff to be sure that the control is operating properly and to clear any ice which may have formed during taxiing. Avoid prolonged ground operation with carburetor heat ON as the air is unfiltered.

Operation of the engine driven fuel pump should be checked while taxiing or during pretake-off engine run up by switching off the electric fuel pump and observing fuel pressure. The electric fuel pump should be on during takeoff to prevent loss of power should the engine driven pump fail. The engine is warm enough for takeoff when the throttle can be opened without the engine faltering. For air conditioner ground check refer to page 7-8.

TAKEOFF

Just before takeoff the following items should be checked:

1. Fuel on proper tank
2. Electric fuel pump - on
3. Engine gauges - checked
4. Flaps - set
5. Carburetor heat - off
6. Mixture - set
7. Seat backs - erect
8. Safety Belts/harness - fastened
9. Empty seats - seat belts snugly fastened
10. Trim tab - set
11. Controls - free
12. Door - latched
13. Air conditioner - off

The takeoff technique is conventional for the Cherokee. The tab should be set slightly aft of neutral, with the exact setting determined by the loading of the aircraft. Allow the airplane to accelerate to 50 to 60 MPH, then ease back on the wheel enough to let the airplane fly itself from the ground. Premature raising of the nose or raising it to an excessive angle will result in a delayed takeoff. After takeoff let the aircraft accelerate to the desired climb speed by lowering the nose slightly.

Takeoffs are normally made with flaps up. However, for short field takeoffs, and for takeoffs under difficult conditions such as deep grass or on a soft surface, distances can be reduced appreciably by lowering flaps to 25°.

CLIMB

The best rate of climb at gross weight will be obtained at 85 MPH. The best angle of climb may be obtained at 74 MPH. At lighter than gross weight these speeds are reduced somewhat. For climbing en route a speed of 100 MPH is recommended. This will produce better forward speed and increased visibility over the nose during the climb. The air conditioner may be turned on after all obstacles have been cleared.

*Serial nos. 7505001 and up

STALLS

All controls are effective at speeds down through the stalling speed, and stalls are gentle and easily controlled.

The stall speed chart is at gross weight. Stall speeds at lower weights will be correspondingly less.

STALL SPEED TABLE

Angle of Bank	Flaps 40°	Flaps Retracted
0°	61 MPH	68 MPH
20°	63 MPH	70 MPH
40°	70 MPH	78 MPH
60°	86 MPH	96 MPH
Power Off – Gross Weight 2450 Lbs.		

CRUISING

The cruising speed is determined by many factors, including power setting, altitude, temperature, loading and equipment installed on the airplane.

The normal cruising power is 75% of the rated horsepower of the engine. True airspeeds, which may be obtained at various altitudes and power settings, can be determined from the charts in Performance Charts Section.

Use of the mixture control in cruising flight reduces fuel consumption significantly, especially at higher altitudes. The mixture should be leaned during cruising operation above 5000 feet altitude and at pilot's discretion at lower altitudes when 75% power or less is being used. If any doubt exists as to the amount of power being used, the mixture should be in the FULL RICH position for all operations under 5000 feet.

To lean the mixture, pull the mixture control until the engine becomes rough, indicating that the lean mixture limit has been reached in the leaner cylinders. Then enrich the mixture by pushing the control towards the instrument panel until engine operation becomes smooth.

If the airplane is equipped with the optional exhaust gas temperature (EGT) gauge, a more accurate means of leaning is available to the pilot. For this procedure, refer to the AVCO Lycoming Operator's Manual.

In order to keep the airplane in best lateral trim during cruising flight, the fuel should be used alternately from each tank. It is recommended that one tank be used for one hour after takeoff, then the other tank be used for two hours; then return to the first tank, which will have approximately one and one half hours of fuel remaining if the tanks were full at takeoff. The second tank will contain approximately one half hour of fuel. Do not run tanks completely dry in flight.

ROUGH AIR OPERATION

In keeping with good operating practice used in all aircraft, it is recommended that when turbulent air is encountered or expected the airspeed be reduced to maneuvering speed to reduce the structural loads caused by gusts and to allow for inadvertent speed build-ups which may occur as a result of the turbulence or distractions caused by the conditions.

APPROACH AND LANDING

Before landing check list:

1. Fuel on proper tank
2. Mixture - rich
3. Electric fuel pump on
4. Seat backs - erect
5. Flaps - set (115 MPH)
6. Safety Belts/harness - fastened
7. Air conditioner - off

The airplane should be trimmed to an approach speed of about 85 MPH with flaps up. The flaps can be lowered at speeds up to 115 MPH, if desired, and the approach speed reduced 3 MPH for each additional notch of flaps. Carburetor heat should not be applied unless there is an indication of carburetor icing, since the use of carburetor heat causes a reduction in power which may be critical in case of a go-around. Full throttle operation with carburetor heat on is likely to cause detonation.

The amount of flap used during landings and the speed of the aircraft at contact with the runway should be varied according to the landing surface and conditions of wind and airplane loading. It is generally good practice to contact the ground at the minimum possible safe speed consistent with existing conditions.

Normally, the best technique for short and slow landings is to use full flap and enough power to maintain the desired airspeed and approach flight path. Reduce the airspeed during flareout and contact the ground close to stalling speed. After ground contact hold the nose wheel off as long as possible. As the airplane slows down, drop the nose and apply brakes. There will be less chance of skidding the tires if the flaps are retracted before applying the brakes. Braking is most effective when back pressure is applied to the control wheel, putting most of the airplane weight on the main wheels. In high wind conditions, particularly in strong crosswinds, it may be desirable to approach the ground at higher than normal speeds with partial or no flaps.

STOPPING ENGINE

At the pilot's discretion, the flaps should be raised and the electric fuel pump turned off. After parking, the air conditioner and radios should be turned off and the engine stopped by pulling the mixture control to idle cut-off. The throttle should be left full aft to avoid engine vibration while stopping. Then the magneto and master switches should be turned off and the parking brake set.

AIRSPEED DATA

All airspeeds quoted in this manual are calibrated unless otherwise noted. Calibrated airspeed is indicated airspeed corrected for instrument and position errors. The following table gives the correlation between indicated airspeed and calibrated airspeed if zero instrument error is assumed. This calibration is valid only when flown at maximum gross weight in level flight.

AIRSPEED CORRECTION TABLE

Flaps 0°											
IAS - MPH	60	70	80	90	100	110	120	130	140	150	160
CAS - MPH	68	76	84	92	101	110	119	128	137	147	156
Flaps 40°											
IAS - MPH	60	70	80	90	100	110	120				
CAS - MPH	66	74	82	90	99	109	118				

ENGINE POWER LOSS

The most common cause of engine power loss is mismanagement of the fuel. Therefore, the first step to take after engine power loss is to move the fuel selector valve to the tank not being used. This will often keep the engine running even if there is no apparent reason for the engine to stop on the tank being used.

If changing to another tank does not restore the engine:

1. Check fuel pressure and turn on electric fuel pump if off.
2. Push mixture control to full "RICH."
3. Check ignition switch. Turn to best operating magneto - left, right, or both.

MOORING

The Cherokee should be moved on the ground with the aid of the nose wheel tow bar provided with each plane and secured in the baggage compartment. Tie down ropes may be secured to rings provided under each wing and to the tail skid. The aileron and stabilator controls should be secured by looping the seat belt through the control wheel and pulling it snug. The rudder is held in position by its connections to the nose wheel steering and normally does not have to be secured. The flaps are locked when in the full up position and should be left retracted.

WEIGHT AND BALANCE

It is the responsibility of the owner and pilot to determine that the airplane remains within the allowable weight vs. center of gravity envelope while in flight. For weight and balance data see the Airplane Flight Manual and Weight and Balance form supplied with each airplane.

EMERGENCY LOCATOR TRANSMITTER*

The Emergency Locator Transmitter (ELT) when installed, is located in the aft portion of the fuselage just below the stabilator leading edge and is accessible through a plate on the right side of the fuselage. (On aircraft manufactured prior to mid-1975, this plate is retained by three steel Phillips head screws. On aircraft manufactured from mid-1975 and on, this plate is attached with three slotted-head nylon screws for ease of removal; these screws may be readily removed with a variety of common items such as a dime, a key, a knife blade, etc. If there are no tools available in an emergency the screw heads may be broken off by any means.) The ELT is an emergency locator transmitter which meets the requirements of FAR 91.52. The unit operates on a self-contained battery.

A battery replacement date is marked on the transmitter label. To comply with FAA regulations, the battery must be replaced on or before this date. The battery must also be replaced if the transmitter has been used in an emergency situation or if the accumulated test time exceeds one hour, or if the unit has been inadvertently activated for an undetermined time period.

On the unit itself is a three position selector switch labeled "OFF," "ARM" and "ON." The "ARM" position is provided to set the unit to the automatic position so that it will transmit only after impact and will continue to transmit until the battery is drained to depletion or until the switch is manually moved to the "OFF" position. The "ARM" position is selected when the transmitter is installed at the factory and the switch should remain in that position whenever the unit is installed in the airplane. The "ON" position is provided so the unit can be used as a portable transmitter or in the event the automatic feature was not triggered by impact or to periodically test the function of the transmitter.

Select the "OFF" position when changing the battery, when rearming the unit if it has been activated for any reason, or to discontinue transmission.

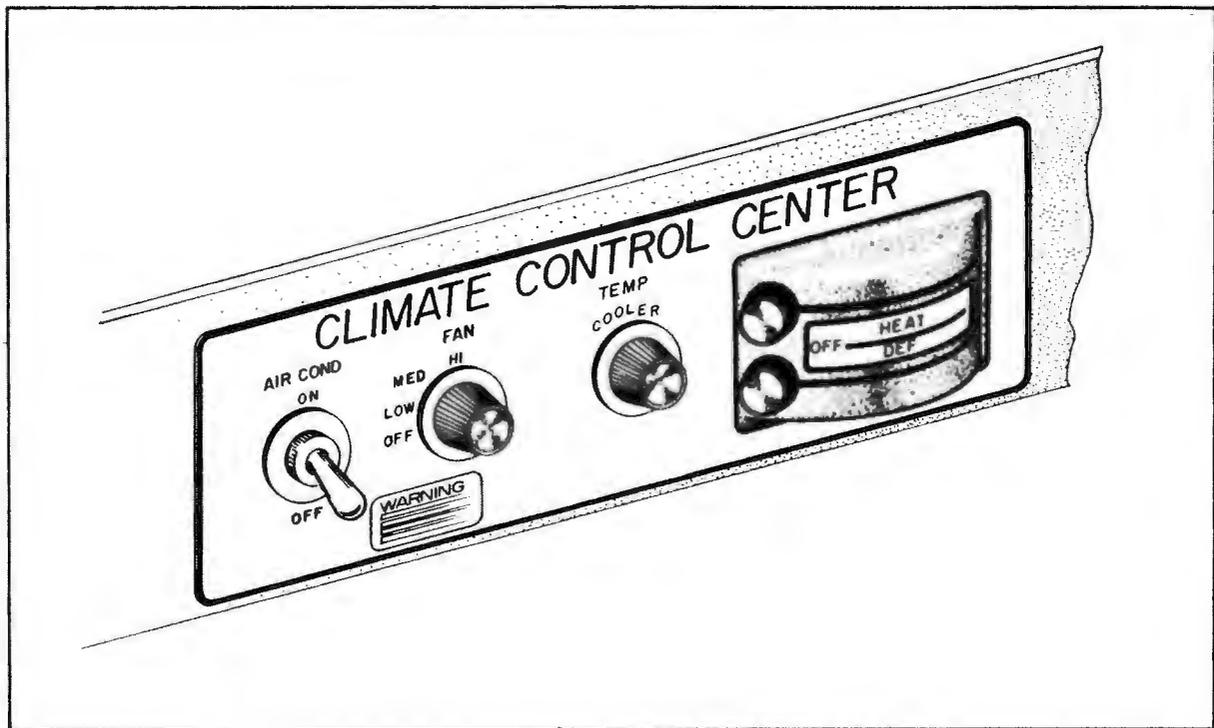
NOTE

If the switch has been placed in the "ON" position for any reason, the "OFF" position must be selected before selecting "ARM." If "ARM" is selected directly from the "ON" position, the unit will continue to transmit in the "ARM" position.

A pilot's remote switch, located on the left side panel, is provided to allow the transmitter to be controlled from inside the cabin. The pilot's remote switch is placarded "ON," "ARM," "OFF RESET." If the pilot's remote switch has been placed in the "ON" position for any reason, the "OFF RESET" position must be selected for one second before the switch is placed in the "ARM" position.

The unit is equipped with a portable antenna to allow the locator to be removed from the airplane in case of an emergency and used as a portable signal transmitter.

*Optional equipment



Air Conditioner Controls

The locator should be checked during the ground check to make certain the unit has not been accidentally activated. Check by tuning a radio receiver to 121.5 MHz. If there is an oscillating sound, the locator may have been activated and should be turned off immediately. Reset to the "ARM" position and check again to insure against outside interference.

NOTE

If for any reason a test transmission is necessary, the test transmission should be conducted only in the first five minutes of any hour and limited to three audio sweeps. If tests must be made at any other time, the tests should be coordinated with the nearest FAA tower or flight service station.

AIR CONDITIONING*

To operate the air conditioning system either on the ground or in flight:

1. Start the engine (ground operation).
2. Turn the air conditioning Master Switch to "ON."
3. Turn "TEMP" control to desired temperature. Clockwise rotation increases cooling.
4. Select desired "FAN" position, "LOW," "MED" or "HIGH."

*Optional equipment

AIR CONDITIONER OPERATIONAL CHECK PROCEDURE

Prior to takeoff the air conditioner should be checked for proper operation as follows:

1. Check aircraft Master Switch ON.
2. Select desired "FAN" position, "LOW," "MED" or "HIGH."
3. Turn the air conditioner control switch to "ON." The "Air Cond. Door Open" warning light will turn on, thereby indicating proper air conditioner condenser door actuation.
4. Turn the air conditioner control switch to "OFF." The "Air Cond. Door Open" warning light will go out, thereby indicating the air conditioner condenser door is in the up position.
5. If the "Air Cond. Door Open" light does not respond as specified above, an air conditioner system or indicator bulb malfunction is indicated, and further investigation should be conducted prior to flight.

The above operational check may be performed during flight if an inflight failure is suspected.

AIR CONDITIONER EFFECTS ON AIRPLANE PERFORMANCE

Operation of the air conditioner will cause slight decreases in the cruise speed and range of the Cherokee 180. Power from the engine is required to run the compressor, and the condenser door, when extended, causes a slight increase in drag. When the air conditioner is turned off there is normally no measurable difference in climb, cruise or range performance of the airplane.

NOTE

To insure maximum climb performance the air conditioner must be turned off manually before takeoff to disengage the compressor and retract the condenser door. Also the air conditioner must be turned off manually before the landing approach in preparation for a possible go-around.

Although the cruise speed and range are only slightly affected by the air conditioner operation, these changes should be considered in preflight planning. To be conservative, the following figures assume that the compressor is operating continuously while the airplane is airborne. This will be the case only in extremely hot weather.

1. The decrease in true airspeed is approximately 5 mph at all power settings.
2. The decrease in range may be as much as 37 statute miles for the 50 gal. capacity.

NOTE

To read power from the Power vs. Density Altitude Chart in this manual, add 50 rpm to the value observed on the tachometer when the air conditioner is operating.

The climb performance of Cherokee 180 is not compromised measurably with the air conditioner operating since the compressor is declutched and the condenser door is retracted, both automatically, when a full throttle position is selected. When the full throttle position is not used or in the event of a malfunction which would cause the compressor to operate and the condenser door to be extended, a decrease in rate of climb of as much as 100 fpm can be expected. Should a malfunction occur which prevents condenser door retraction when the compressor is turned off, a decrease in rate of climb of as much as 50 fpm can be expected.

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OPERATING TIPS

Operating Tips 8-1

OPERATING TIPS

The following Operating Tips are of particular value in the operation of the Cherokee.

1. Learn to trim for takeoff so that only a very light back pressure on the wheel is required to lift the airplane off the ground.
2. The best speed for takeoff is about 60 MPH under normal conditions. Trying to pull the airplane off the ground at too low an airspeed decreases the controllability of the airplane in event of engine failure.
3. Flaps may be lowered at airspeeds up to 115 MPH. To reduce flap operating loads, it is desirable to have the airplane at a slower speed before extending the flaps.
4. Before attempting to reset any circuit breaker, allow a two to five minute cooling off period.
5. Before starting the engine, check that all radio switches, light switches and the pitot heat switch are in the off position so as not to create an overloaded condition when the starter is engaged.
6. The overvoltage relay is provided to protect the electronics equipment from a momentary overvoltage condition (approximately 16.5 volts and up), or a catastrophic regulator failure. In the event of a momentary condition, the relay will open and the ammeter will indicate "0" output from the alternator. The relay may be reset by switching the ALT switch to OFF for approximately 1 second and then returning the ALT switch to ON. The ALT light on the annunciator panel* will illuminate if the alternator fails. Recycle the ALT switch and check the ALT FIELD circuit breaker. If the failure persists after this action, reduce electrical loads and land as soon as practical.
7. The vacuum gauge is provided to monitor the pressure available to assure the correct operating speed of the vacuum driven gyroscopic flight instruments. It also monitors the condition of the common air filter by measuring the flow of air through the filter.
 If the vacuum gauge does not register 5" ± .10" Hg at 2000 RPM, the following items should be checked before flight:
 - a. Common air filter could be dirty or restricted.
 - b. Vacuum lines could be collapsed or broken.
 - c. Vacuum pump worn.
 - d. Vacuum regulator, not adjusted correctly. The pressure, even though set correctly, can read lower under two conditions: (1) Very high altitude, above 12,000 feet, (2) Low engine RPM, usually on approach or during training maneuvers. This is normal and should not be considered a malfunction.
8. The shape of the wing fuel tanks is such that in certain maneuvers the fuel may move away from the tank outlet. If the outlet is uncovered, the fuel flow will be interrupted and a temporary loss of power may result. Pilots can prevent inadvertent uncovering of the outlet by avoiding maneuvers which could result in uncovering the outlet.

Extreme running turning takeoffs should be avoided as fuel flow interruption may occur.

Prolonged slips or skids which result in excess of 2000 feet of altitude loss, or other radical or extreme maneuvers which could cause uncovering of the fuel outlet must be avoided as fuel flow interruption may occur when tank being used is not full.

*Serial nos. 7505001 and up

OPERATING TIPS

REVISED: NOVEMBER 21, 1975

9. Anti-collision lights should not be operating when flying through overcast and clouds, since reflected light can produce spacial disorientation. Do not operate strobe lights when taxiing in the vicinity of other aircraft.
10. The rudder pedals are suspended from a torque tube which extends across the fuselage. The pilot should become familiar with the proper positioning of his feet on the rudder pedals so as to avoid interference with the torque tube when moving the rudder pedals or operating the toe brakes.
11. In an effort to avoid accidents, pilots should obtain and study the safety related information made available in FAA publications such as regulations, advisory circulars, Aviation News, AIM and safety aids.

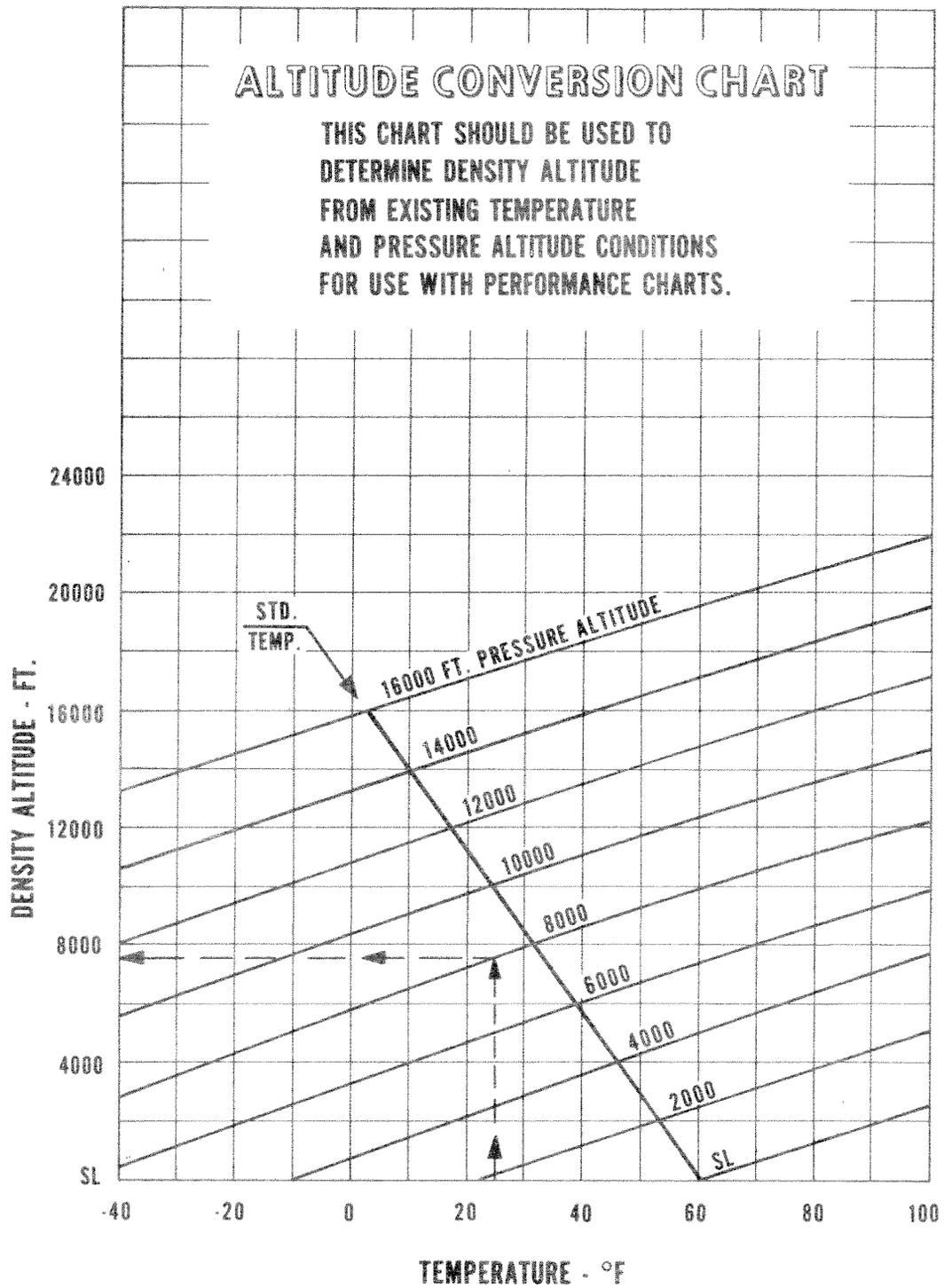
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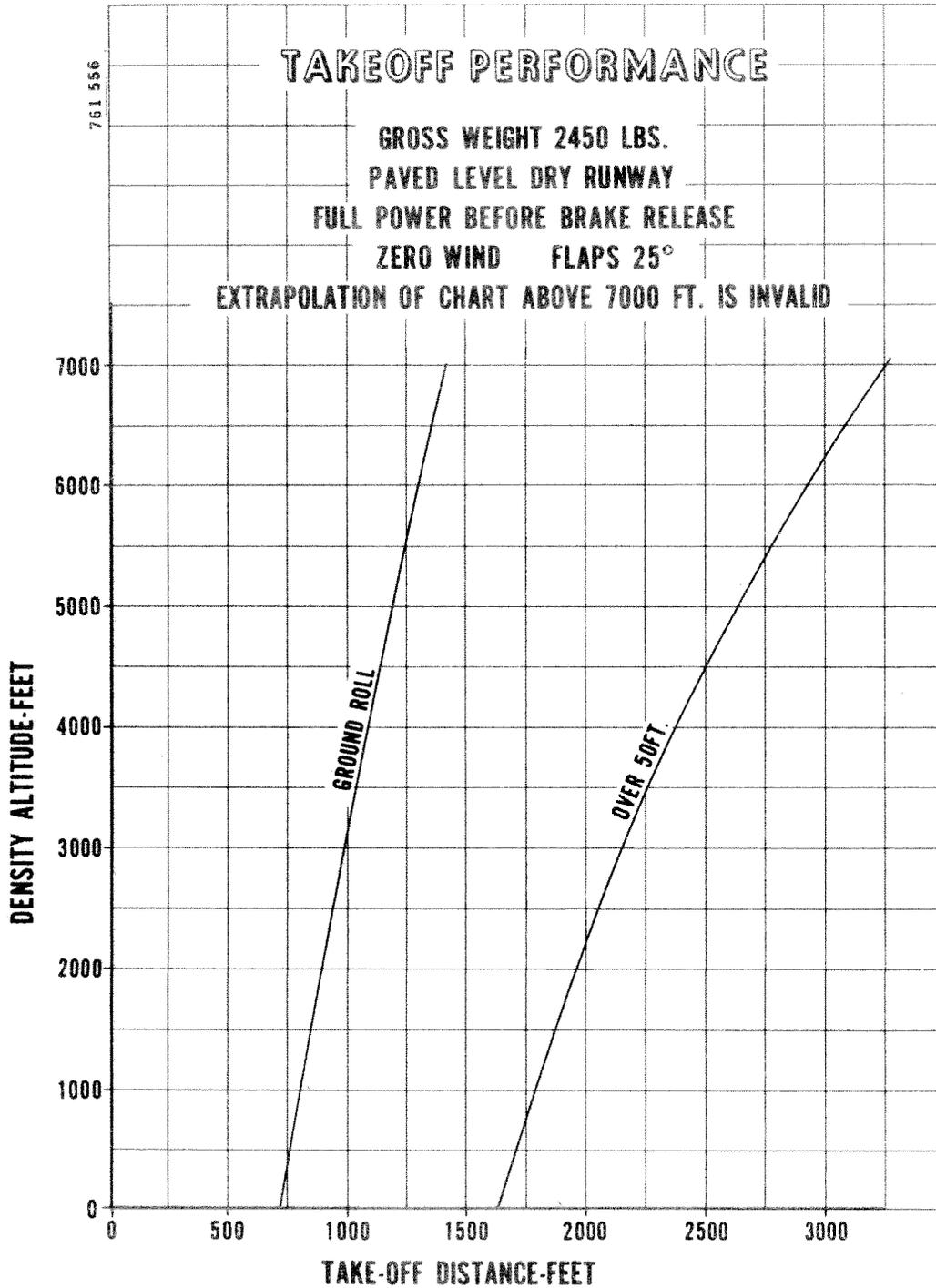
WARNING

Performance information derived by extrapolation beyond the limits shown on the charts should not be used for flight planning purposes.

PA-28-180 CHEROKEE

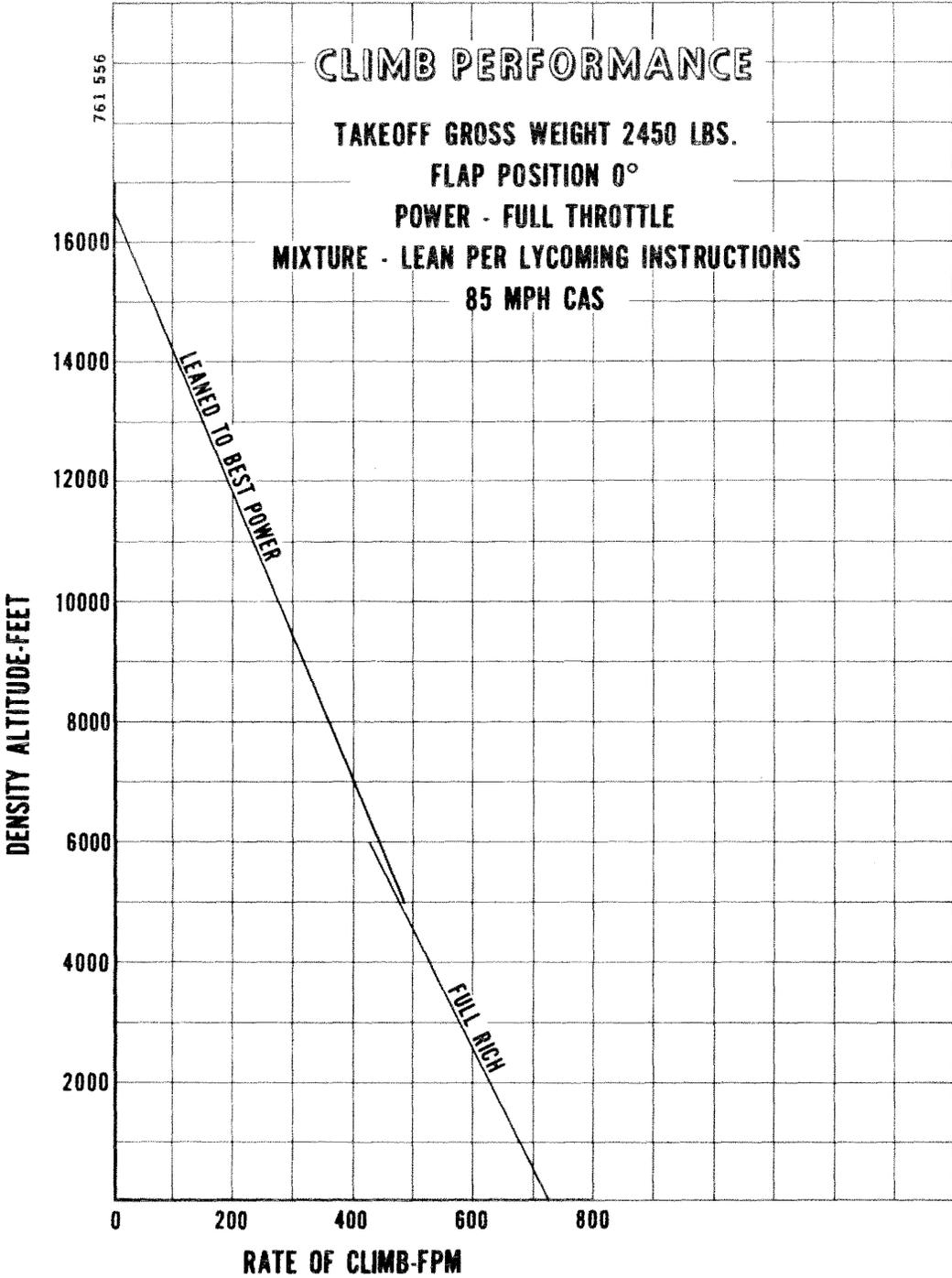


PA-28-180 CHEROKEE



NOTE: SEE SECTION 7 FOR EFFECTS OF AIR CONDITIONING
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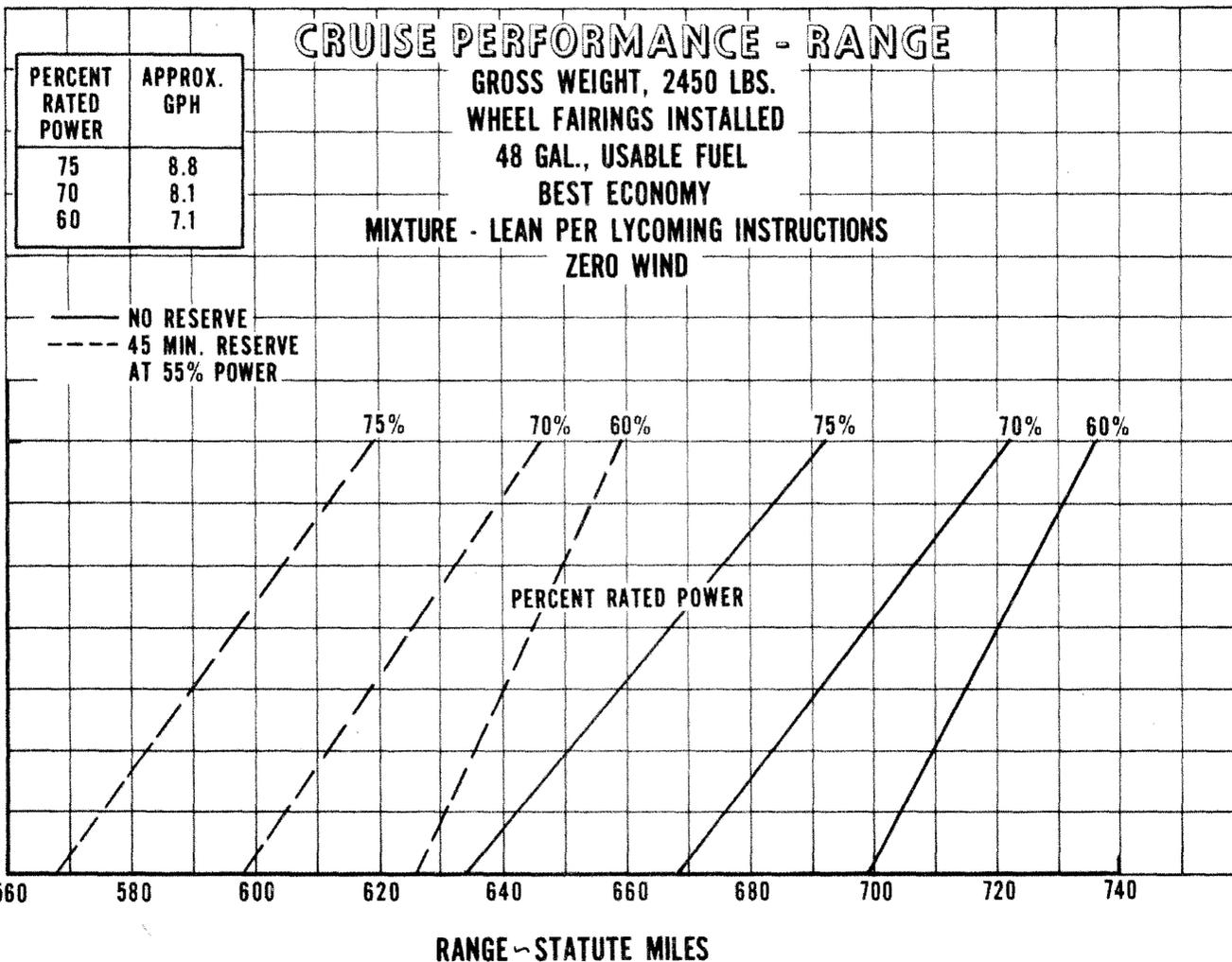


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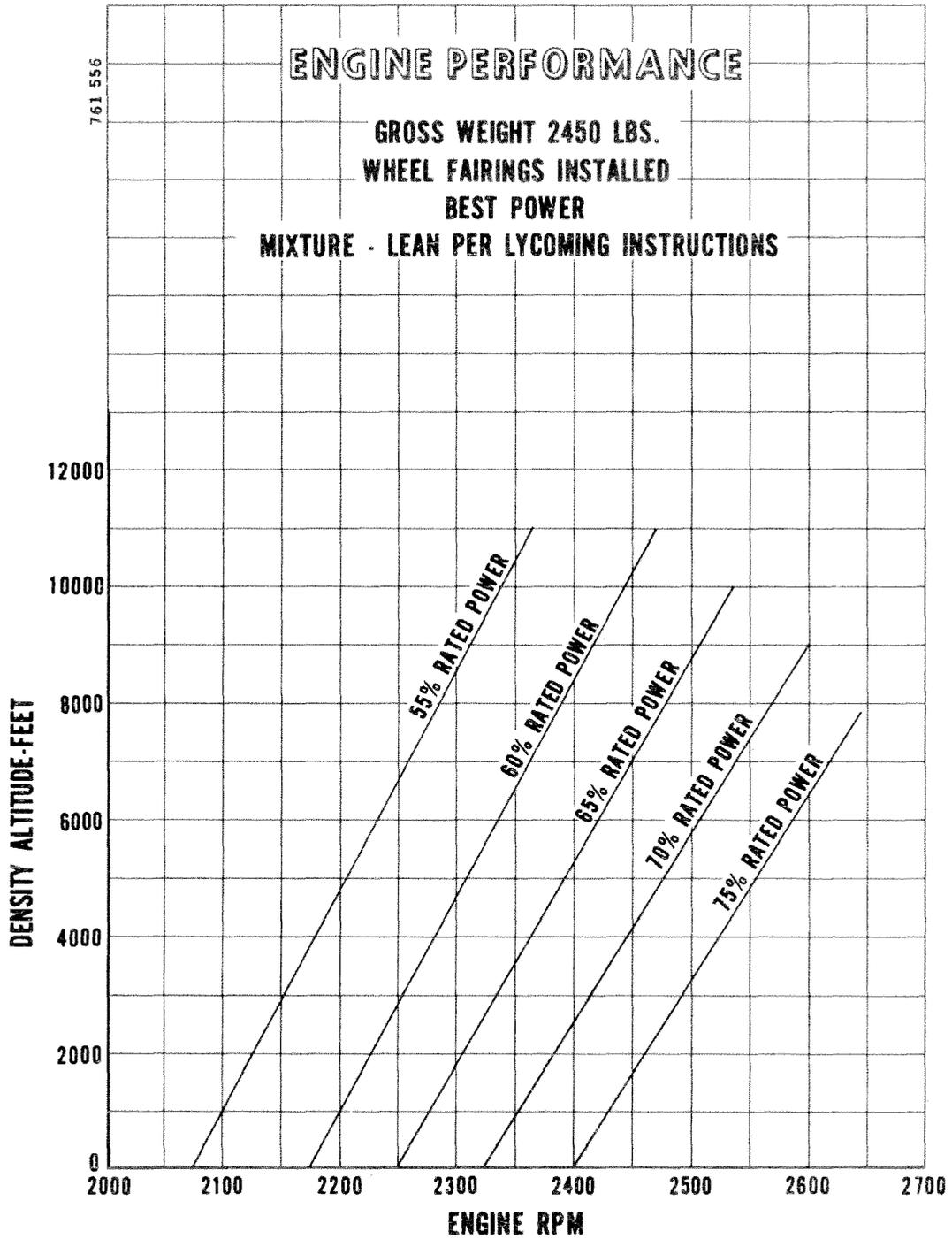
CHEROKEE ARCHER



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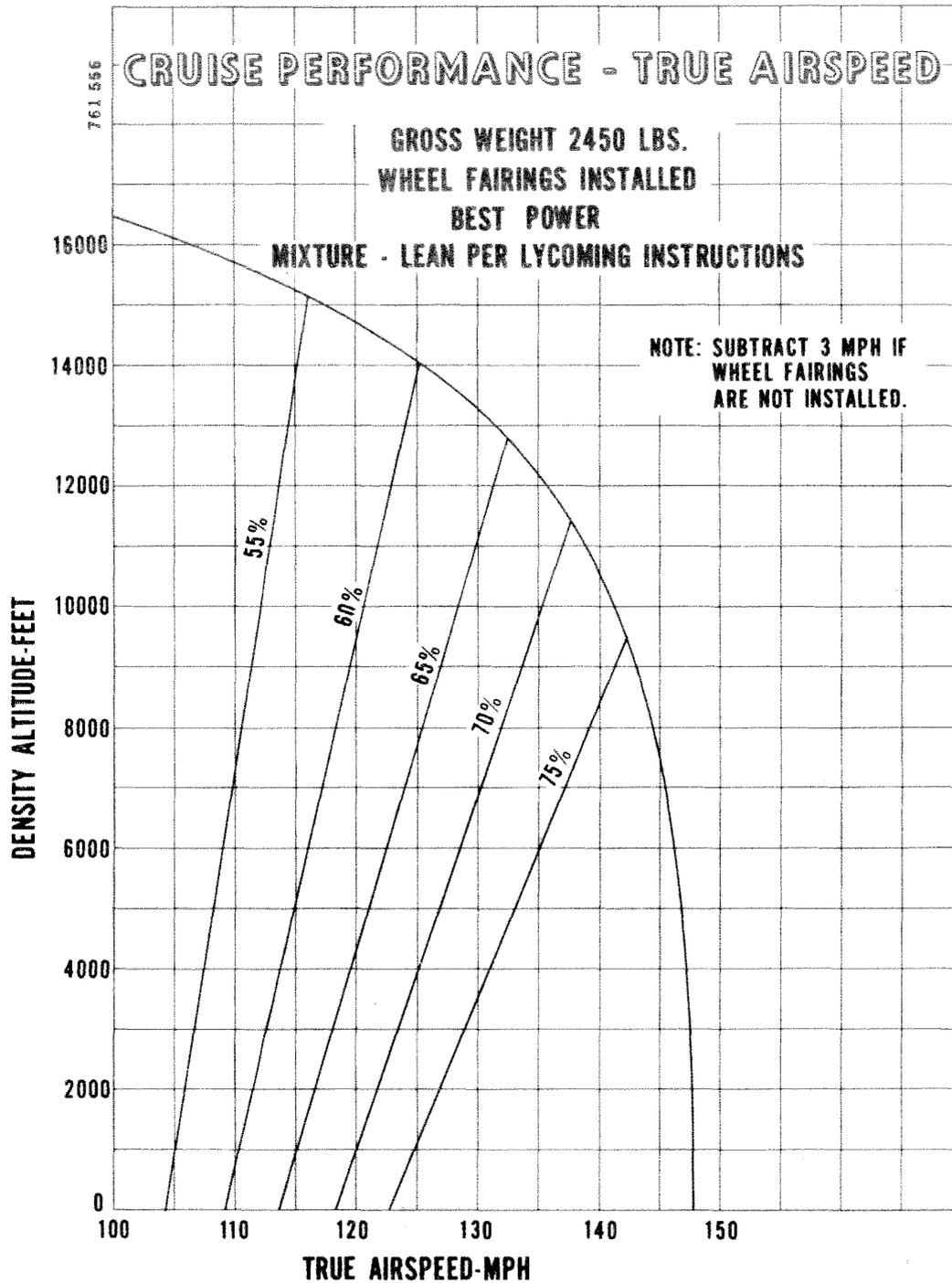
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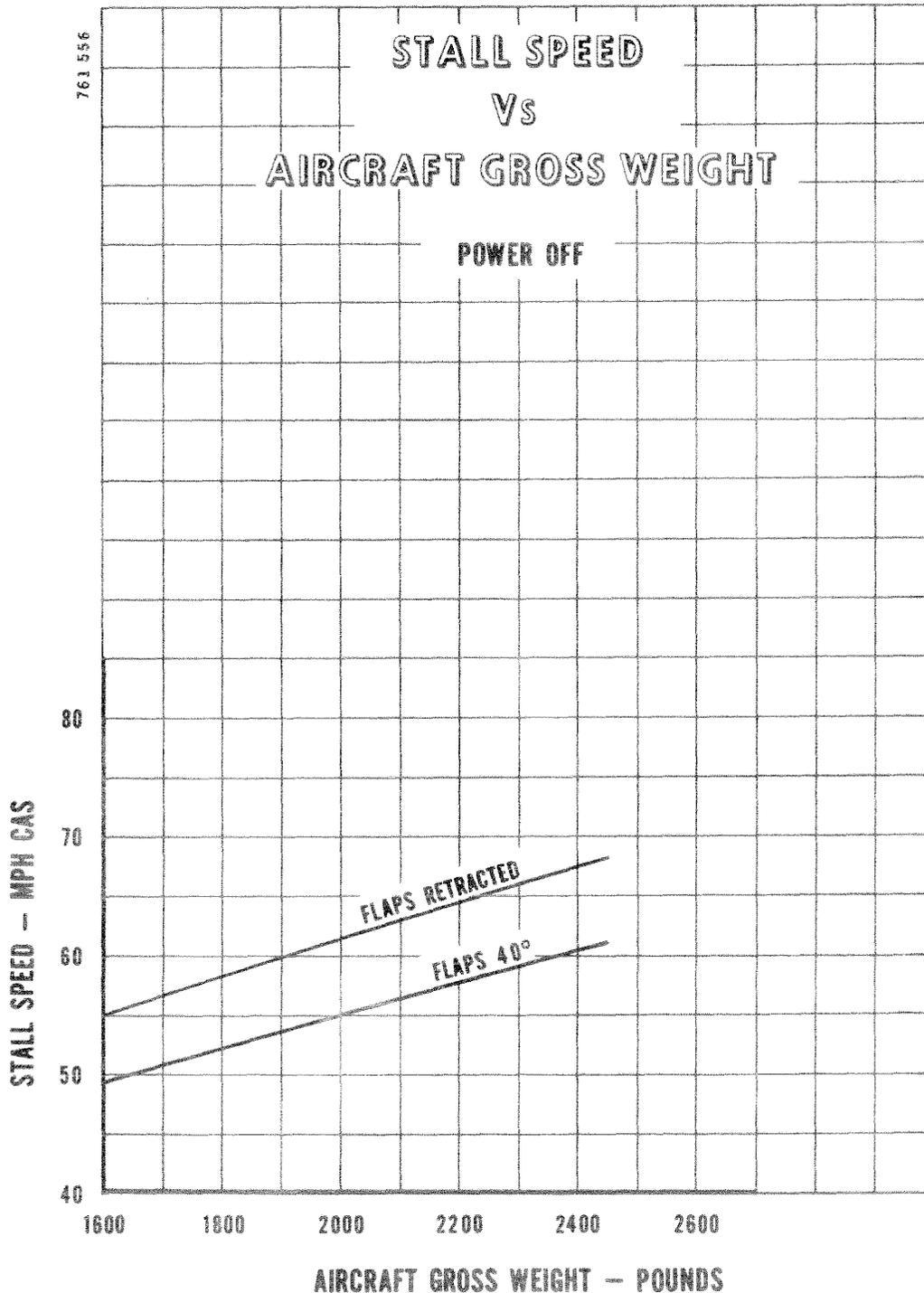
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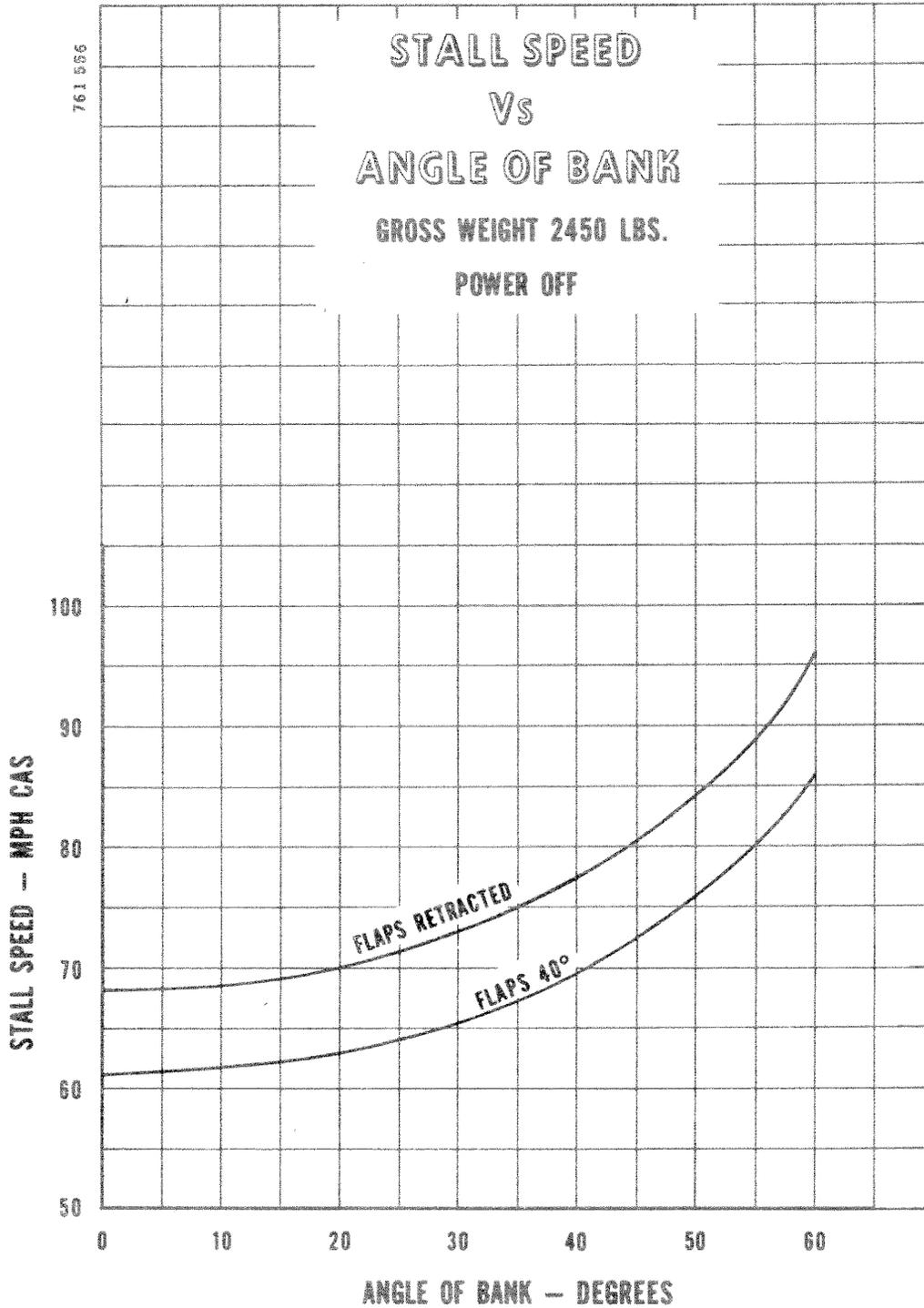
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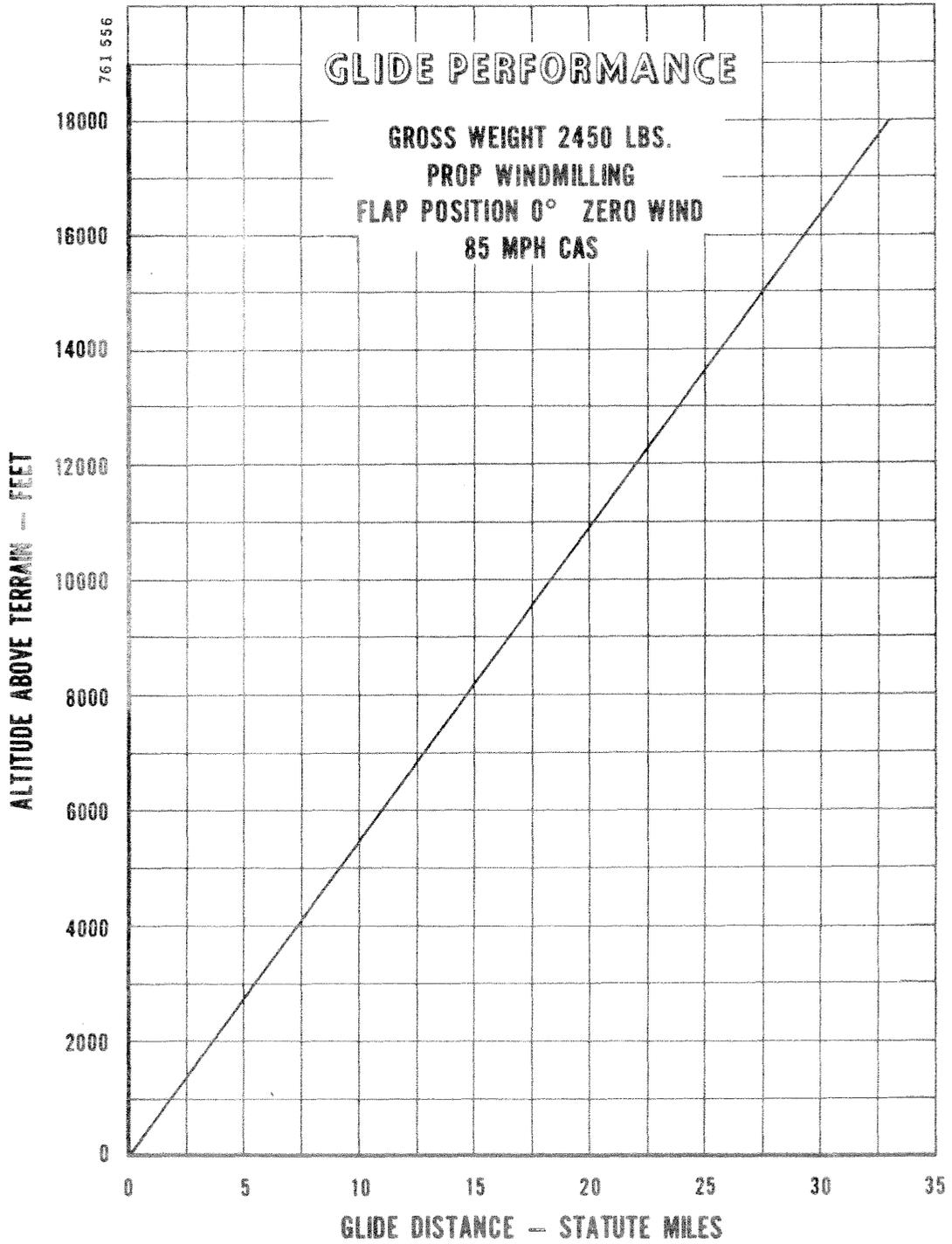
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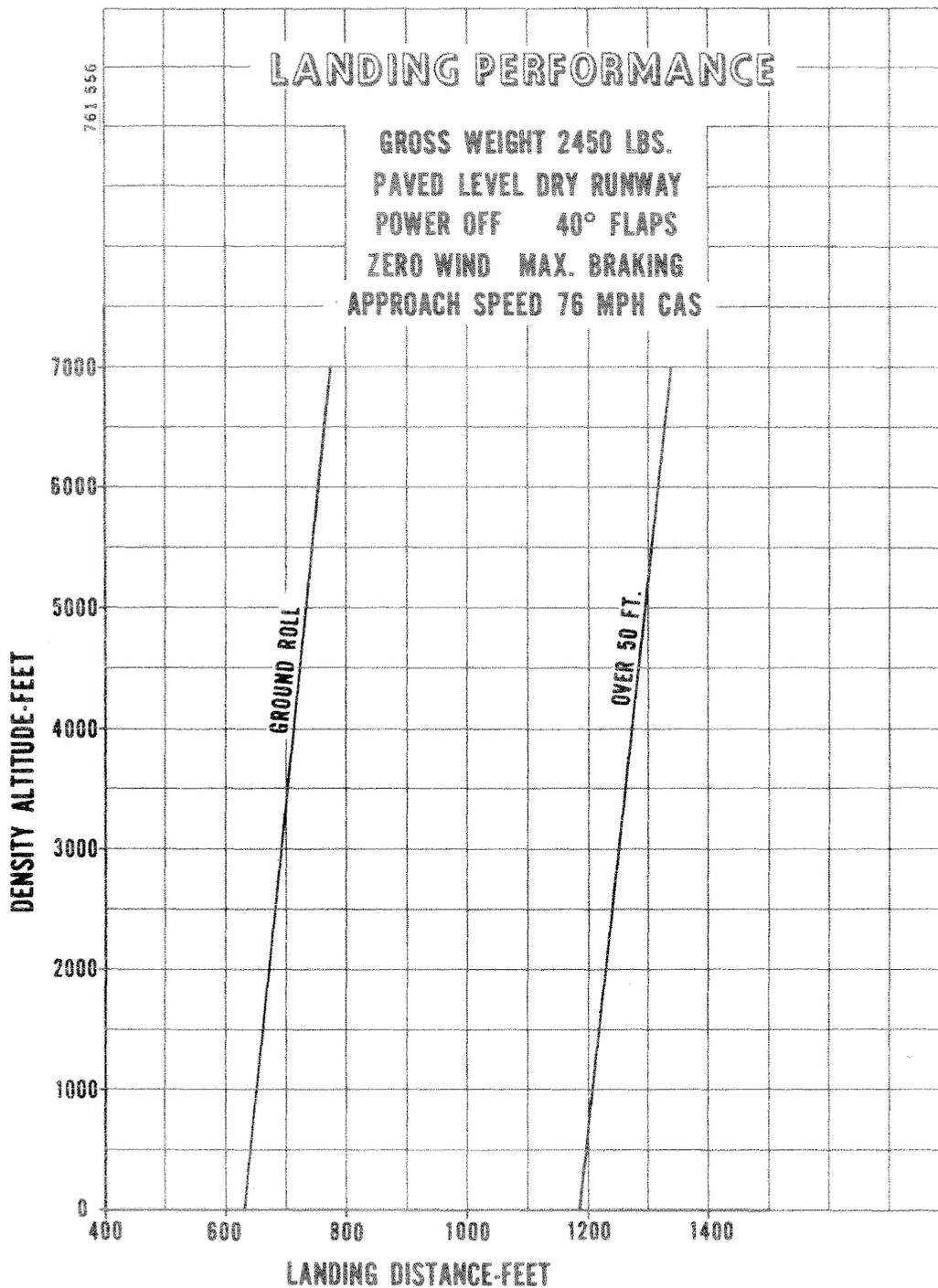
NOTE: SEE SECTION 7 FOR EFFECTS OF AIR CONDITIONING
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PA-28-180 CHEROKEE



NOTE: SEE SECTION 7 FOR EFFECTS OF AIR CONDITIONING
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PA-28-180 CHEROKEE



NOTE: SEE SECTION 7 FOR EFFECTS OF AIR CONDITIONING
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HANDLING AND SERVICING

This section contains information on preventive maintenance. Refer to the PA-28 Service Manual for further maintenance procedures. Any complex repair or modification should be accomplished by a Piper Certified Service Center.

GROUND HANDLING

TOWING

The airplane may be moved by using the nose wheel steering bar provided, or power equipment that will not damage or cause excess strain to the nose gear assembly. The steering bar is stowed in the baggage compartment.

CAUTION

When towing with power equipment, do not turn nose gear more than 30 degrees in either direction, as this will result in damage to the nose gear and steering mechanism.

TAXIING

Before attempting to taxi the airplane, ground personnel should be instructed and approved by a qualified person authorized by the owner. Engine starting and shut-down procedures as well as taxi techniques should be covered. When it is ascertained that the propeller back blast and taxi areas are clear, power should be applied to start the taxi roll, and the following checks should be performed, after releasing the parking brake.

- a. Taxi forward a few feet and apply brakes to determine their effectiveness.
- b. While taxiing, make slight turns to ascertain the effectiveness of the steering.
- c. Observe wing clearances when taxiing near buildings or other stationary objects. If possible, station a guide outside the airplane to observe.
- d. When taxiing on uneven ground, look for holes and ruts.
- e. Do not operate the engine at high RPM when running up or taxiing over ground containing loose stones, gravel or any loose material that may cause damage to the propeller blades.

PARKING

When parking the airplane, insure that it is sufficiently protected against adverse weather conditions and presents no danger to other aircraft. When parking the airplane for any length of time or overnight, it is recommended that it be moored securely.

- a. To park the airplane, head it into the wind, if possible.
- b. Set the parking brake. (Use wheel chocks if available.)

NOTE

Care should be taken when setting brakes that are overheated or during cold weather when accumulated moisture may freeze a brake.

MOORING

The airplane should be moored to insure its immovability, protection and security under varying weather conditions. The following procedure should be used for proper mooring of the airplane.

- a. Head the airplane into the wind, if possible.
- b. Lock the aileron and stabilator controls by looping the seat belt through the control wheel and pulling it snug.
- c. Block the wheels.
- d. Secure tie down ropes to the wing tie down rings and tail skid at approximately 45-degree angles to the ground. When using rope of non-synthetic material, leave sufficient slack to avoid damage to the airplane should the ropes contract.

CAUTION

Use bowline knots or locked slip knots. Do not use a plain slip knot.

NOTE

Additional preparations for high winds include using tie down ropes from the landing gear forks, and securing the rudder.

- e. Install a pitot head cover, if available.

CLEANING

CLEANING ENGINE COMPARTMENT

Before cleaning the engine compartment, place a strip of tape on the magneto vents to prevent solvent from entering these units.

- a. Place a large pan under the engine to catch waste.
- b. With the engine cowling removed, spray or brush the engine with solvent or a mixture of solvent and degreaser, as desired. Where heavy grease and dirt deposits have collected, it may be necessary to brush areas that were sprayed in order to clean them.

CAUTION

Do not spray solvent into the alternator, vacuum pump, starter or air intakes.

- c. Allow the solvent to remain on the engine from five to ten minutes. Then rinse the engine clean with additional solvent and allow to dry.

CAUTION

Do not operate the engine until excess solvent has evaporated or otherwise been removed.

- d. Remove the protective covers from the magnetos.
- e. Lubricate controls, bearing surfaces, etc., in accordance with the Lubrication Chart.

CLEANING LANDING GEAR

Before cleaning the landing gear, place a plastic cover or similar material over the wheel and brake assembly.

- a. Place a pan under the gear to catch waste.
- b. Spray or brush the gear area with solvent or a mixture of solvent and degreaser, as desired. Where heavy grease and dirt deposits have collected, it may be necessary to brush areas that were sprayed in order to clean them.
- c. Allow the solvent to remain on the gear from five to ten minutes. Then rinse the gear with additional solvent and allow to dry.
- d. Remove the cover from the wheel and remove the catch pan.
- e. Lubricate the gear in accordance with the Lubrication Chart.

CLEANING EXTERIOR SURFACES

The airplane should be washed with a mild soap and water. Harsh abrasive or alkaline soaps or detergents used on painted or plastic surfaces could make scratches or cause corrosion of metal surfaces. Cover areas where cleaning solution could cause damage. To wash the airplane, the following procedure may be used:

- a. Flush away loose dirt with water.
- b. Apply cleaning solution with a rag, sponge or soft bristle brush.
- c. To remove stubborn oil and grease, use a cloth dampened with naphtha.
- d. Where exhaust stains exist, allow solution to remain on the surface longer.
- e. Any good automotive wax may be used to preserve the painted surfaces. Soft cleaning cloths or a chamois should be used to prevent scratches when cleaning or polishing. A heavier coating of wax on the leading surfaces will reduce the abrasion problems in these areas.

CLEANING WINDSHIELD AND WINDOWS

A certain amount of care is needed to keep the plexiglas windows clean and unmarred. The following procedure is recommended:

- a. Remove dirt, mud, and other marks from exterior surface with clean water.
- b. Wash with mild soap and warm water or an aircraft plastic cleaner. Use a soft cloth or sponge using a straight rubbing motion. Do not rub surface harshly.
- c. Remove oil and grease with a cloth moistened with kerosene.

NOTE

Do not use gasoline, alcohol, benzene, carbon tetrachloride, thinner, acetone, or window cleaning sprays.

- d. After cleaning plastic surfaces, apply a thin coat of hard polishing wax. Rub lightly with a soft cloth. Do not use a circular motion.
- e. A severe scratch or mar in plastic can be removed by using jeweler's rouge to rub out the scratch. Smooth both sides and apply wax.

CLEANING HEADLINER, SIDE PANELS AND SEATS

- a. Clean headliner, side panels and seats with a stiff bristle brush, and vacuum where necessary.
- b. Soiled upholstery, except leather, may be cleaned by using an approved foam upholstery cleaner. Carefully follow the manufacturer's instructions. Avoid soaking or harsh rubbing.

CAUTION

Solvent cleaners require adequate ventilation.

CLEANING CARPETS

Use a small whisk broom or vacuum cleaner to remove dirt. For soiled spots, use a noninflammable dry cleaning fluid.

POWER PLANT INDUCTION AIR FILTER

The air filter must be cleaned at least once every fifty hours. Under extremely adverse conditions of operation it may be necessary to clean the filter daily. Extra filters are inexpensive and a spare should be kept on hand and used as a rapid replacement.

REMOVAL OF INDUCTION AIR FILTER

The filter is located on the right side and midway in the engine compartment and may be removed by the following procedure:

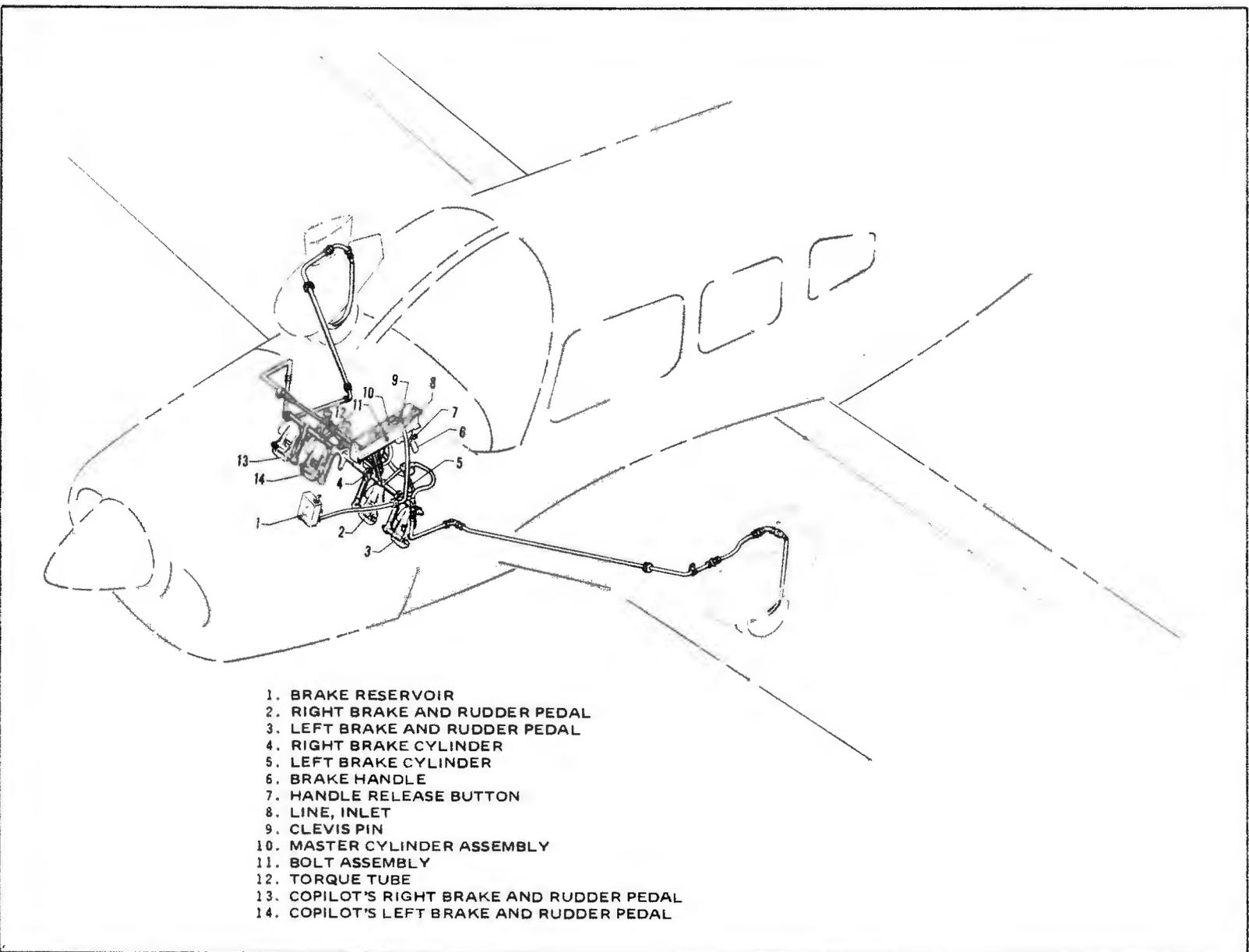
- a. Remove the two nuts and washers on the cover assembly and pull off the cover.
- b. Remove the filter.

CLEANING INDUCTION AIR FILTER

- a. Tap filter gently to remove dirt particles. Do not use compressed air or cleaning solvents.
- b. Inspect filter. If paper element is torn or ruptured or gasket is damaged, the filter should be replaced. The usable life of the filter should be restricted to one year or 500 hours, whichever comes first.

INSTALLATION OF INDUCTION AIR FILTER

After cleaning or replacing the filter, install it in the reverse order of removal.



1. BRAKE RESERVOIR
2. RIGHT BRAKE AND RUDDER PEDAL
3. LEFT BRAKE AND RUDDER PEDAL
4. RIGHT BRAKE CYLINDER
5. LEFT BRAKE CYLINDER
6. BRAKE HANDLE
7. HANDLE RELEASE BUTTON
8. LINE, INLET
9. CLEVIS PIN
10. MASTER CYLINDER ASSEMBLY
11. BOLT ASSEMBLY
12. TORQUE TUBE
13. COPILOT'S RIGHT BRAKE AND RUDDER PEDAL
14. COPILOT'S LEFT BRAKE AND RUDDER PEDAL

Brake System

BRAKE SERVICE

The brake system is filled with MIL-H-5606 (petroleum base) hydraulic brake fluid. This should be checked at every 50 hour inspection and replenished when necessary by filling the brake reservoir on the fire wall to the indicated level. If the entire system has to be refilled, it should be done by filling from the brake end of the system with fluid under pressure. This will eliminate air from the system.

No adjustment of brake clearances is necessary on the Cherokee. If after extended service the brake blocks become worn excessively, they are easily replaced with new segments.

LANDING GEAR SERVICE

The three landing gears use Cleveland 6.00 x 6 wheels, the main gear wheels being provided with brake drums and Cleveland single disc hydraulic brake assemblies. All three wheels use 6.00 x 6, four-ply rating, Type III tires with tubes.

Main wheels are removed by taking off the wheel fairings, hub cap, axle nut, and the two bolts holding the brake segment in place. The wheel will slip easily from the axle.

Tires are removed from the wheels by first deflating the tire, then removing the through bolts, and separating the wheel halves.

Landing gear oleo struts should be checked for proper strut exposures and fluid leaks. The required extensions for the strut when under normal static load (empty weight of airplane plus full fuel and oil) are 3-1/4 inches for the nose gear and 4-1/2 inches for the main gear. Should the strut exposure be below that required, it should be determined whether air or oil is required by first raising the airplane on jacks. Depress the valve core to allow air to escape from the strut housing chamber. Remove the filler plug and slowly raise the strut to full compression. If the strut has sufficient fluid, it will be visible up to the bottom of the filler plug hole and will then require only proper inflation.

Should fluid be below the bottom of the filler plug hole, oil should be added. Replace the plug with valve core removed; attach a clear plastic hose to the valve stem of the filler plug and submerge the other end in a container of hydraulic fluid (MIL-H-5606). Fully compress and extend the strut several times, thus drawing fluid from the container and expelling air from the strut chamber. To allow fluid to enter the bottom chamber of the main gear strut housing, the torque link assembly must be disconnected to let the strut be extended a minimum of 10 inches (the nose gear torque links need not be disconnected). Do not allow the strut to extend more than 12 inches. When air bubbles cease to flow through the hose, compress the strut fully and again check fluid level. Reinstall the valve core and filler plug, and the main gear torque links, if disconnected.

With fluid in the strut housing at the correct level, attach a strut pump to the air valve and with the airplane on the ground, inflate the oleo strut to the correct height.

In jacking the Cherokee for landing gear service, a jack kit (available through the Piper Dealers and Distributors) should be used. This kit consists of two hydraulic jacks and a tail stand. At least 250 pounds of ballast should be placed on the tail stand before jacking the aircraft. The jacks should be placed under the jack points on the wing and the airplane jacked up until the tail skid is at the right height to attach the tail stand. After attaching the tail stand and adding ballast, jacking may be continued until the aircraft is at the height desired.

The steering arms from the rudder pedals to the nose wheel are adjusted at the rudder pedals or at the nose wheel by turning in or out the threaded rod end bearings. Adjustment is normally accomplished at the forward end of the rods and should be done in such a way that the nose wheel is in line with the fore and aft axis of the plane when the rudder pedals and rudder are centered. Alignment of the nose wheel can be checked by pushing the airplane back and forth with the rudder centered to determine that the plane follows a perfectly straight line. The turning arc of the nose wheel is 30 degrees in either direction and is factory adjusted at stops on the bottom of the forging.

The steering arm stops should be carefully adjusted so that the nose wheel reaches its full travel just after the rudder hits its stops. This guarantees that the rudder will be allowed to move through its full travel.

PROPELLER SERVICE

The spinner and backing plate should be cleaned and inspected frequently for cracks. The propeller should be inspected before each flight for nicks, scratches and corrosion. If found, they should be taken care of as soon as possible by a rated mechanic, because nicks and scratches cause areas of increased stress which can cause serious damage or loss of a propeller tip. The back face of the blades should be painted when necessary with flat black paint to retard glare to the pilot's eyes. To prevent corrosion the surface should be cleaned and waxed periodically.

OIL REQUIREMENTS

The oil capacity of the Lycoming O-360-A4A and O-360-A4M series engines is 8 quarts and the minimum safe quantity is 2 quarts. It is recommended that the oil filter element be changed every 50 hours or sooner under unfavorable conditions. Engine oil is normally changed with the filter. However, if the full flow (cartridge type) oil filter is used and changed every 50 hours of operation, the intervals between oil changes may be increased as much as 100 percent. The following grades are recommended for the specified temperatures:

Temperatures above 60°F	S.A.E. 50
Temperatures between 30°F and 90°F	S.A.E. 40
Temperatures between 0°F and 70°F	S.A.E. 30
Temperatures below 10°F	S.A.E. 20

FUEL SYSTEM

FUEL REQUIREMENTS (AVGAS ONLY)

Aviation grade 100/130 Octane (minimum) fuel must be used in the Cherokee. The use of lower grades can cause serious engine damage in a very short period of time, and is considered of such importance that the engine warranty is invalidated by such use.

FILLING FUEL TANKS

Observe all required precautions for handling gasoline. Fuel is stored in two twenty-five gallon (24 gal. usable) tanks. To obtain the standard quantity of thirty-six U.S. gallons total, fill the tanks only to the bottom of the filler neck tube or visual indicator. To obtain the standard plus reserve quantity, fill the tanks to the top of the filler neck.

DRAINING FUEL VALVES AND LINES

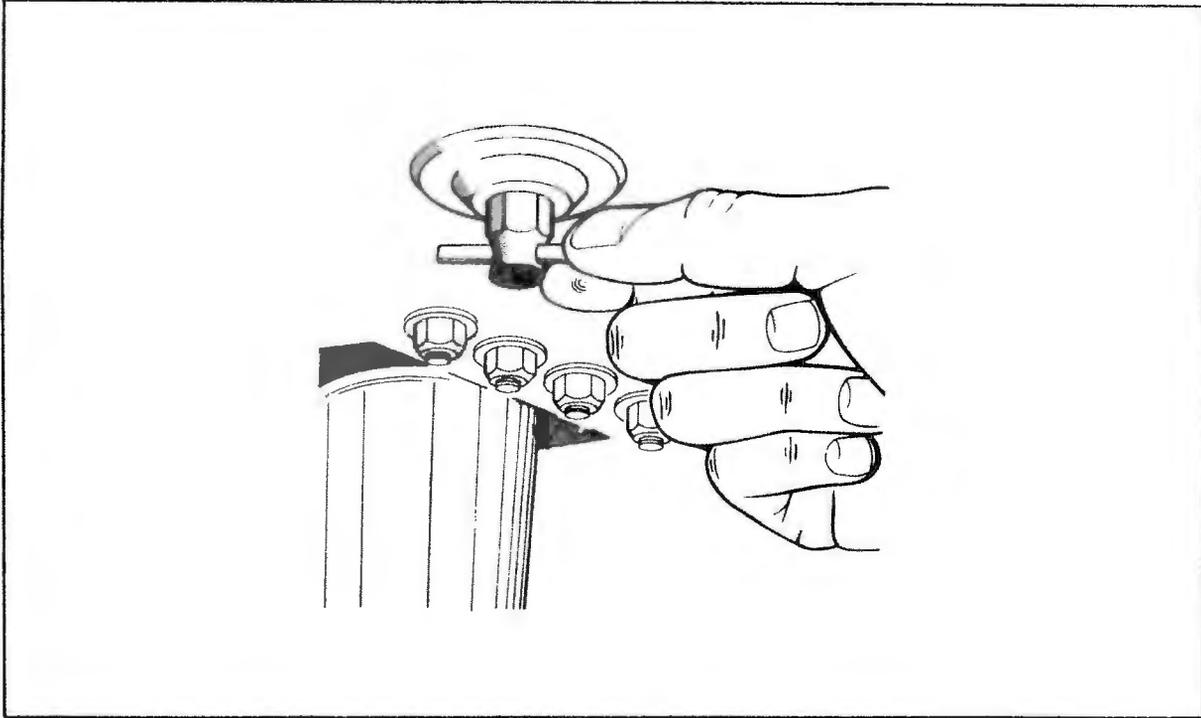
The fuel system should be drained daily prior to first flight and after refueling to avoid the accumulation of water or sediment. Each fuel tank is equipped with an individual quick drain located at the lower inboard rear corner of the tank. The fuel strainer is equipped with an easy drain valve. It is located on the front lower left corner of the fire wall. It is important that the fuel system be drained in the following manner:

- a. Open the easy drain valve with the fuel selector valve on one tank, and allow fuel to flow for a few seconds.
- b. Place a container under the drain and examine the contents for sediment, water and proper fuel.
- c. When enough fuel has flowed to ensure that the lines and strainers are free of water and sediment, close the drain and dispose of the contents of the container.
- d. Repeat the procedure with the fuel selector valve changed to the other tank.

CAUTION

When draining any amount of fuel, care should be taken to ensure that no fire hazard exists before starting engine.

After using the quick drain, it should be checked to make sure it has closed completely and is not leaking.



Fuel Drain

DRAINING FUEL SYSTEM

The bulk of the fuel may be drained from the system by opening the valve at the inboard end of each fuel tank. Push up on the arms of the drain valve and turn counterclockwise to hold the drain open. The remaining fuel in the system may be drained through the filter bowl. Any individual tank may be drained by closing the selector valve and then draining the desired tank.

TIRE INFLATION

For maximum service from the tires, keep all three tires inflated to a pressure of 24 pounds. If necessary, interchange the tires on the main wheels to produce even wear. All wheels and tires are balanced before original installation, and the relationship of the tire, tube and wheel should be maintained if possible. Out of balance wheels can cause extreme vibration on takeoff. In the installation of new components, it may be necessary to rebalance the wheel with the tire mounted.

BATTERY SERVICE

Access to the 12-volt battery is through the removal of the panel at the right rear side of the baggage compartment. The battery box has a plastic drain tube which should be opened occasionally to drain off any accumulation of liquid. Check the battery for proper fluid level. (Do not fill above the baffle plates.) Use only water - no acid. A hydrometer check should be performed to determine the percent of charge present in the battery.

If the battery is not up to charge, recharge starting at 4 amp rate and finishing with a 2 amp rate. Quick charges are not recommended.

FACTS YOU SHOULD KNOW

The Federal Aviation Administration (FAA) occasionally publishes **Airworthiness Directives (AD's)** that apply to specific groups of aircraft. They are mandatory changes and are to be complied with within a time limit set by the FAA. When an AD is issued, it is sent by the FAA to the latest registered owner of the affected aircraft and also to subscribers of their service. The owner should periodically check with his Piper dealer or A & P mechanic to see if he has the latest issued AD against his aircraft.

Piper Aircraft Corporation takes a **continuing interest** in having the owner get the most efficient use from his aircraft and keeping it in the best mechanical condition. Consequently, Piper Aircraft from time to time issues Service Bulletins, Service Letters and Service Spares Letters relating to the aircraft.

Service Bulletins are of special importance and should be complied with promptly. These are sent to the latest registered owners, distributors and dealers. Depending on the nature of the bulletin, material and labor allowances are usually applicable.

Service Letters deal with product improvements and service hints pertaining to the aircraft. They are sent to dealers and distributors so they can properly service the aircraft and keep it up to date with the latest changes. Owners should give careful attention to the Service Letter information.

Service Spares Letters offer improved parts, kits and optional equipment which were not available originally and which may be of interest to the owner.

If an owner is not having his aircraft serviced by an **Authorized Piper Service Center**, he should periodically check with a Piper dealer or distributor to find out the latest information to keep his aircraft up to date.

Piper Aircraft Corporation has a **Subscription Service** for the Service Bulletins, Service Letters and Service Spares Letters. This service is offered to interested persons such as owners, pilots and mechanics at a nominal fee, and may be obtained through Piper dealers and distributors. A Service Manual and revisions are available from a Piper dealer.

CHEROKEE ARCHER

Pilot's Operating Manual supplements are distributed by the manufacturer as necessary. These revisions and additions should be studied and put into the operating manual to keep it up to date. This manual contains important information about the operation of the aircraft and should be kept with the aircraft at all times, even after resale. Every owner, to avail himself of the Piper Aircraft Service Back-Up, should stay in close contact with a Piper dealer or distributor so that he can receive the latest information.

If the owner desires to have his aircraft modified, he must obtain FAA approval for the alteration. **Major alterations** accomplished in accordance with Advisory Circular 43.13-2, when performed by an A & P mechanic, may be approved by the local FAA office. Major alterations to the basic airframe or systems not covered by AC 43.13-2 require a Supplemental Type Certificate.

The owner or pilot is required to ascertain that the following **Aircraft Papers** are in order and in the aircraft.

- a. To be displayed in the aircraft at all times:
 1. Aircraft Airworthiness Certificate Form FAA-1362B.
 2. Aircraft Registration Certificate Form FAA-500A.
 3. Aircraft Radio Station License Form FCC-404A, if transmitters are installed.
- b. To be carried in the aircraft at all times:
 1. Aircraft Flight Manual.
 2. Weight and Balance data plus a copy of the latest Repair and Alteration Form FAA-337, if applicable.
 3. Aircraft equipment list.

Although the aircraft and engine log books are not required to be in the aircraft, they should be made available upon request. Log books should be complete and up to date. Good records will reduce maintenance cost by giving the mechanic information about what has or has not been accomplished.

PREVENTIVE MAINTENANCE

The holder of a Pilot Certificate issued under FAR Part 61 may perform certain preventive maintenance described in FAR Part 43. This maintenance may be performed only on an aircraft which the pilot owns or operates and which is not used to carry persons or property for hire, except as provided in applicable FAR's. Although such maintenance is allowed by law, each individual should make a self-analysis as to whether he has the ability to perform the work.

All other maintenance required on the airplane should be accomplished by appropriately licensed personnel.

If maintenance is accomplished, an entry must be made in the appropriate logbook. The entry should contain:

1. The date the work was accomplished.
2. Description of the work.
3. Number of hours on the aircraft.
4. The certificate number of pilot performing the work.
5. Signature of the individual doing the work.

REQUIRED SERVICE AND INSPECTION PERIODS

The **Owner Service Agreement** which the owner receives upon delivery of the aircraft should be kept in the aircraft at all times. This identifies him to authorized Piper dealers and entitles the owner to receive service in accordance with the regular service agreement terms. This agreement also entitles the transient owner full warranty by any Piper dealer in the world.

One hundred hour inspections are required by law if the aircraft is used commercially. Otherwise this inspection is left to the discretion of the owner. This inspection is a complete check of the aircraft and its systems, and should be accomplished by a Piper Authorized Service Center or by a qualified aircraft and power plant mechanic who owns or works for a reputable repair shop. The inspection is listed, in detail, in the inspection report of the appropriate Service Manual.

An **annual inspection** is required once a year to keep the Airworthiness Certificate in effect. It is the same as a 100-hour inspection except that it must be signed by an Inspection Authorized (IA) mechanic or a General Aviation District Office (GADO) representative. This inspection is required whether the aircraft is operated commercially or for pleasure.

A **Progressive Maintenance** program is approved by the FAA and is available to the owner. It involves routine and detailed inspections at 50-hour intervals. The purpose of the program is to allow maximum utilization of the aircraft, to reduce maintenance inspection cost and to maintain a maximum standard of continuous airworthiness. Complete details are available from Piper dealers.

A **spectographic analysis** of the oil is available from several sources. This system, if used intelligently, provides a good check of the internal condition of the engine. For this system to be accurate, oil samples must be sent in at regular intervals, and induction air filters must be cleaned or changed regularly.

TYPE OF LUBRICANT		PREFERRED PRODUCT AND VENDOR
IDENTIFICATION LETTER	LUBRICANT	SPECIFICATION
A	LUBRICATING OIL - GENERAL PURPOSE, LOW TEMP.	MIL-L-7870
B	LUBRICATING OIL - AIRCRAFT RECIPROCATING ENGINE (PISTON) GRADE AS SPECIFIED SAE 50 ABOVE 60°F AIR TEMP. SAE 40 30° TO 60°F AIR TEMP. SAE 30 0° TO 30°F AIR TEMP. SAE 20 BELOW 0°F AIR TEMP.	MIL-L-8082
C	HYDRAULIC FLUID.	MIL-H-5606
D	PETROLEUM BASE GREASE, AIRCRAFT AND INSTRUMENT, GEAR AND ACTUATOR SCREW GREASE, AIRCRAFT, HIGH TEMP.	MIL-G-2327
E		TEXACO WAREAK ALL PURPOSE GREASE, MOBIL GREASE 77, MOBIL GREASE 77, MOBIL GREASE 77, SHELL ALVANIA EP GREASE 2
F	PARKER "O" RING LUBRICANT	FISKE BROS. REFINING CO.
G	AERO LUBRIFLATE	
H	FLUOROCARBON RELEASE AGENT DRY LUBRICANT	* MG-122
I	GEN. PURPOSE AIRCRAFT GREASE - LUBRICATION	MIL-G-7711

SPECIAL INSTRUCTIONS (cont)

1. STABILATOR TIE-IN PULLEYS - LUBRICATION MAY BE EXTENDED TO 250 HOURS WHEN DUSTY CONDITIONS ARE AT A MINIMUM.
2. ALERON HINGES WITH TEFLON SLEEVES SHOULD NOT BE LUBRICATED. ALERON HINGES WITHOUT TEFLON SLEEVES SHOULD FIRST BE CLEARED WITH A DRY TYPE SOLVENT THEN LUBRICATED WITH MIL-L-7870 LUBRICATING OIL.
3. THIS TRANSMISSION TO BE 1/2 FULL OF GREASE. APPLY GREASE DURING ASSEMBLY AND LUBRICATE TRANSMISSION BALL NUT AND SCREW WITH MIL-G-2327 GREASE.

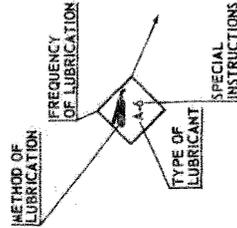
NOTES

1. PILOT AND PASSENGER SEATS - LUBRICATE TRACK ROLLERS AND STOP PINS AS REQUIRED. (TYPE OF LUBRICANT: "A")
2. WHEEL BEARINGS REQUIRE CLEANING AND REPACKING AFTER EXPOSURE TO AN ABNORMAL QUANTITY OF WATER.
3. FUEL SELECTOR VALVE - LUBRICATE FUEL SELECTOR VALVE AS REQUIRED. REFER TO PIPER SERVICE LETTER NO. 261.
4. SEE LYCOMING SERVICE INSTRUCTIONS NO. 1014 FOR USE OF DETERGENT OIL.

CAUTIONS

1. DO NOT USE HYDRAULIC FLUID WITH A CASTOR OIL OR ESTER BASE.
2. DO NOT OVER-LUBRICATE COCKPIT CONTROLS.
3. DO NOT APPLY LUBRICANT TO RUBBER PARTS.

EXAMPLE



SPECIAL INSTRUCTIONS

1. AIR FILTER - TO CLEAN FILTER, TAP GENTLY TO REMOVE DIRT PARTICLES. DO NOT BLOW OUT WITH COMPRESSED AIR OR USE OIL. REPLACE FILTER IF PUNCTURED OR DAMAGED.
2. BEARINGS AND BUSHINGS - CLEAN EXTERIOR WITH A DRY TYPE SOLVENT BEFORE LUBRICATING.
3. WHEEL BEARINGS - DISASSEMBLE AND CLEAN WITH A DRY TYPE SOLVENT. ASCERTAIN THAT GREASE IS PACKED BETWEEN THE BEARING ROLLER AND CONE. DO NOT PACK GREASE IN WHEEL HOUSING.
4. OLEO STRUTS AND BRAKE RESERVOIR - FILL PER INSTRUCTIONS ON UNIT OR CONTAINER, OR REFER TO SERVICE MANUAL, SECTION II.
5. "O" RING CONTROL SHAFT BUSHING (WITH 1.125 INCH SHAFT ONLY) - DISASSEMBLE "O" RING RETAINER PLATES FROM INSTRUMENT PANEL, LUBRICATE "O" RING AND REASSEMBLE.
6. LUBRICATION POINTS - WIPE ALL LUBRICATION POINTS CLEAN OF OLD GREASE, OIL, DIRT, ETC. BEFORE LUBRICATING.
7. INTERVALS BETWEEN OIL CHANGES CAN BE INCREASED AS MUCH AS 100% ON ENGINES EQUIPPED WITH FULL FLOW (CARTRIDGE TYPE) OIL FILTERS - PROVIDED THE ELEMENT IS REPLACED EACH 50 HOURS OF OPERATION.

Lubrication Nomenclature

PRECISE FLIGHT, INC.

AIRPLANE FLIGHT MANUAL SUPPLEMENT

FOR Piper MODEL PA28-180
REG. NO. N1185X
SER. NO. 28-7505222

This supplement must be attached to the FAA Approved Flight Manual when the airplane is modified by the installation of Precise Flight Inc. Standby Vacuum System in accordance with STC Number SA2167NM. The information contained herein supplements or supersedes the basic manual only in those areas listed. For limitations, procedures and performance information not contained in this supplement, consult the basic Airplane Flight Manual.

FAA Approved D.L. Riggins
Manager, Seattle Aircraft Certification Office
Northwest Mountain Region

Date December 7, 1984

STANDBY VACUUM SYSTEM (SVS)
OPERATING INSTRUCTIONS

I. LIMITATIONS

1. The Standby Vacuum system is for emergency or standby use only and not for dispatch purposes.
2. Vacuum powered and/or Vacuum gyro directed auto pilot operation may be unreliable when the SVS is sole source of vacuum. Vacuum powered or vacuum gyro directed auto pilot should be OFF when operating with failed primary vacuum system.
3. The SVS is not designed to operate pneumatic de-ice systems. DO NOT operate this type de-ice system when operating with a failed primary vacuum system.
4. Above 10,000 feet pressure altitude, engine power settings may have to be significantly reduced to provide adequate vacuum power for proper gyro instrument operation.

II. EMERGENCY PROCEDURES

1. In the event of (warning light) primary vacuum system failure, pull the standby vacuum knob out (ON) and reduce throttle setting as required to maintain adequate vacuum power (suction gauge reading in green arc). If necessary descend to a lower altitude to obtain a larger differential between atmospheric pressure and engine manifold pressure. Vacuum power must be closely monitored by checking vacuum gauge frequently.
2. CONTINUED IFR FLIGHT IS NOT RECOMMENDED AND IMMEDIATE ACTION SHOULD BE TAKEN TOWARD VFR CONDITION OR LANDING.
3. If descent is impracticable:
 - a. Periodically reduce power as required to "spool up" the gyros.
 - b. Reapply power as required while comparing vacuum driven gyros against the turn and bank, turn coordinator, VSI, and other flight instruments.
 - c. When an obvious discrepancy is noted between the vacuum driven instrument and the other flight instrument REPEAT the above "spool up" procedure as needed.

II. NORMAL PROCEDURES

1. Before starting engine push standby vacuum knob IN/OFF.
2. During run-up idle engine at low speed and momentarily pull standby vacuum knob out (ON) and check vacuum gauge. Normally, the vacuum reading will be slightly higher. After checking system push vacuum system control knob in (OFF).
3. Regularly check vacuum gauge for proper vacuum system operation.
4. After landing turn Standby Vacuum System OFF.

Approx. Standby Vacuum Available - Altitude - Power Chart
 for Aircraft with C.S. Prop. - Max. Cont. RPM

Press. Alt.	RPM	Man. Press.	SVS Vacuum In.Hg. Min.
2000	Max. cont.		
4000	Max. cont.		
6000	Max. cont.		
8000	Max. cont.		
10000	Max. cont.		

Approx. Standby Vacuum Available - Altitude - Power Chart
 for Aircraft with Fixed Pitch Prop

Press. Alt.	RPM	SVS Vacuum In.Hg. Min.
2000		
4000		
6000		
8000		
10000		

V. PERFORMANCE

No Change.

AA APPROVED
 at December 7, 1984

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed Precise Flight Inc. Standby Vacuum System in accordance with
STC #SA2167NM.

E N D

ADDITIONAL SHEETS ARE ATTACHED

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

FOR FAA USE ONLY

OFFICE IDENTIFICATION

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form.

1. AIRCRAFT	MAKE Piper	MODEL PA28-180
	SERIAL NO. 28-7505222	NATIONALITY AND REGISTRATION MARK N1185X
2. OWNER	NAME (As shown on registration certificate) Halzel, Lenny	ADDRESS (As shown on registration certificate) 144 President Ave. Providence, R.I. 02906

3. FOR FAA USE ONLY

4. UNIT IDENTIFICATION

5. TYPE

UNIT	MAKE	MODEL	SERIAL NO.	5. TYPE	
				REPAIR	ALTERATION
AIRFRAME	***** (As described in item 1 above) *****				
POWERPLANT	Lycoming	O-360-A4A	L-20202-36A		X
PROPELLER					
APPLIANCE	TYPE				
	MANUFACTURER				

6. CONFORMITY STATEMENT

A. AGENCY'S NAME AND ADDRESS	B. KIND OF AGENCY	C. CERTIFICATE NO.
Harold Thistlewaite 289 Pequot Trail E. Greenwich, R.I. 02818	<input checked="" type="checkbox"/> U.S. CERTIFICATED MECHANIC	A&P1522270
	<input type="checkbox"/> FOREIGN CERTIFICATED MECHANIC	
	<input type="checkbox"/> CERTIFICATED REPAIR STATION	
	<input type="checkbox"/> MANUFACTURER	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

DATE 10-2-87	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>Harold Thistlewaite</i>
-----------------	--

7. APPROVAL FOR RETURN TO SERVICE

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED

BY	FAA FLT. STANDARDS INSPECTOR	MANUFACTURER	<input checked="" type="checkbox"/> INSPECTION AUTHORIZATION	OTHER (Specify)
	FAA DESIGNEE	REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	
DATE OF APPROVAL OR REJECTION 10-2-87	CERTIFICATE OR DESIGNATION NO. 1522270	SIGNATURE OF AUTHORIZED INDIVIDUAL <i>Harold Thistlewaite</i>		

10-5-1

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
Installed Precise Flight Inc. Standby Vacuum System in accordance with Lycoming STC #SE1779NM.

E N D

ADDITIONAL SHEETS ARE ATTACHED

PULL OPERATED
CABLE VALVE

Standby Vacuum System
AFM SUPPLEMENT

FAA APPROVED AIRPLANE FLIGHT MANUAL
SUPPLEMENT
OR
SUPPLEMENTAL FLIGHT MANUAL
FOR

AIRCRAFT MODEL: Piper PA28-180
SERIAL NUMBER: 28-1505222
REGISTRATION NUMBER: N1185X

This supplement must be attached to the FAA approved Airplane Flight Manual, when the Precise Flight Standby Vacuum has been installed in accordance with STC(s).

SA2160NM, SA2161NM, SA2162NM, SA2164NM, SA2167NM,
SA2168NM, SA2683NM - Aircraft
&
SE1779NM - Lycoming Engine
or
SE1780NM - Continental Engine

The information contained in this document supplements or supersedes the basic manual only in those areas listed. For Limitations, Procedures, and Performance information not contained in this supplement, consult the basic Airplane Flight Manual.

FAA APPROVED: 
Manager,
Special Certification Branch
Seattle Aircraft Certification Office

DATE OF APPROVAL: Feb. 4, 2000

ISSUED: February 4, 2000

B. EMERGENCY PROCEDURES February 4, 2000**1. PRIMARY VACUUM FAILURE WARNING LIGHT ILLUMINATES**

- a. Pull the Standby Vacuum System knob **OUT-ON-** and adjust throttle setting as required to maintain adequate vacuum for the primary instruments - Suction Gauge Reading in the Green Arc - If necessary descend to a lower altitude to obtain a larger differential between manifold and ambient pressure. Vacuum power must be closely monitored by checking the vacuum gauge frequently.
- b. The SVS is not designed for continued IFR flight. Immediate steps should be taken to return to VFR conditions or to land. If this is not possible, IFR flight should be continued only as long as necessary to return to VFR conditions or land the airplane.

WARNING: FAILURE OF THE VACUUM SYSTEM STILL CONSTITUTES AN EMERGENCY SITUATION REGARDLESS OF THE INSTALLATION OF THE SVS. IT MAY NOT BE POSSIBLE TO MAINTAIN A SAFE ALTITUDE AND MAKE USE OF THE SVS. IN SUCH A SITUATION THE AIRPLANE MUST BE FLOWN USING NON-VACUUM POWERED INSTRUMENTS.

ISSUED: February 4, 2000

**PULL OPERATED
CABLE VALVE**

**Standby Vacuum System
AFM SUPPLEMENT**

II. OPERATING PROCEDURES

A. NORMAL PROCEDURES

1. GROUND CHECK

- a. Cycle the Standby Vacuum Control Knob **OUT - ON -**, and return Control Knob **IN - OFF -** position.

2. BEFORE TAKEOFF

- a. Idle Engine at low speed, momentarily pull the standby vacuum knob **OUT - ON -** and check vacuum gauge. Normally, the vacuum reading will be slightly higher. After checking system push Standby Vacuum System knob **IN - OFF -**. Check that vacuum gauge has returned to the previous reading.

3. ENROUTE

- a. Regularly check vacuum gauge and monitor warning light for proper vacuum system operation.

I. OPERATING LIMITATIONS (CONT.)

B. PLACARDS

One of the following placards must be placed in full view of the pilot near the instrument vacuum indicator after appropriate entries have been made.

Approximate Standby Vacuum Available - Altitude - Power Chart
for aircraft with Constant Speed Propeller - Maximum Continuous
RPM.

PRESS ALT. (FT.)	RPM	MAN. PRESSURE	SVS VACUUM IN. HG MIN.
2000	Max. Cont.		
4000	Max. Cont.		
6000	Max. Cont.		
8000	Max. Cont.		
10,000	Max. Cont.		

Approximate Standby Vacuum Available - Altitude - Power Chart
for aircraft with a Fixed Pitch Propeller

PRESS ALT. (FT.)	RPM	SVS VACUUM IN. HG MIN.
2000		
4000		
6000		
8000		
10,000		

**PULL OPERATED
CABLE VALVE**

**Standby Vacuum System
AFM SUPPLEMENT**

I. OPERATING LIMITATIONS (CONT.)

B. PLACARDS

Placard to be located on the push/pull control cable

Placard to be located around the LED for the pump inop warning light.

Placard to be placed in front and in full view of the pilot.

**STANDBY VACUUM SYSTEM EQUIPPED: FOR
OPERATING INSTRUCTIONS AND LIMITATIONS SEE
SUPPLEMENT IN OWNERS MANUAL OR PILOTS
OPERATING HANDBOOK**

ISSUED: February 4, 2000

I. OPERATING LIMITATIONS

A. INSTRUCTIONS

1. The Standby Vacuum System is for emergency or standby use only and not for dispatch purposes.
2. Vacuum powered and/or Vacuum gyro directed autopilot operation may be unreliable when the Standby Vacuum System is the sole source of vacuum. Vacuum powered or vacuum gyro directed autopilot should be OFF when operating with a failed primary vacuum system.
3. The Supplemental Vacuum System is not designed to operate pneumatic de-ice systems. DO NOT operate a pneumatic de-ice system when operating with a failed primary vacuum system.
4. Above 10,000 ft. pressure altitude, engine power settings may have to be significantly reduced to provide adequate vacuum power for proper gyro instrument operation.
5. The following placards are required to be in full view of pilot:

ISSUED: February 4, 2000

SYSTEM DESCRIPTION

A Precise Flight Standby Vacuum System may be installed to provide a temporary vacuum system in the event of a primary vacuum failure. The Standby Vacuum System operates on the differential between the intake manifold and ambient air pressure and is directed through a shuttle valve system to drive your flight instruments.

CAUTION: The use of the Standby Vacuum System requires a degree of Pilot skill and proficiency that is best maintained through practice. It is recommended, upon recurrent IFR training, in VFR conditions, in the presence of a CFI, that the aircraft be flown at the RPM and or Manifold Pressure settings found on the required placard and entered in this AFMS. This procedure will familiarize the pilot with limitations of using engine manifold vacuum for instrument power and maintaining level flight.

ISSUED: February 4, 2000

GARMIN Ltd. or its subsidiaries
c/o Garmin International
1200 E. 151st Street, Olathe, KS 66062 USA

FAA APPROVED

AIRPLANE FLIGHT MANUAL SUPPLEMENT
or
SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
for
GARMIN 400W SERIES GPS-WAAS NAVIGATION SYSTEM
as installed in

Piper PA28-180
Make and Model Airplane

Reg. No. N1185X S/N 28-7505222

This document serves as an Airplane Flight Manual Supplement or as a Supplemental Airplane Flight Manual when the aircraft is equipped with the Garmin 400W Series unit. This document must be carried in the airplane at all times when the Garmin 400W Series unit is installed in accordance with STC SA01933LA-D.

The information contained herein supplements or supersedes the information made available to the operator by the manufacturer in the form of clearly stated placards, markings, or manuals or in the form of an FAA approved Airplane Flight Manual, only in those areas listed herein. For limitations, procedures and performance information not contained in this document, consult the basic placards, markings, or manuals or the basic FAA approved Airplane Flight Manual.

FAA Approved By:



David G Armstrong
ODA STC Unit Administrator
Garmin International, Inc.
ODA-240087-CE

Date: 7/31/09

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
 or SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
 for a Garmin 400W Series Navigation System

LOG OF REVISIONS					
Rev No.	No.	Page		Description	FAA Approved
		Date			
A	All	11-20-07		Complete Supplement	<u>Sayed-Youssef Hashemi</u> Mgr. Flt. Test Br., ANM-160L FAA, Los Angeles ACO Transport Airplane Directorate Date <u>Nov. 20, 2007</u>
B	All	7/31/09		Added '-D' to STC number, added LP approach type	<u>David G. Armstrong</u> ODA STC Unit Administrator ODA-240087-CE Garmin International, Inc.

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
or SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
for a Garmin 400W Series Navigation System

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
 or SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
 for a Garmin 400W Series Navigation System

Section 1. GENERAL

1.1 Garmin 400W Series GPS/WAAS Nav Com

The Garmin 400W Series GPS/WAAS Navigator is a panel-mounted product that contains a GPS/WAAS receiver for GPS approved primary navigation, under TSO C146a (plus optional VHF Com and VHF Nav radios) in an integrated unit with a moving map and color display. The 400W Series unit features a graphical display which may also be used to depict traffic, weather, or terrain data.

The navigation functions are operated by dedicated keys and graphical menus which are controlled by the buttons and the dual concentric rotary knob along the bottom and right side of the display.

Optional VHF Com and VHF Nav radio functions are controlled via dedicated buttons and knobs on the left side of the display and adjacent to frequencies they are controlling.

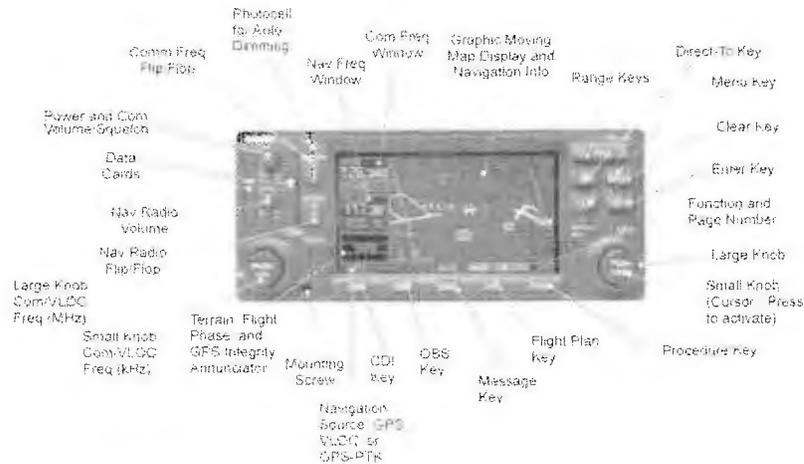


Figure 1 - 400W Series Control and Display Layout

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
or SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
for a Garmin 400W Series Navigation System

1.2 Operation

GPS/WAAS TSO-C146a Class 3 Operation: The Garmin 400W Series unit, when installed in accordance with STC SA01933LA-D, uses GPS and WAAS (within the coverage of a Space-Based Augmentation System complying with ICAO Annex 10) for enroute, terminal area, non-precision approach operations (including “GPS”, “or GPS”, “RNAV”, “LNAV”, and “LP” approaches), and approach procedures with vertical guidance (including “LNAV/VNAV” and “LPV”).

Navigation is accomplished using the WGS-84 (NAD-83) coordinate reference datum. GPS navigation data is based upon use of only the Global Positioning System (GPS) operated by the United States of America.

1.3 Class II Oceanic, Remote, and other Operations:

The Garmin 400W Series, as installed, has been found to comply with the requirements for GPS primary means of Class II navigation in oceanic and remote airspace, when used in conjunction with WAAS Garmin Prediction Program part number 006-A0154-03. Oceanic operations are supported when the 400W Series unit annunciates OCN. This provides an alarm limit of four NMI and a mask angle of five degrees. The 400W series unit also has the ability to predict RAIM availability at any waypoint in the database or if WAAS corrections are expected to be absent or disabled. This AFMS does not constitute an operational approval for Oceanic or Remote area operations. Additional equipment installations or operational approvals may be required.

- a) Oceanic navigation requires an additional approved long range oceanic and/or remote area navigation system with independent display, sensors, antenna, and power source. (It may be a second 400W/500W Series unit.)
- b) Redundant VHF Com and VHF Nav systems may be required for other than U.S. 14 CFR Part 91 operations. Check foreign regulation requirements as applicable. (It may be a second 400W/500W Series unit.)
- c) Operations approval may be granted for the use of the 400W Series unit RAIM prediction function in lieu of the Prediction Program for operators requiring this capability. Refer to your appropriate civil aviation authorities for these authorizations.

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
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for a Garmin 400W Series Navigation System

Section 2. LIMITATIONS

2.1 Pilot's Guide

The GARMIN 400W Series Pilot's Guide, part number and revision listed below (or later applicable revisions), must be immediately available for the flight crew whenever navigation is predicated on the use of the 400W Series unit.

- 400W Series Pilot's Guide & Reference P/N 190-00356-00 Rev E
- 400W/500W Series Optional Displays P/N 190-00356-30 Rev F
- 400W/500W Series Display Interfaces P/N 190-00356-31 Rev B

This AFM supplement does not grant approval for IFR operations to aircraft limited to VFR operations. Additional aircraft systems may be required for IFR operational approval. Systems limited to VFR shall be placarded in close proximity to the 400W Series unit
"GPS LIMITED TO VFR USE ONLY".

2.2 System Software:

The system must utilize the Main and GPS software versions listed below (or later FAA approved versions). The software versions are displayed on the self-test page immediately after turn-on for approximately 5 seconds or they can be accessed in the AUX pages.

Subsequent software versions may support different functions. Check the 400W Series Pilot's Guide for further information.

Table 1 - Approved Software Versions

Software Item	Approved Software Version <i>(or later FAA approved versions for this STC)</i>	
	SW version	As displayed on unit
Main SW Version	3.30	3.30
GPS SW Version	3.2	3.2

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AIRPLANE FLIGHT MANUAL SUPPLEMENT
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for a Garmin 400W Series Navigation System

2.3 Navigation Database

The 400W Series unit database card must be installed. (IAW the TSO deviations granted to Garmin for the 400W unit, navigation database cards may not be marked with the part number. The software automatically precludes invalid databases for use by the 400W)

- a) IFR enroute and terminal navigation is prohibited unless the pilot verifies the currency of the database or verifies each selected waypoint for accuracy by reference to current approved data.
- b) GPS instrument approaches using the 400W Series units are prohibited, unless the 400W Series unit's approach data is verified by the pilot or crew to be current. Instrument approaches must be accomplished in accordance with an approved instrument approach procedure that is loaded from the 400W Series unit database.
- c) Installations with dual 400W/500W Series units will only crossfill between units when they contain the same database cycle. Updating of each database must be accomplished on the ground prior to flight.

2.4 Terrain Database

The 400W Series unit supports Terrain and requires a Terrain database card to be installed in order for the feature to operate. The table below lists compatible database cards for the 400W series. Each of the data base cards contains the following data:

- a) The Terrain Database has an area of coverage from North 75° Latitude to South 60° Latitude in all longitudes.
- b) The Airport Terrain Database has an area of coverage that includes the United States, Canada, Mexico, Latin America, and South America.
- c) The Obstacle Database has an area of coverage that includes the United States, and is updated as frequently as every 56 days.

NOTE: The area of coverage may be modified as additional terrain data sources become available.

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Table 2 – Approved Terrain Database Cards

Part Number	Description
010-10201-20	Data Card, TAWS / Terrain, 128MB
010-10201-21	Data Card, TAWS / Terrain, 256MB

2.5 Navigation

No navigation is authorized north of 89° (degrees) north latitude or south of 89° (degrees) south latitude.

2.6 Approaches

- a) During GPS approaches, the pilot must verify the 400W Series unit is operating in the approach mode. (LNAV, LNAV+V, L/VNAV, LP, or LPV)
- b) When conducting approaches referenced to true North, the heading selection on the AUX pages must be adjusted to TRUE.
- c) Accomplishment of an ILS, LOC, LOC-BC, LDA, SDF, MLS, VOR approach, or any other type of approach not approved for GPS overlay, is not authorized with GPS navigation guidance.
- d) Use of the GNS 430W VOR/LOC/GS receiver to fly approaches not approved for GPS requires VOR/LOC/GS navigation data to be present on the external indicator (i.e. proper CDI source selection).
- e) For aircraft with remote source selection annunciation or remote GPS navigation annunciations installed, conducting IFR approaches is prohibited if the remote annunciation is found to be inoperative during pre-flight. (This limitation does not prohibit the conduct of an IFR approach if the required remote annunciation fails during flight. The indications provided on the 400W Series unit display may be used as a backup).
- f) Except in emergency conditions, IFR approaches are prohibited whenever any physical or visual obstruction (such as a throw-over yoke) restricts pilot view or access to the 400W Series unit or the affected CDI.

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2.7 Autopilot Coupling

IFR installations of a Garmin 400W Series unit allow the operator to fly all phases of flight based on the navigation information presented to the pilot; however, not all modes may be coupled to the autopilot. All autopilots may be coupled in Oceanic (OCN), Enroute (ENR), and Terminal (TERM) modes; however, the FAA requires that vertical coupling of an autopilot for approaches be demonstrated to meet their intended function and provide safe and proper operation to published minimums. This installation is limited to:

- No limitations for autopilot coupling.
- Lateral GPS coupling (LNAV only). For 430W units: The GS of an ILS (VLOC) may be coupled to the autopilot without any limitations.

This limitation may be removed after an FAA Flight Test demonstration. Contact Garmin International, Tech Support for additional information.

2.8 Terrain Display

Terrain refers to the display of terrain information. Pilots are NOT authorized to deviate from their current ATC clearance to comply with terrain/obstacle alerts. Terrain unit alerts are advisory only and are not equivalent to warnings provided by TAWS. Navigation must not be predicated upon the use of the terrain display.

The terrain display is intended to serve as a situational awareness tool only. By itself, it may not provide either the accuracy or the fidelity on which to base decisions and plan maneuvers to avoid terrain or obstacles.

2.9 VNAV

VNAV information may be utilized for advisory information only. Use of VNAV information for Instrument Approach Procedures does not guarantee Step-Down Fix altitude protection, or arrival at approach minimums in a normal position to land.

2.10 Weather Display

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If an optional weather receiver is interfaced to the 400W Series unit, the weather information displayed is limited to supplemental use only and may not be used in lieu of an official weather data source.

2.11 Traffic Display

Traffic may be displayed on the 400W Series unit when connected to an approved optional TCAS, TAS, or TIS traffic device. These systems are capable of providing traffic monitoring and alerting to the pilot. Traffic shown on the display may or may not have traffic alerting available. The display of traffic is an aid to visual acquisition and may not be utilized for aircraft maneuvering. Display of this traffic data and related operations are described in the 400W Series unit Pilot's Guide.

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Section 3. EMERGENCY PROCEDURES

3.1 Emergency Procedures

No change.

3.2 Abnormal Procedures

- a) If the Garmin 400W Series unit GPS navigation information is not available, or is invalid, utilize other remaining operational navigation equipment installed in the airplane as appropriate. If the 400W Series unit loses GPS position and reverts to Dead Reckoning mode (indicated by the annunciation of "DR" in the lower left of the display), the moving map will continue to be displayed. Aircraft position will be based upon the last valid GPS position and estimated by Dead Reckoning methods. Changes in airspeed or winds aloft can affect the estimated position substantially. Dead Reckoning is only available in Enroute mode; Terminal and Approach modes do not support DR.
- b) If a "Loss of Integrity" (INTEG) message is displayed during:
 - Enroute/Terminal: continue to navigate using GPS equipment and periodically cross-check the GPS guidance to other approved means of navigation.
 - GPS Approach: GPS approaches are not authorized under INTEG - Execute missed approach or revert to alternate navigation.
- c) During a GPS LPV precision approach or GPS LNAV/VNAV approach, the 400W Series unit will downgrade the approach if the Vertical alarm limits are exceeded. This will cause the vertical guidance to flag as unavailable. The procedure may be continued using the LNAV only minimums.
- d) During a GPS LP approach, the 400W Series may downgrade the approach prior to the Final Approach Fix if alarm limits are exceeded. If this occurs, a message will be displayed advising the pilot to use LNAV minimums. If alarm limits are exceeded after the Final Approach Fix, the 400W Series unit will flag the lateral guidance and generate a system message "ABORT APPROACH loss of navigation". Immediately upon viewing the message the unit will revert to Terminal alarm limits. If the position integrity is within these limits lateral guidance will be restored

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and the GPS may be used to execute the missed approach, otherwise alternate means of navigation should be utilized.

- e) During any GPS approach in which precision and non-precision alarm limits are exceeded, the 400W Series unit will flag the lateral guidance and generate a system message "ABORT APPROACH loss of navigation". Immediately upon viewing the message the unit will revert to Terminal alarm limits. If the position integrity is within these limits lateral guidance will be restored and the GPS may be used to execute the missed approach, otherwise alternate means of navigation should be utilized.

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Section 4. NORMAL PROCEDURES

Refer to the 400W Series unit Pilot's Guide defined in paragraph 2.1 on page 6 of this document for normal operating procedures. This includes all GPS operations, VHF COM and NAV, and Multi-Function Display information. For information on TIS traffic, or data linked weather see the Pilot's Guide addendum for optional displays. For information on active traffic sensor or Stormscope operation and displays see the Pilot's Guide addendum for display interfaces.

Although intuitive and user friendly the 400W Series unit requires a reasonable degree of familiarity to prevent operations without becoming too engrossed at the expense of basic instrument flying in IMC and basic see-and-avoid in VMC. Pilot workload will be higher for pilots with limited familiarity in using the unit in an IFR environment, particularly without the autopilot engaged. Garmin provides excellent training tools with the Pilot's Guide and PC based simulator. Pilots should take full advantage of these training tools to enhance system familiarization. Use of an autopilot is strongly encouraged when using the 400W Series unit in IMC conditions

4.1 Approaches with Vertical Guidance

The 400W Series unit supports three types of GPS approaches with vertical guidance: LPV approaches, LNAV/VNAV (annunciated as L/VNAV) approaches, and LNAV approaches with advisory vertical guidance (annunciated as LNAV+V). For LNAV approaches with advisory vertical guidance, the 400W Series will annunciate LNAV+V indicating vertical guidance is available. LNAV minimums will be controlling in this case.

NOTE:

If flying an LPV or LNAV/VNAV approach, be prepared to fly the LNAV only approach prior to reaching the final approach fix (FAF). If the GPS integrity is not within vertical approach limits, the system will flag the vertical guidance. This may be annunciated by a downgrade to LNAV message.

For additional information on approaches with vertical guidance refer to the 400W Series unit Pilot's Guide.

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4.2 Approaches without Vertical Guidance

The 400W Series unit supports Localizer Performance approaches (annunciated as LP). Published LP minimums will be controlling in this case.

NOTE:

If flying an LP approach, be prepared to fly the LNAV only approach prior to reaching the final approach fix (FAF). If the GPS integrity is not within LP approach limits, the system will notify the pilot by a downgrade to LNAV message.

For additional information on LP approaches refer to the 400W Series unit Pilot's Guide.

4.3 Autopilot Operation

The Garmin 400W Series may be coupled to an optional autopilot if installed in the aircraft when operating as prescribed in the LIMITATIONS section of this manual. For lateral guidance, some installations may utilize GPSS or GPS Roll Steering in lieu of the analog deviation information. If an HSI is used with GPSS engaged, the pilot should rotate the course pointer as prompted on the 400W Series unit to prevent loss of situational awareness and to prevent the aircraft from turning inappropriately if the autopilot is switched from digital (GPSS) to analog mode. For autopilot operational instructions, refer to the FAA approved Flight Manual or Flight Manual Supplement for the autopilot.

4.4 Coupling the Autopilot during approaches

The Garmin 400W Series supports analog and digital (GPSS) control interfaces to an optionally installed autopilot. Some autopilots revert to ROLL mode (wings level) and/or flag a NAV failure if the digital data becomes unavailable or is inhibited. The CDI selection of VLOC should inhibit the digital control interface. When switching between GPS and VLOC the pilot should be aware that the autopilot may need to be re-engaged into APR or NAV mode after changing the CDI source.

Autopilot coupling to GPS vertical guidance requires that the autopilot be engaged in an analog APR mode identical to coupling to an ILS. Some autopilots may revert to ROLL mode when the navigation outputs of the 400W Series unit sequence to the final approach fix. In these installations

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the unit will be configured to PROMPT the pilot to “Enable the autopilot approach outputs” in order to prevent the autopilot from entering ROLL mode without the pilot being aware of the transition.

- This installation prompts the pilot and requires the pilot to enable the A/P outputs just prior to engaging the autopilot in APR mode.
- This installation supports a seamless transition from digital (GPSS) to analog guidance for the autopilot. To capture the vertical guidance, the pilot may engage the autopilot in APR mode at any time when the GPS Glide Slope (VDI) becomes valid (displayed without a FLAG).
- This installation interfaces to the autopilot in analog mode only. To capture the vertical guidance, the pilot may engage the autopilot in APR mode at any time when the GPS Glide Slope (VDI) becomes valid.
- The autopilot does not support any vertical capture or tracking in this installation.

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Analog only autopilots should use APR mode for coupling to LNAV approaches. Autopilots which support digital roll steering commands (GPSS) may utilize NAV mode and take advantage of the digital tracking during LNAV only approaches.

4.5 WFDE Prediction Program

The Garmin WAAS Fault Detection and Exclusion (WFDE) Prediction Program is required for Remote/Oceanic operations.

The Prediction Program should be used in conjunction with the Garmin 400W/500W Simulator. After entering the intended route of flight in the Simulator flight plan the pilot selects the FDE Prediction Program under the Options menu of the Simulator program.

For detailed information refer to the WFDE prediction program instructions (190-00643-01). The availability of FDE is only required for Oceanic or Remote operations.

Section 5. PERFORMANCE

No change.

Section 6. WEIGHT AND BALANCE

See current weight and balance data.

Section 7. SYSTEM DESCRIPTIONS

See Garmin 400W Series unit Pilot's Guide for a complete description of the 400W Series unit.